

of 5,000 hours TIS to the total hours TIS indicated on the component history card or equivalent record for the collective lever P/N 412-010-408-101.

Note 1 to paragraph (g)(1)(i): Bell Helicopter service information identifies helicopters with serial numbers 36570, 36579, 36587, and 36593 through 36602 inclusive, as helicopters originally delivered with a Strake and FastFin system installed.

(ii) For helicopters with a Strake and FastFin system P/N 412-705-040-101 installed after delivery from the manufacturer, add a permanent penalty of 50% of the total hours TIS accumulated by the collective lever P/N 412-010-408-101 on the component history card or equivalent record for the collective lever P/N 412-010-408-101.

Note 2 to paragraph (g)(1)(ii): The Accomplishment Instructions, part II, paragraph 2., of Bell Helicopter Alert Service Bulletin 412-12-151, Revision A, dated July 8, 2014, provides an example of calculating and adding a permanent penalty of 50%.

(iii) For helicopters without a Strake and FastFin system P/N 412-705-040-101 installed, but with a collective lever P/N 412-010-408-101 installed, add a permanent penalty of 50% of the total hours TIS accumulated by the collective lever on the component history card or equivalent record for the collective lever.

(2) Before further flight, remove from service any collective lever P/N 412-010-408-101 that has reached or exceeded its life limit of 10,000 total hours TIS. Thereafter, remove from service each collective lever P/N 412-010-408-101 on or before reaching its life limit of 10,000 total hours TIS.

(3) As of the effective date of this AD, do not install a new (zero total hours TIS) collective lever P/N 412-010-408-101 unless a permanent penalty of 5,000 hours TIS has been added to the total hours TIS on its component history card or equivalent record.

(4) As of the effective date of this AD, do not install a used collective lever P/N 412-010-408-101 unless a permanent penalty of 50% of the total hours TIS accumulated by the collective lever has been added to the total hours TIS on its component history card or equivalent record.

(h) Special Flight Permits

Special flight permits are prohibited.

(i) Alternative Methods of Compliance (AMOCs)

(1) The Manager, DSCO Branch, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the certification office, send it to the attention of the person identified in paragraph (j) of this AD. Information may be emailed to: 9-ASW-190-COS@faa.gov.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(j) Related Information

For more information about this AD, contact Haytham Alaidy, Aerospace Engineer, Certification & Program Management Section, DSCO Branch, Compliance & Airworthiness Division, FAA, 10101 Hillwood Pkwy., Fort Worth, TX 76177; telephone (817) 222-5224; email haytham.alaidy@faa.gov.

Issued on August 19, 2021.

Ross Landes,

Deputy Director for Regulatory Operations, Compliance & Airworthiness Division, Aircraft Certification Service.

[FR Doc. 2021-18383 Filed 8-26-21; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2021-0661; Project Identifier AD-2020-01349-E]

RIN 2120-AA64

Airworthiness Directives; Pratt & Whitney Turbofan Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: The FAA proposes to supersede Airworthiness Directive (AD) 2011-07-02, which applies to all Pratt & Whitney (P&W) JT8D-209, JT8D-217, JT8D-217A, JT8D-217C, and JT8D-219 model turbofan engines. AD 2011-07-02 requires initial and repetitive torque inspections of the 3rd-stage and 4th-stage low-pressure turbine (LPT) blades. AD 2011-07-02 also requires replacement of the LPT blade if wear limits are exceeded, replacement of the LPT-to-exhaust case bolts and nuts, and installation of crushable sleeve spacers on the bolts. Since the FAA issued AD 2011-07-02, the FAA received a report of an MD-82 airplane, equipped with a JT8D-217 engine, experiencing an engine surge that resulted in the fracture of an LPT blade. This proposed AD would retain certain requirements of AD 2011-07-02, while revising the inspection thresholds and replacement intervals for the 3rd-stage and 4th-stage LPT blades. The FAA is proposing this AD to address the unsafe condition on these products.

DATES: The FAA must receive comments on this proposed AD by October 12, 2021.

ADDRESSES: You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- **Federal eRulemaking Portal:** Go to <https://www.regulations.gov>. Follow the instructions for submitting comments.

- **Fax:** (202) 493-2251.

- **Mail:** U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE, Washington, DC 20590.

- **Hand Delivery:** Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this NPRM, contact Pratt & Whitney, 400 Main Street, East Hartford, CT 06118; phone: (800) 565-0140; email: help24@prattwhitney.com; website: <https://fleetcare.prattwhitney.com>. You may view this service information at the FAA, Airworthiness Products Section, Operational Safety Branch, 1200 District Avenue, Burlington, MA 01803. For information on the availability of this material at the FAA, call (781) 238-7759.

Examining the AD Docket

You may examine the AD docket at <https://www.regulations.gov> by searching for and locating Docket No. FAA-2021-0661; or in person at Docket Operations between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this NPRM, any comments received, and other information. The street address for Docket Operations is listed above.

FOR FURTHER INFORMATION CONTACT: Nicholas Paine, Aviation Safety Engineer, ECO Branch, FAA, 1200 District Avenue, Burlington, MA 01803; phone: (781) 238-7116; fax: (781) 238-7199; email: nicholas.j.paine@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

The FAA invites you to send any written relevant data, views, or arguments about this proposal. Send your comments to an address listed under **ADDRESSES**. Include "Docket No. FAA-2021-0661; Project Identifier AD-2020-01349-E" at the beginning of your comments. The most helpful comments reference a specific portion of the proposal, explain the reason for any recommended change, and include supporting data. The FAA will consider all comments received by the closing date and may amend the proposal because of those comments.

Except for Confidential Business Information (CBI) as described in the following paragraph, and other information as described in 14 CFR 11.35, the FAA will post all comments received, without change, to <https://www.regulations.gov>.

www.regulations.gov, including any personal information you provide. The agency will also post a report summarizing each substantive verbal contact received about this proposed AD.

Confidential Business Information

CBI is commercial or financial information that is both customarily and actually treated as private by its owner. Under the Freedom of Information Act (FOIA) (5 U.S.C. 552), CBI is exempt from public disclosure. If your comments responsive to this NPRM contain commercial or financial information that is customarily treated as private, that you actually treat as private, and that is relevant or responsive to this NPRM, it is important that you clearly designate the submitted comments as CBI. Please mark each page of your submission containing CBI as "PROPIN." The FAA will treat such marked submissions as confidential under the FOIA, and they will not be placed in the public docket of this NPRM. Submissions containing CBI should be sent to Nicholas Paine, Aviation Safety Engineer, ECO Branch, FAA, 1200 District Avenue, Burlington, MA 01803. Any commentary that the FAA receives which is not specifically designated as CBI will be placed in the public docket for this rulemaking.

Background

The FAA issued AD 2011-07-02, Amendment 39-16639 (76 FR 16526, March 24, 2011), (AD 2011-07-02), for all P&W JT8D-209, JT8D-217, JT8D-217A, JT8D-217C, and JT8D-219 model turbofan engines. AD 2011-07-02 was prompted by nine reports of failure of Tinidur material LPT-to-exhaust case bolts. AD 2011-07-02 requires initial and repetitive torque inspections of the 3rd-stage and 4th-stage LPT blades, replacement of the LPT blade if wear limits are exceeded, and replacement of the LPT-to-exhaust case bolts and nuts with longer bolts and nuts made of Tinidur material. AD 2011-07-02 also

requires installation of crushable sleeve spacers on the bolts. The agency issued AD 2011-07-02 to prevent an LPT blade failure that could result in uncontained engine debris and damage to the airplane.

Actions Since AD 2011-07-02 Was Issued

Since the FAA issued AD 2011-07-02, the agency received a report of an MD-82 airplane, equipped with JT8D-217C model turbofan engines that, on approach to Taipei Songshan Airport, experienced an engine surge on the number one engine resulting in LPT blade fracture and uncontained LPT blade failure. An inspection by the manufacturer determined that this event was caused by shroud notch wear of the LPT blades, which led to changes in the vibration mode and subsequent high-cycle fatigue of the airfoil. In addition to this event, the FAA received reports of five events that involved uncontained failure of the LPT blades on the affected engines. Based on its investigation of these events, P&W determined that revised or more restrictive inspection thresholds and replacement intervals of the 3rd-stage and 4th-stage LPT blades are necessary and revised its service information accordingly.

FAA's Determination

The FAA is issuing this NPRM after determining that the unsafe condition described previously is likely to exist or develop on other products of the same type design.

Related Service Information Under 1 CFR Part 51

The FAA reviewed Pratt & Whitney Alert Service Bulletin (ASB) No. JT8D A6224, Revision No. 7, dated August 26, 2019. This service information specifies procedures for the initial and repetitive torque inspections of the 3rd-stage and 4th-stage LPT blades for shroud notch wear at revised inspection thresholds and intervals. This service information is reasonably available because the

interested parties have access to it through their normal course of business or by the means identified in ADDRESSES.

Other Related Service Information

The FAA reviewed Pratt & Whitney ASB No. JT8D A6494, Revision No. 1, dated January 26, 2010, Pratt & Whitney ASB JT8D A6507, dated November 2, 2020, and Sections 72-53-12 through 72-53-13 of Pratt & Whitney Engine Maintenance Manual (EMM), Part No. 773128, Revision 107, dated October 15, 2020. Pratt & Whitney ASB No. JT8D A6494, Revision No. 1, dated January 26, 2010, describes procedures for replacing the LPT-to-exhaust case bolts and nuts and installing the crushable sleeve spacers. Pratt & Whitney ASB JT8D A6507, dated November 2, 2020, describes procedures for replacing the 3rd-stage and 4th-stage LPT blades. Sections 72-53-12 through 72-53-13 of Pratt & Whitney EMM, Part No. 773128, Revision 107, dated October 15, 2020, describe procedures for inspecting and repairing the 3rd-stage and 4th-stage LPT blades.

Proposed AD Requirements in This NPRM

This proposed AD would retain certain requirements of AD 2011-07-02. This proposed AD would require an initial torque inspection of certain 3rd-stage LPT blades and repetitive torque inspections of 4th-stage LPT blades for shroud notch wear at revised inspection thresholds and intervals. This proposed AD would also require replacement of the 3rd-stage and 4th-stage LPT blades before accumulating 5,000 hours time-in-service.

Costs of Compliance

The FAA estimates that this AD, if adopted as proposed, would affect 42 engines installed on airplanes of U.S. registry.

The FAA estimates the following costs to comply with this proposed AD:

ESTIMATED COSTS

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Inspect 3rd-stage and 4th-stage LPT blades	1 work-hour × \$85 per hour = \$85 ..	\$0	\$85	\$3,570
Replace 3rd-stage and 4th-stage LPT blades	150 work-hours × \$85 per hour = \$12,750.	350,000	362,750	15,235,500
Replace the LPT-to-exhaust case bolts and nuts and install the crushable sleeve spacers.	1.5 work-hours × \$85 per hour = 127.50.	4,576	4,703.50	197,547

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue

rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII,

Aviation Programs, describes in more detail the scope of the Agency's authority.

The FAA is issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, General requirements. Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

The FAA determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the proposed regulation:

- (1) Is not a “significant regulatory action” under Executive Order 12866,
- (2) Would not affect intrastate aviation in Alaska, and
- (3) Would not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by:
- a. Removing Airworthiness Directive AD 2011–07–02, Amendment 39–16639 (76 FR 16526, March 24, 2011); and
 - b. Adding the following new airworthiness directive:

Pratt & Whitney: Docket No. FAA–2021–0661; Project Identifier AD–2020–01349–E.

(a) Comments Due Date

The FAA must receive comments on this airworthiness directive (AD) action by October 12, 2021.

(b) Affected ADs

This AD replaces AD 2011–07–02, Amendment 39–16639 (76 FR 16526, March 24, 2011).

(c) Applicability

This AD applies to Pratt & Whitney (P&W) JT8D–209, JT8D–217, JT8D–217A, JT8D–217C, and JT8D–219 model turbofan engines.

(d) Subject

Joint Aircraft System Component (JASC) Code 7230, Turbine Engine Compressor Section.

(e) Unsafe Condition

This AD was prompted by a report of an MD–82 airplane, equipped with a JT8D–217C model turbofan engine, experiencing an engine surge that resulted in the fracture of the low-pressure turbine (LPT) blade and uncontained release of the LPT blade. Five prior uncontained LPT blade failures were also reported on affected model turbofan engines. The FAA is issuing this AD to prevent LPT blade fracture and uncontained release of the LPT blade. The unsafe condition, if not addressed, could result in uncontained engine debris, damage to the engine, and damage to the aircraft.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Required Actions

(1) For JT8D–209, JT8D–217, and JT8D–217A model turbofan engines, within the compliance times specified in the Accomplishment Instructions, Part 1: JT8D–209, –217, –217A Engines (Part 1), paragraph 1.A., of P&W Alert Service Bulletin No. JT8D A6224, Revision No. 7, dated August 26, 2019 (the ASB), perform an initial torque inspection for shroud notch wear of the 3rd-stage LPT blades using the procedures in Part 1, paragraph 1, of the ASB.

(i) Thereafter, within the applicable reinspection interval specified in Table 1—Reinspection Interval for all 3rd Stage Blades, of the ASB, repeat the torque inspection for shroud notch wear required by paragraph (g)(1) of this AD.

(ii) If the results of the torque inspection required by paragraphs (g)(1) or (g)(1)(i) of this AD meet the criteria for engine removal specified in Table 1—Reinspection Interval for all 3rd Stage Blades, of the ASB, perform piece-part inspections in accordance with the Instructions for Continued Airworthiness (ICA) on all 3rd-stage LPT blades before exceeding 20 hours time-in-service (TIS) since the last torque inspection.

(2) For JT8D–209, JT8D–217, and JT8D–217A model turbofan engines, within the compliance times specified in Table A or Table B, of the ASB, as applicable, perform an initial torque inspection for shroud notch wear of the 4th-stage LPT blades using the procedures in Part 1, paragraph 1, of the ASB. Wherever the ASB refers to “Revision

7 Release Date” and “At SB Release Date,” use the effective date of this AD.

(i) For engines in which the last inspection prior to the effective date of this AD had a torque inspection result of less than 15 LB–IN on any 4th-stage LPT blade, perform piece-part inspections in accordance with the ICA on all 3rd-stage and 4th-stage LPT blades within 20 hours TIS after the effective date of this AD.

(ii) Thereafter, within the applicable reinspection interval specified in Table 2—Reinspection Interval for all 4th Stage Blades, of the ASB, repeat the torque inspection for shroud notch wear required by paragraph (g)(2) of this AD.

(iii) If the results of the torque inspection required by paragraphs (g)(2) or (g)(2)(ii) of this AD meet the criteria for engine removal specified in Table 2—Reinspection Interval for all 4th Stage Blades, of the ASB, perform piece-part inspections in accordance with the ICA on all 3rd-stage and 4th-stage LPT blades before exceeding 20 hours TIS since the last torque inspection.

(3) For JT8D–217C and JT8D–219 model turbofan engines, within the compliance times specified in Table A or Table B, of the ASB, as applicable, perform an initial torque inspection for shroud notch wear of the 4th-stage LPT blades using the procedures in the Accomplishment Instructions, Part 2: JT8D–217C, –219 Engines (Part 2), paragraph 1, of the ASB. Wherever the ASB refers to “Revision 7 Release Date” and “At SB Release Date,” use the effective date of this AD.

(i) For engines in which the last inspection prior to the effective date of this AD had a torque inspection result of less than 15 LB–IN on any 4th-stage LPT blade, perform piece-part inspections in accordance with the ICA on all 3rd-stage and 4th-stage LPT blades within 20 hours TIS after the effective date of this AD.

(ii) Thereafter, within the reinspection interval specified in Table 3—Reinspection Interval for all 4th Stage Blades, of the ASB, repeat the torque inspection for shroud notch wear required by paragraph (g)(3) of this AD.

(iii) If the results of the torque inspection required by paragraph (g)(3) and (g)(3)(ii) of this AD meet the criteria for engine removal specified in Table 3—Reinspection Interval for all 4th Stage Blades, of the ASB, perform piece-part inspections in accordance with the ICA on all 3rd-stage and 4th-stage LPT blades before exceeding 20 hours TIS since the last torque inspection.

(4) At the first engine shop visit after January 1, 2023, or prior to accumulating 5,000 TIS on the 3rd-stage and 4th-stage LPT blades, whichever occurs later, but not to exceed 6 years after the effective date of the AD, replace the 3rd-stage and 4th-stage LPT blades with parts eligible for installation.

(5) Thereafter, prior to accumulating 5,000 hours TIS on the 3rd-stage and 4th-stage LPT blades since their last replacement, replace the 3rd-stage and 4th-stage LPT blades with parts eligible for installation.

(6) After every replacement of the 3rd-stage or 4th-stage LPT blades, perform initial and repetitive torque inspections of the 3rd-stage or 4th-stage LPT blades using, as applicable, the accomplishment instructions and

compliance times in Part 1, paragraph 1, or Part 2, paragraph 1, of the ASB.

(i) If the results of the torque inspection required by paragraph (g)(6) of this AD meet the criteria for engine removal specified in Table 1, 2 or 3, of the ASB, as applicable, perform piece-part inspections in accordance with the ICA on all 3rd-stage and 4th-stage LPT blades before exceeding 20 hours TIS since the last torque inspection.

(ii) [Reserved]

(7) The initial inspection or the reinspection interval should not be reset unless the blades are refurbished. Whenever a used blade is reinstalled in a rotor, the previous used time should be subtracted from the initial inspection threshold.

(8) Whenever a refurbished or used blade is intermixed with zero hours time-since-new (TSN) blades in a rotor, use the lowest initial inspection threshold that is applicable.

(9) At the next accessibility to the LPT-to-exhaust case bolts and nuts after the effective date of this AD, do the following:

(i) Replace the bolts with part number (P/N) MS9557–26 bolts;

(ii) Replace the nuts with P/N 375095 nuts or P/N 490270 nuts; and

(iii) Install crushable sleeve spacers, P/N 822903, under the head of the bolts.

Note 1 to paragraph (g): Guidance on replacing the 3rd-stage and 4th-stage LPT blades can be found in P&W ASB JT8D A6507, dated November 2, 2020.

Note 2 to paragraph (g): Guidance on replacing the LPT-to-exhaust case bolts and nuts and installing the crushable sleeve spacers can be found in P&W ASB No. JT8D A6494, Revision No. 1, dated January 26, 2010.

(h) Definitions

For the purpose of this AD:

(1) An “engine shop visit” is the induction of an engine into the shop for maintenance involving the separation of pairs of major mating engine flanges, except that the separation of engine flanges solely for the purposes of transportation without subsequent engine maintenance does not constitute an engine shop visit.

(2) Accessibility to the LPT-to-exhaust case bolts refers to maintenance involving the inner turbine fan ducts being removed from the engine.

(3) Parts eligible for installation are 3rd-stage or 4th-stage LPT blades with less than 5,000 hours TIS.

(4) A “piece-part inspection” is when the blades are removed from the rotor.

(5) A “used blade” refers to a 3rd-stage or 4th-stage LPT blade that has more than zero hours TSN.

(i) Credit for Previous Actions

You may take credit for any initial torque inspection for shroud notch wear required by paragraphs (g)(1) through (3) of this AD if you performed the initial inspection before the effective date of this AD using P&W ASB No. JT8D A6224, Revision No. 5, dated June 11, 2004, or Revision No. 6, dated May 3, 2007.

(j) Alternative Methods of Compliance (AMOCs)

(1) The Manager, ECO Branch, FAA, has the authority to approve AMOCs for this AD,

if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ECO Branch, send it to the attention of the person identified in Related Information. You may email your request to: ANE-AD-AMOC@faa.gov.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(k) Related Information

(1) For more information about this AD, contact Nicholas Paine, Aviation Safety Engineer, ECO Branch, FAA, 1200 District Avenue, Burlington, MA 01803; phone: (781) 238–7116; fax: (781) 238–7199; email: nicholas.j.paine@faa.gov.

(2) For service information identified in this AD, contact Pratt & Whitney, 400 Main Street, East Hartford, CT 06118; phone: (800) 565–0140; email: help24@prattwhitney.com; website: <https://fleetcare.prattwhitney.com>. You may view this referenced service information at the FAA, Airworthiness Products Section, Operational Safety Branch, 1200 District Avenue, Burlington, MA 01803. For information on the availability of this material at the FAA, call (781) 238–7759.

Issued on August 5, 2021.

Lance T. Gant,

Director, Compliance & Airworthiness Division, Aircraft Certification Service.

[FR Doc. 2021–18489 Filed 8–26–21; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2021–0712; Project Identifier 2019–CE–018–AD]

RIN 2120–AA64

Airworthiness Directives; ASI Aviation (Type Certificate Previously Held by Reims Aviation S.A.) Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: The FAA proposes to supersede Airworthiness Directive (AD) 2015–16–07 R1, which applies to certain Reims Aviation S.A. (type certificate now held by ASI Aviation) Model F406 airplanes. AD 2015–16–07 R1 requires inspecting the left-hand and right-hand rudder control pedal torque tubes and replacing with a serviceable part as necessary. Since the FAA issued AD 2015–16–07 R1, the European Aviation Safety Agency (EASA)

superseded its mandatory continuing airworthiness information (MCAI) to correct an unsafe condition on these products. This proposed AD would retain the requirements of AD 2015–16–07 R1, expand the applicability, and require repeating the inspections using updated procedures. The FAA is proposing this AD to address the unsafe condition on these products.

DATES: The FAA must receive comments on this proposed AD by October 12, 2021.

ADDRESSES: You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- *Federal eRulemaking Portal:* Go to <https://www.regulations.gov>. Follow the instructions for submitting comments.

- *Fax:* (202) 493–2251.

- *Mail:* U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12 140, 1200 New Jersey Avenue SE, Washington, DC 20590.

- *Hand Delivery:* Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this NPRM, contact ASI Aviation, Aérodrôme de Reims Prunay, 51360 Prunay, France; telephone: +33 3 26 48 46 84; fax: +33 3 26 49 18 57; email: contact@asi-aviation.fr; website: <https://asi-aviation.fr/page-Accueil.html>. You may view this service information at the Airworthiness Products Section, Operational Safety Branch, FAA, 901 Locust, Kansas City, MO 64106. For information on the availability of this material at the FAA, call (816) 329–4148.

Examining the AD Docket

You may examine the AD docket at <https://www.regulations.gov> by searching for and locating Docket No. FAA–2021–0712; or in person at Docket Operations between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this NPRM, the MCAI, any comments received, and other information. The street address for Docket Operations is listed above.

FOR FURTHER INFORMATION CONTACT: Gregory Johnson, Aviation Safety Engineer, FAA, General Aviation & Rotorcraft Section, International Validation Branch, 901 Locust, Room 301, Kansas City, MO 64106; phone: (720) 626–5462; fax: (816) 329–4090; email: gregory.johnson@faa.gov.

SUPPLEMENTARY INFORMATION: