

for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the certification office, send it to the attention of the person identified in paragraph (j)(2) of this AD. Information may be emailed to: *9-ANM-LAACO-AMOC-Requests@faa.gov*.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair, modification, or alteration required by this AD if it is approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has been authorized by the Manager, Los Angeles ACO Branch, FAA, to make those findings. To be approved, the repair method, modification deviation, or alteration deviation must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(4) Except as required by paragraph (h)(2) of this AD: For service information that contains steps that are labeled as RC, the provisions of paragraphs (i)(4)(i) and (i)(4)(ii) of this AD apply.

(i) The steps labeled as RC, including substeps under an RC step and any figures identified in an RC step, must be done to comply with the AD. If a step or substep is labeled "RC Exempt," then the RC requirement is removed from that step or substep. An AMOC is required for any deviations to RC steps, including substeps and identified figures.

(ii) Steps not labeled as RC may be deviated from using accepted methods in accordance with the operator's maintenance or inspection program without obtaining approval of an AMOC, provided the RC steps, including substeps and identified figures, can still be done as specified, and the airplane can be put back in an airworthy condition.

(j) Related Information

(1) For more information about this AD, contact Lu Lu, Aerospace Engineer, Airframe Section, FAA, Seattle ACO Branch, 2200 South 216th St., Des Moines, WA 98198; phone and fax: 206-231-3525; email: *lu.lu@faa.gov*.

(2) For information about AMOCs, contact George Garrido, Aerospace Engineer, Airframe Section, FAA, Los Angeles ACO Branch, 3960 Paramount Boulevard, Lakewood, CA 90712-4137; phone: 562-627-5232; fax: 562-627-5210; email: *george.garrido@faa.gov*.

(3) For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Contractual & Data Services (C&DS), 2600 Westminister Blvd., MC 110-SK57, Seal Beach, CA 90740-5600; telephone 562-797-1717; internet <https://www.myboeingfleet.com>. You may view this referenced service information at the FAA, Transport Standards Branch, 2200 South 216th St., Des Moines, WA. For information on the availability of this material at the FAA, call 206-231-3195.

Issued in Des Moines, Washington, on October 4, 2018.

Michael Kaszycki,

Acting Director, System Oversight Division, Aircraft Certification Service.

[FR Doc. 2018-22277 Filed 10-15-18; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2018-0900; Product Identifier 2018-NM-101-AD]

RIN 2120-AA64

Airworthiness Directives; The Boeing Company Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to adopt a new airworthiness directive (AD) for certain The Boeing Company Model 737-100, -200, -200C, -300, -400, and -500 series airplanes. This proposed AD was prompted by reports of cracking in the frame web, frame integral inboard chord, and fail-safe chord on multiple airplanes in multiple locations below the passenger floor, in addition to an evaluation by the design approval holder (DAH) indicating that certain fuselage frame splices are subject to widespread fatigue damage (WFD). This proposed AD would require repetitive inspections of certain fuselage upper frames, side frames, fail-safe chords, inboard chords, frame webs, and stringers; an inspection for the presence of repairs in certain inspections zones and open tooling holes; and applicable on-condition actions. We are proposing this AD to address the unsafe condition on these products.

DATES: We must receive comments on this proposed AD by November 30, 2018.

ADDRESSES: You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- *Fax:* 202-493-2251.

- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE, Washington, DC 20590.

- *Hand Delivery:* Deliver to Mail address above between 9 a.m. and 5

p.m., Monday through Friday, except Federal holidays.

For service information identified in this NPRM, contact Boeing Commercial Airplanes, Attention: Contractual & Data Services (C&DS), 2600 Westminister Blvd., MC 110-SK57, Seal Beach, CA 90740-5600; telephone 562-797-1717; internet <https://www.myboeingfleet.com>. You may view this referenced service information at the FAA, Transport Standards Branch, 2200 South 216th St., Des Moines, WA. For information on the availability of this material at the FAA, call 206-231-3195. It is also available on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2018-0900.

Examining the AD Docket

You may examine the AD docket on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2018-0900; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this NPRM, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (phone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Lu Lu, Aerospace Engineer, Airframe Section, FAA, Seattle ACO Branch, 2200 South 216th St., Des Moines, WA 98198; phone and fax: 206-231-3525; email: *lu.lu@faa.gov*.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposal. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2018-0900; Product Identifier 2018-NM-101-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this NPRM. We will consider all comments received by the closing date and may amend this NPRM because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

Fatigue damage can occur locally, in small areas or structural design details, or globally, in widespread areas. Multiple-site damage is widespread damage that occurs in a large structural element such as a single rivet line of a lap splice joining two large skin panels. Widespread damage can also occur in multiple elements such as adjacent frames or stringers. Multiple-site damage and multiple-element damage cracks are typically too small initially to be reliably detected with normal inspection methods. Without intervention, these cracks will grow, and eventually compromise the structural integrity of the airplane. This condition is known as WFD. It is associated with general degradation of large areas of structure with similar structural details and stress levels. As an airplane ages, WFD will likely occur, and will certainly occur if the airplane is operated long enough without any intervention.

The FAA’s WFD final rule (75 FR 69746, November 15, 2010) became effective on January 14, 2011. The WFD rule requires certain actions to prevent structural failure due to WFD throughout the operational life of certain existing transport category airplanes and all of these airplanes that will be certificated in the future. For existing and future airplanes subject to the WFD rule, the rule requires that DAHs establish a limit of validity (LOV) of the engineering data that support the structural maintenance program. Operators affected by the WFD rule may not fly an airplane beyond its LOV, unless an extended LOV is approved.

The WFD rule (75 FR 69746, November 15, 2010) does not require identifying and developing maintenance actions if the DAHs can show that such actions are not necessary to prevent WFD before the airplane reaches the LOV. Many LOVs, however, do depend on accomplishment of future maintenance actions. As stated in the WFD rule, any maintenance actions necessary to reach the LOV will be mandated by airworthiness directives through separate rulemaking actions.

In the context of WFD, this action is necessary to enable DAHs to propose LOVs that allow operators the longest operational lives for their airplanes, and still ensure that WFD will not occur. This approach allows for an implementation strategy that provides flexibility to DAHs in determining the timing of service information development (with FAA approval), while providing operators with certainty regarding the LOV applicable to their airplanes.

We have received a report indicating that cracking is being found in the frame web, frame integral inboard chord, and fail-safe chord on multiple frame locations, below the passenger floor, on multiple Model 737–100, –200, –200C, –300, –400 and –500 series airplanes. In addition, the fuselage frame splices from station (STA) 380 to STA 520 and STA 727A to STA 907 between stringers S–13 and S–14, are subject to WFD. This condition, if not addressed, could result in the cracks growing large enough to sever frames. Continued operation with multiple adjacent severed frames or a combination of a severed frame adjacent to fuselage skin cracks in chem-milled pockets could result in a loss of structural integrity or uncontrolled decompression.

Related Service Information Under 1 CFR Part 51

We reviewed Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018. The service information describes procedures for repetitive inspections of certain fuselage upper frames, side frames, fail-safe chords, inboard chords, frame webs, and stringers; an inspection for the presence of repairs in certain inspections zones and open tooling holes; and applicable on-condition actions. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the ADDRESSES section.

FAA’s Determination

We are proposing this AD because we evaluated all the relevant information

and determined the unsafe condition described previously is likely to exist or develop in other products of the same type design.

Proposed AD Requirements

This proposed AD would require accomplishment of the actions identified as “RC” (required for compliance) in the Accomplishment Instructions of Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018, described previously, except for any differences identified as exceptions in the regulatory text of this proposed AD and except as discussed under “Differences Between this Proposed AD and the Service Information.”

For information on the procedures and compliance times, see this service information at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2018–0900.

Differences Between This Proposed AD and the Service Information

Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018, specifies that the additional inspections and applicable on-condition actions identified in Table 9, “Inspection of the Fuselage Frame Integral Inboard Chord and Web from STA 360 to STA 400, Right Side,” of Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018, must be done only for Group 3 airplanes. However, this proposed AD also requires that for Group 2 and Groups 4 through 9 airplanes identified in Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018, that have been modified to a cargo configuration, those additional inspections and applicable on-condition actions must also be done. We have coordinated this difference with Boeing.

Costs of Compliance

We estimate that this proposed AD affects 262 airplanes of U.S. registry. We estimate the following costs to comply with this proposed AD:

ESTIMATED COSTS FOR REQUIRED ACTIONS

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Inspections	Up to 243 work-hours × \$85 per hour = \$20,655 per inspection cycle.	None	Up to \$20,655 per inspection cycle.	Up to \$5,411,610 per inspection cycle.

We have received no definitive data that would enable us to provide cost estimates for the on-condition actions specified in this proposed AD.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

This proposed AD is issued in accordance with authority delegated by the Executive Director, Aircraft Certification Service, as authorized by FAA Order 8000.51C. In accordance with that order, issuance of ADs is normally a function of the Compliance and Airworthiness Division, but during this transition period, the Executive Director has delegated the authority to issue ADs applicable to transport category airplanes to the Director of the System Oversight Division.

Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. FAA amends § 39.13 by adding the following new airworthiness directive (AD):

The Boeing Company: Docket No. FAA–2018–0900; Product Identifier 2018–NM–101–AD.

(a) Comments Due Date

We must receive comments by November 30, 2018.

(b) Affected ADs

None.

(c) Applicability

This AD applies to The Boeing Company Model 737–100, –200, –200C, –300, –400, and –500 series airplanes, certificated in any category, as identified in Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018.

(d) Subject

Air Transport Association (ATA) of America Code 53, Fuselage.

(e) Unsafe Condition

This AD was prompted by reports of cracking in the frame web, frame integral inboard chord, and fail-safe chord on multiple airplanes in multiple locations below the passenger floor, in addition to an evaluation by the design approval holder (DAH) indicating that the fuselage frame splices from station (STA) 380 to STA 520 and STA 727A to STA 907 between stringers S–13 and S–14 are subject to widespread fatigue damage (WFD). We are issuing this AD to address cracks in these locations, which could grow large enough to sever frames. Continued operation with multiple adjacent severed frames or a combination of a severed frame adjacent to fuselage skin cracks in chem-milled pockets could result in a loss of structural integrity or uncontrolled decompression.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Actions for Group 1

For airplanes identified as Group 1 in Boeing Alert Service Bulletin 737–53A1360,

dated June 21, 2018: Within 120 days after the effective date of this AD, inspect the airplane and do all applicable on-condition actions using a method approved in accordance with the procedures specified in paragraph (k) of this AD.

(h) Inspection for Groups 2 Through 9

For airplanes identified as Groups 2 through 9 in Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018, except as specified in paragraph (i) of this AD: At the applicable times specified in paragraph 1.E., "Compliance," of Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018, do all applicable actions identified as "RC" (required for compliance) in, and in accordance with, the Accomplishment Instructions of Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018.

(i) Exceptions to Service Information Specifications

(1) For purposes of determining compliance with the requirements of this AD: Where Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018, uses the phrase "the original issue date of this service bulletin," this AD requires using "the effective date of this AD."

(2) Where Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018, specifies contacting Boeing for repair instructions: This AD requires repair and applicable on-condition actions using a method approved in accordance with the procedures specified in paragraph (k) of this AD.

(3) Where Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018, specifies contacting Boeing for alternative inspections: This AD requires alternative inspections using a method approved in accordance with the procedures specified in paragraph (k) of this AD.

(4) For airplanes identified as Group 2 and Groups 4 through 9 in Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018, that have been modified to a cargo configuration: In addition to the actions required by paragraph (h) of this AD, the actions specified in Table 9, "Inspection of the Fuselage Frame Integral Inboard Chord and Web from STA 360 to STA 400, Right Side," of Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018, must be done by doing all applicable actions identified as "RC" (required for compliance) in, and in accordance with, the Accomplishment Instructions of Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018, at the applicable compliance times specified in Table 9, "Inspection of the Fuselage Frame Integral Inboard Chord and Web from STA 360 to STA 400, Right Side," of Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018, except as specified in paragraphs (i)(1) and (i)(2) of this AD.

(j) Terminating Actions for Repetitive Inspections

(1) Accomplishment of a preventative modification specified in Part 7 of the Accomplishment Instructions of Boeing Alert Service Bulletin 737–53A1360, dated June 21, 2018, at a tooling hole location, terminates the repetitive inspections specified in Part 6 of the Accomplishment

Instructions of Boeing Alert Service Bulletin 737-53A1360, dated June 21, 2018, that are required by paragraph (h) of this AD, for that modified tooling hole location only.

(2) Accomplishment of an high frequency eddy current inspection specified in Part 9 of the Accomplishment Instructions of Boeing Alert Service Bulletin 737-53A1360, dated June 21, 2018, terminates the repetitive inspections specified in Part 2 of the Accomplishment Instructions of Boeing Alert Service Bulletin 737-53A1360, dated June 21, 2018, that are required by paragraph (h) of this AD, at the uppermost frame splice fastener location only.

(k) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Los Angeles ACO Branch, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the certification office, send it to the attention of the person identified in paragraph (l)(2) of this AD. Information may be emailed to: 9-ANM-LAACO-AMOC-Requests@faa.gov.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair, modification, or alteration required by this AD if it is approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has been authorized by the Manager, Los Angeles ACO Branch, FAA, to make those findings. To be approved, the repair method, modification deviation, or alteration deviation must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(4) Except as required by paragraph (i) of this AD: For service information that contains steps that are labeled as RC, the provisions of paragraphs (k)(4)(i) and (k)(4)(ii) of this AD apply.

(i) The steps labeled as RC, including substeps under an RC step and any figures identified in an RC step, must be done to comply with the AD. If a step or substep is labeled "RC Exempt," then the RC requirement is removed from that step or substep. An AMOC is required for any deviations to RC steps, including substeps and identified figures.

(ii) Steps not labeled as RC may be deviated from using accepted methods in accordance with the operator's maintenance or inspection program without obtaining approval of an AMOC, provided the RC steps, including substeps and identified figures, can still be done as specified, and the airplane can be put back in an airworthy condition.

(l) Related Information

(1) For more information about this AD, contact Lu Lu, Aerospace Engineer, Airframe Section, FAA, Seattle ACO Branch, 2200 South 216th St., Des Moines, WA 98198;

phone and fax: 206-231-3525; email: lu.lu@faa.gov.

(2) For information about AMOCs, contact George Garrido, Aerospace Engineer, Airframe Section, FAA, Los Angeles ACO Branch, 3960 Paramount Boulevard, Lakewood, CA 90712-4137; phone: 562-627-5232; fax: 562-627-5210; email: george.garrido@faa.gov.

(3) For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Contractual & Data Services (C&DS), 2600 Westminister Blvd., MC 110-SK57, Seal Beach, CA 90740-5600; telephone 562-797-1717; internet <https://www.myboeingfleet.com>. You may view this referenced service information at the FAA, Transport Standards Branch, 2200 South 216th St., Des Moines, WA. For information on the availability of this material at the FAA, call 206-231-3195.

Issued in Des Moines, Washington, on October 4, 2018.

Michael Kaszycki,

Acting Director, System Oversight Division, Aircraft Certification Service.

[FR Doc. 2018-22278 Filed 10-15-18; 8:45 am]

BILLING CODE 4910-13-P

LIBRARY OF CONGRESS

Copyright Office

37 CFR Part 201

[Docket No. 2018-8]

Noncommercial Use of Pre-1972 Sound Recordings That Are Not Being Commercially Exploited

AGENCY: U.S. Copyright Office, Library of Congress.

ACTION: Notice of inquiry.

SUMMARY: The U.S. Copyright Office is issuing a notice of inquiry regarding the Classics Protection and Access Act, title II of the recently enacted Orrin G. Hatch-Bob Goodlatte Music Modernization Act. In connection with the establishment of federal remedies for unauthorized uses of sound recordings fixed before February 15, 1972 ("Pre-1972 Sound Recordings"), Congress also established an exception for certain noncommercial uses of Pre-1972 Sound Recordings that are not being commercially exploited. To qualify for this exemption, a user must file a notice of noncommercial use after conducting a good faith, reasonable search to determine whether the Pre-1972 Sound Recording is being commercially exploited, and the rights owner of the sound recording must not object to the use within 90 days. To promulgate the regulations required by the new statute, the Office is soliciting comments regarding specific steps that

a user should take to demonstrate she has made a good faith, reasonable search. The Office also solicits comments regarding the filing requirements for the user to submit a notice of noncommercial use and for a rights owner to submit a notice objecting to such use.

DATES: Initial written comments must be received no later than 11:59 p.m. Eastern Time on November 15, 2018. Written reply comments must be received no later than 11:59 p.m. Eastern Time on November 30, 2018. Rather than reserving time for potential extensions of time to file comments, commenting parties should be aware that the Office has already established what it believes to be the most reasonable deadlines consistent with the statutory deadlines by which it must promulgate the regulations described in this notice of inquiry.

ADDRESSES: For reasons of government efficiency, the Copyright Office is using the *regulations.gov* system for the submission and posting of public comments in this proceeding. All comments are therefore to be submitted electronically through *regulations.gov*. Specific instructions for submitting comments are available on the Copyright Office's website at <https://www.copyright.gov/rulemaking/pre1972-soundrecordings-noncommercial/>. If electronic submission of comments is not feasible due to lack of access to a computer and/or the internet, please contact the Office using the contact information below for special instructions.

FOR FURTHER INFORMATION CONTACT:

Regan A. Smith, General Counsel and Associate Register of Copyrights, by email at regans@copyright.gov, Anna Chauvet, Assistant General Counsel, by email at achau@copyright.gov, or Jason E. Sloan, Assistant General Counsel, by email at jslo@copyright.gov. Each can be contacted by telephone by calling (202) 707-8350.

SUPPLEMENTARY INFORMATION:

I. Background

On October 11, 2018, the president signed into law the Orrin G. Hatch-Bob Goodlatte Music Modernization Act, H.R. 1551 ("MMA"). Title II of the MMA, the Classics Protection and Access Act, created chapter 14 of the copyright law, title 17, United States Code, which, among other things, extends remedies for copyright infringement to owners of sound recordings fixed before February 15, 1972 ("Pre-1972 Sound Recordings"). Under the provision, rights owners may be eligible to recover statutory damages