

1065, Revision 3, dated June 30, 2015: Do an external low frequency eddy current (LFEC) inspection for cracking of the skin at the critical fastener row of the repair doubler; and do all applicable corrective actions; in accordance the Accomplishment Instructions of Boeing Special Attention Service Bulletin 737-53-1065, Revision 3, dated June 30, 2015, except as required by paragraph (i)(3) of this AD. Do all applicable corrective actions before further flight. Repeat the LFEC inspection thereafter at the applicable intervals specified in Boeing Special Attention Service Bulletin 737-53-1065, Revision 3, dated June 30, 2015.

#### (m) Skin Panel Replacement

Except for Group 1 airplanes identified in Boeing Special Attention Service Bulletin 737-53-1065, Revision 3, dated June 30, 2015: At the later of the times specified in paragraphs (m)(1) and (m)(2) of this AD: Replace the applicable skin panels, and do all applicable related investigative and corrective actions, in accordance with the Accomplishment Instructions of Boeing Special Attention Service Bulletin 737-53-1065, Revision 3, dated June 30, 2015. Do all applicable related investigative and corrective actions before further flight. Doing the skin panel replacement required by this paragraph terminates the inspection requirements of paragraph (h) of this AD for that skin panel only, provided the skin panel was replaced with a production skin panel after 59,000 total flight cycles.

(1) Before 60,000 total flight cycles, but not at or before 59,000 total flight cycles.

(2) Within 6,000 flight cycles after the effective date of this AD, but not at or before 59,000 total flight cycles.

#### (n) Credit for Previous Actions

This paragraph provides credit for the actions required by paragraph (h) of this AD, if those actions were performed before the effective date of this AD using Boeing Special Attention Service Bulletin 737-53-1065, Revision 2, dated April 19, 2001, which was incorporated by reference in AD 2005-13-30.

#### (o) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Seattle Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in paragraph (p)(1) of this AD. Information may be emailed to: [9-ANM-Seattle-ACO-AMOC-Requests@faa.gov](mailto:9-ANM-Seattle-ACO-AMOC-Requests@faa.gov).

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair, modification, or alteration required by this AD if it is approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has

been authorized by the Manager, Seattle ACO, to make those findings. To be approved, the repair method, modification deviation, or alteration deviation must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(4) AMOCs approved previously for AD 2005-13-30, are approved as AMOCs for the corresponding provisions of paragraph (h) of this AD.

#### (p) Related Information

(1) For more information about this AD, contact Wade Sullivan, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle ACO, 1601 Lind Avenue SW., Renton, WA 98057-3356; phone: 425-917-6430; fax: 425-917-6590; email: [wade.sullivan@faa.gov](mailto:wade.sullivan@faa.gov).

(2) For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P. O. Box 3707, MC 2H-65, Seattle, WA 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>. You may view this referenced service information at the FAA, 1601 Lind Avenue SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221.

Issued in Renton, Washington, on May 4, 2016.

**Michael Kaszycki,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2016-11095 Filed 5-10-16; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2016-6428; Directorate Identifier 2015-NM-119-AD]

RIN 2120-AA64

#### Airworthiness Directives; The Boeing Company Airplanes

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** We propose to adopt a new airworthiness directive (AD) for certain The Boeing Company Model 787-8 airplanes. This proposed AD was prompted by reports indicating that certain wing side-of-body stringer fittings have been installed with faying surface mismatch beyond the allowed machining tolerance. This proposed AD would require inspection of certain stringer fittings for faying surface mismatch common to the side-of-body rib chord, replacement if necessary, and replacement of the clearance fit fasteners common to the side-of-body

fittings and upper side-of-body rib chord with tapered sleeve bolts. We are proposing this AD to prevent an unacceptable reduction of the fatigue life in the upper side-of-body rib chord. Associated fatigue cracks can reduce the structural capability to a point where it cannot sustain limit load, which could adversely affect the structural integrity of the airplane.

**DATES:** We must receive comments on this proposed AD by June 27, 2016.

**ADDRESSES:** You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- *Fax:* 202-493-2251.

- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

- *Hand Delivery:* Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this NPRM, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, WA 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-6428.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-6428; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (phone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

**FOR FURTHER INFORMATION CONTACT:** Allen Rauschendorfer, Aerospace Engineer, Airframe Branch, ANM-120S,

FAA, Seattle Aircraft Certification Office (ACO), 1601 Lind Avenue SW., Renton, WA 98057-3356; phone: 425-917-6487; fax: 425-917-6590; email: allen.rauschenborfer@faa.gov.

**SUPPLEMENTARY INFORMATION:**

**Comments Invited**

We invite you to send any written relevant data, views, or arguments about this proposal. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2016-6428; Directorate Identifier 2015-NM-119-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

**Discussion**

We have received reports indicating that certain wing side-of-body stringer fittings have been installed with faying surface mismatch beyond the allowed machining tolerance. The fittings are assembled to the mating side-of-body rib chord. The faying surface mismatch

produces a gouge in the mating surface which reduces the fatigue life, and could grow into a widespread fatigue condition on the upper side-of-body rib chord. We are proposing this AD to prevent an unacceptable reduction of the fatigue life in the upper side-of-body rib chord. Associated fatigue cracks can reduce the structural capability to a point where it cannot sustain limit load, which could adversely affect the structural integrity of the airplane.

**Related Service Information Under 1 CFR Part 51**

We reviewed Boeing Alert Service Bulletin B787-81205-SB570018-00, Issue 001, dated July 1, 2015. The service information describes procedures for inspection of the left and right hand side stringer 1 fittings for faying surface mismatch common to the side-of-body rib chord. If faying surface mismatch is found, instructions are also given to replace the stringer 1 fitting, and removal and replacement of the clearance fit fasteners common to the side-of-body fittings and upper side-of-body rib chord with tapered sleeve bolts from stringer 5 to stringer 11. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.

**FAA's Determination**

We are proposing this AD because we evaluated all the relevant information

and determined the unsafe condition described previously is likely to exist or develop in other products of the same type design.

**Proposed AD Requirements**

This proposed AD would require accomplishing the actions specified in the service information described previously, except as described in "Differences Between this Proposed AD and the Service Information."

**Differences Between This Proposed AD and the Service Information**

Boeing Alert Service Bulletin B787-81205-SB570018-00, Issue 001, dated July 1, 2015, specifies to contact the manufacturer for instructions on how to repair certain conditions, but this proposed AD would require repairing those conditions in one of the following ways:

- In accordance with a method that we approve; or
- Using data that meet the certification basis of the airplane, and that have been approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) whom we have authorized to make those findings.

**Costs of Compliance**

We estimate that this proposed AD affects 5 airplanes of U.S. registry.

We estimate the following costs to comply with this proposed AD:

**ESTIMATED COSTS**

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Modification and inspection.	144 work-hours × \$85 per hour = \$12,240 .....	\$100,079	\$112,319	\$561,595

We estimate the following costs to do any necessary corrective action for fretting damage or cutter mismatch that

would be required based on the results of the proposed inspection. We have no way of determining the number of

aircraft that might need these corrective actions:

**ON-CONDITION COSTS**

Action	Labor cost	Parts cost	Cost per product
Repair for fretting damage or cutter mismatch.	9 work-hours × \$85 per hour = \$765 .....	\$0	\$765

We have received no definitive data that would enable us to provide cost estimates for the crack repair specified in this proposed AD.

**Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA's authority to issue

rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations

for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

### Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

(1) Is not a “significant regulatory action” under Executive Order 12866,

(2) Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),

(3) Will not affect intrastate aviation in Alaska, and

(4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

### The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

**The Boeing Company:** Docket No. FAA–2016–6428; Directorate Identifier 2015–NM–119–AD.

#### (a) Comments Due Date

We must receive comments by June 27, 2016.

#### (b) Affected ADs

None.

#### (c) Applicability

This AD applies to The Boeing Company Model 787–8 airplanes, certificated in any category, as identified in Boeing Alert Service Bulletin B787–81205–SB570018–00, Issue 001, dated July 1, 2015.

#### (d) Subject

Air Transport Association (ATA) of America Code 57, Wings.

#### (e) Unsafe Condition

This AD was prompted by reports indicating that certain wing side-of-body stringer fittings have been installed with faying surface mismatch beyond the allowed machining tolerance. We are issuing this AD to prevent an unacceptable reduction of the fatigue life in the upper side-of-body rib chord. Associated fatigue cracks can reduce the structural capability to a point where it cannot sustain limit load, which could adversely affect the structural integrity of the airplane.

#### (f) Compliance

Comply with this AD within the compliance times specified, unless already done.

#### (g) Inspection

Before the accumulation of 18,000 total flight cycles, or within 13 years after the effective date of this AD, whichever occurs first, do the inspections specified in paragraphs (g)(1), (g)(2), and (g)(3) of this AD, and all applicable corrective actions, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin B787–81205–SB570018–00, Issue 001, dated July 1, 2015, except as required by paragraph (i) of this AD. Do all applicable corrective actions before further flight.

(1) Do a detailed inspection for fretting damage of the faying surface of the aluminum T-chord.

(2) Do an eddy current inspection for cracking of the fastener holes.

(3) Do a detailed inspection for a machine mismatch condition of the stringer 1 fitting faying surface.

#### (h) Modifications

Concurrently with accomplishment of the requirements of paragraph (g) of this AD: Modify the stringer fitting fasteners, and do an eddy current inspection for cracking of the fastener holes, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin B787–81205–SB570018–00, Issue 001, dated July 1, 2015. If any crack is found, before further flight, repair using a method approved in accordance with the procedures specified in paragraph (j) of this AD.

#### (i) Exception to Service Information Specifications

Where Boeing Alert Service Bulletin B787–81205–SB570018–00, Issue 001, dated July 1, 2015, specifies to contact Boeing for repair of cracking: Before further flight, repair the cracking using a method approved in accordance with the procedures specified in paragraph (j) of this AD.

#### (j) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Seattle Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in paragraph (k)(1) of this AD. Information may be emailed to: [9-ANM-Seattle-ACO-AMOC-Requests@faa.gov](mailto:9-ANM-Seattle-ACO-AMOC-Requests@faa.gov).

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair, modification, or alteration required by this AD if it is approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has been authorized by the Manager, Seattle ACO, to make those findings. To be approved, the repair method, modification deviation, or alteration deviation must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(4) Except as required by paragraph (i) of this AD: For service information that contains steps that are labeled as Required for Compliance (RC), the provisions of paragraphs (j)(4)(i) and (j)(4)(ii) apply.

(i) The steps labeled as RC, including substeps under an RC step and any figures identified in an RC step, must be done to comply with the AD. An AMOC is required for any deviations to RC steps, including substeps and identified figures.

(ii) Steps not labeled as RC may be deviated from using accepted methods in accordance with the operator's maintenance or inspection program without obtaining approval of an AMOC, provided the RC steps, including substeps and identified figures, can still be done as specified, and the airplane can be put back in an airworthy condition.

#### (k) Related Information

(1) For more information about this AD, contact Allen Rauschendorfer, Aerospace Engineer, Airframe Branch, ANM–120S, FAA, Seattle Aircraft Certification Office (ACO), 1601 Lind Avenue SW., Renton, WA 98057–3356; phone: 425–917–6487; fax: 425–917–6590; email: [allen.rauschendorfer@faa.gov](mailto:allen.rauschendorfer@faa.gov).

(2) For service information identified in this AD, Boeing Commercial Airplanes, Attention: Data & Services Management, P. O. Box 3707, MC 2H–65, Seattle, WA 98124–2207; telephone 206–544–5000, extension 1; fax 206–766–5680; Internet <https://www.myboeingfleet.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425–227–1221.

Issued in Renton, Washington, on May 3, 2016.

**Michael Kaszycki,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2016-10915 Filed 5-10-16; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2016-6431; Directorate Identifier 2015-NM-182-AD]

RIN 2120-AA64

#### Airworthiness Directives; Airbus Airplanes

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** We propose to adopt a new airworthiness directive (AD) for certain Airbus Model A318-112 airplanes, A319-111, -112, -115, -132, and -133 airplanes, A320-214, -232, and -233 airplanes, and A321-211, -212, -213, -231, and -232 airplanes. This proposed AD was prompted by a quality control review on the final assembly line, which determined that aluminum alloy with inadequate heat treatment had been delivered and used on several structural parts. This proposed AD would require a one-time eddy current conductivity measurement of certain cabin, cargo compartment, and frame structural parts to determine if aluminum alloy with inadequate heat treatment was used, and replacement if necessary. We are proposing this AD to detect and replace structural parts made of aluminum alloy with inadequate heat treatment. This condition could result in reduced structural integrity of the airplane.

**DATES:** We must receive comments on this proposed AD by June 27, 2016.

**ADDRESSES:** You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- **Federal eRulemaking Portal:** Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- **Fax:** 202-493-2251.

- **Mail:** U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

- **Hand Delivery:** Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this NPRM, contact Airbus, Airworthiness Office—EIAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone: +33 5 61 93 36 96; fax: +33 5 61 93 44 51; email: [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com); Internet: <http://www.airbus.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-6431; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

#### FOR FURTHER INFORMATION CONTACT:

Sanjay Ralhan, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone: 425-227-1405; fax: 425-227-1149.

#### SUPPLEMENTARY INFORMATION:

##### Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include “Docket No. FAA-2016-6431; Directorate Identifier 2015-NM-182-AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD based on those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

#### Discussion

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Union, has issued EASA Airworthiness

Directive 2015-0219, dated November 3, 2015 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition for certain Airbus Model A318-112 airplanes, A319-111, -112, -115, -132, and -133 airplanes, A320-214, -232, and -233 airplanes, and A321-211, -212, -213, -231, and -232 airplanes. The MCAI states:

Following an Airbus quality control review on the final assembly line, it was discovered that aluminum alloy with inadequate heat treatment were delivered by a supplier for several structural parts. The results of the investigations highlighted that 1% of the stock could be impacted by this wrong material.

Structural investigations demonstrated the capability to sustain the static limits loads, and sufficient fatigue life up to a certain inspection threshold.

This condition, if not detected and corrected, could reduce the aeroplane structural integrity following fatigue load.

To address this potential unsafe condition, Airbus issued Service Bulletin (SB) A320-53-1292, SB A320-53-1293, and SB A320-53-1294 to provide inspection instructions.

For the reasons described above, this [EASA] AD requires a one-time Special Detailed Inspection (SDI) [*i.e.*, eddy current conductivity measurement] of certain cabin, cargo compartment and frame parts [for material identification] and, depending on findings, replacement with serviceable parts.

You may examine the MCAI in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-6431.

#### Related Service Information Under 14 CFR Part 51

Airbus has issued the following service information:

- Airbus Service Bulletin A320-53-1292, dated July 23, 2015; including Appendices 01 and 02, dated July 23, 2015.

- Airbus Service Bulletin A320-53-1293, dated July 30, 2015; including Appendices 01 and 02, dated July 30, 2015.

- Airbus Service Bulletin A320-53-1294, dated July 23, 2015; including Appendices 01 and 02, dated July 23, 2015.

The service information describes procedures for a one-time eddy current conductivity measurement of certain cabin, cargo compartment, and frame structural parts to determine if aluminum alloy with inadequate heat treatment was used, and replacement of any affected part with a serviceable part. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.