

Worth, TX 76177; telephone (817) 222-5110; email 9-ASW-FTW-AMOC-Requests@faa.gov.

(2) For operations conducted under a 14 CFR part 119 operating certificate or under 14 CFR part 91, subpart K, we suggest that you notify your principal inspector, or lacking a principal inspector, the manager of the local flight standards district office or certificate holding district office, before operating any aircraft complying with this AD through an AMOC.

(g) Additional Information

(1) Airbus Helicopters Alert Service Bulletin ASB No. SA341/SA342-05.40, Revision 0, dated April 28, 2014, which is not incorporated by reference, contains additional information about the subject of this final rule. For Airbus Helicopters service information identified in this final rule, contact Airbus Helicopters, 2701 N. Forum Drive, Grand Prairie, TX 75052; telephone (972) 641-0000 or (800) 232-0323; fax (972) 641-3775; or at <http://www.airbushelicopters.com/techpub>.

You may review a copy of the service information at the FAA, Office of the Regional Counsel, Southwest Region, 10101 Hillwood Pkwy, Room 6N-321, Fort Worth, TX 76177.

(2) The subject of this AD is addressed in European Aviation Safety Agency (EASA) AD No. 2014-0216, dated September 24, 2014. You may view the EASA AD on the Internet at <http://www.regulations.gov> in Docket No. FAA-2015-5914.

(h) Subject

Joint Aircraft Service Component (JASC) Code: 6700, Main Rotor.

(i) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Gazelle Inspection—Check 65.12.607, “Main Rotor Head: Torsion Tie-Back Check (Post MOD 076171),” dated August 2008, of the Eurocopter Gazelle Helicopter Maintenance Manual, Tome 1.

(ii) Reserved.

(3) For Eurocopter service information identified in this final rule, contact Airbus Helicopters, 2701 N. Forum Drive, Grand Prairie, TX 75052; telephone (972) 641-0000 or (800) 232-0323; fax (972) 641-3775; or at <http://www.airbushelicopters.com/techpub>.

(4) You may view this service information at FAA, Office of the Regional Counsel, Southwest Region, 10101 Hillwood Pkwy, Room 6N-321, Fort Worth, TX 76177. For information on the availability of this material at the FAA, call (817) 222-5110.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Fort Worth, Texas, on March 31, 2016.

James A. Grigg,

Acting Manager, Rotorcraft Directorate, Aircraft Certification Service.

[FR Doc. 2016-07979 Filed 4-12-16; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2015-3147; Directorate Identifier 2014-NM-094-AD; Amendment 39-18479; AD 2016-08-03]

RIN 2120-AA64

Airworthiness Directives; The Boeing Company Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for all The Boeing Company Model 777-200, -200LR, -300, and -300ER series airplanes. This AD was prompted by reports of fractured forward attach fittings of the inboard flap outboard aft flap track. The fractured fittings were determined to be the result of corrosion pits forming on the inside diameter of the fittings. This AD requires an inspection for the affected part number and serial number of the main flap; various additional repetitive inspections of the fitting, if necessary; and replacement of the fitting or nested bushing installation, if necessary, which would terminate the inspections. This AD also provides an optional terminating action for the repetitive inspections. We are issuing this AD to detect and correct fracture of the fitting, which could result in the loss of the inboard aft flap and could lead to a punctured fuselage, causing injury to the flightcrew and passengers, and damage to the airplane.

DATES: This AD is effective May 18, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of May 18, 2016.

ADDRESSES: For service information identified in this final rule, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P. O. Box 3707, MC 2H-65, Seattle, WA 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>. You may view this referenced service

information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-3147.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-3147; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Eric Lin, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone: 425-917-6412; fax: 425-917-6590; email: Eric.Lin@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to all The Boeing Company Model 777-200, -200LR, -300, and -300ER series airplanes. The NPRM published in the **Federal Register** on August 25, 2015 (80 FR 51491) (“the NPRM”). The NPRM was prompted by reports of fractured forward attach fittings of the inboard flap outboard aft flap track. The fractured fittings were determined to be the result of corrosion pits forming on the inside diameter of the fittings. The NPRM proposed to require an inspection for the affected part number and serial number of the main flap; various additional repetitive inspections of the fitting, if necessary; and replacement of the fitting or nested bushing installation, if necessary, which would terminate the inspections. The proposed AD also provided an optional terminating action for the repetitive inspections. We are issuing this AD to detect and correct fracture of the fitting, which could result in the loss of the inboard aft flap and could lead to a punctured fuselage, causing injury to

the flightcrew and passengers, and damage to the airplane.

Comments

We gave the public the opportunity to participate in developing this AD. The following presents the comments received on the NPRM and the FAA's response to each comment. Benjamin Kerensa commented that he supports the NPRM.

Request To Change Terminology

Boeing requested that we change the phrase "inboard flap of the main flap" to "main flap of the inboard flap assembly" in paragraphs (g) and (h) of the proposed AD. Boeing stated that this would clarify the inspection location.

We agree with the commenter's request for the reason provided by the commenter. We have revised paragraphs (g) and (h) of this AD accordingly.

Request To Clarify Requirements of a Certain Terminating Action

Boeing requested that we revise the terminating action specified in paragraph (k)(1) of the proposed AD. Boeing recommended that at the end of paragraph (k)(1) of the proposed AD, we add the following language, "Prior to accomplishing this option, the inspections of paragraph (h) must be accomplished and no cracking must have been found during that inspection." Boeing commented that the bushing replacement is only a terminating action if the fitting has been verified to be crack free.

We agree to clarify the terminating action. As specified in Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, if cracking is found during an inspection that is part of the terminating action, a fitting replacement must be done instead of the terminating action. In addition, Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, specifies that if corrosion cannot be removed during the terminating action, a fitting replacement must be done instead of the terminating action.

Therefore, we have revised paragraph (k)(1) of this AD to specify that the terminating action is acceptable provided no cracking is found during any inspection specified in Part 4 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, and all corrosion has been removed.

We have also clarified the terminating actions specified in paragraphs (h), (i), (j), (k)(1), (k)(2), and (k)(3) of this AD to specify the terminating actions apply

only to that fitting on which the terminating action is done.

In addition, we have clarified the actions specified in paragraphs (i)(3)(i), (i)(3)(ii), and (i)(3)(iii) of this AD by specifying the actions include doing all applicable related investigative actions. Also, we have clarified the actions specified in paragraphs (j)(1) and (j)(2) of this AD by specifying the actions include doing all applicable related investigative and corrective actions.

Request To Clarify Terminating Action Is an Option to the Inspections

Boeing and American Airlines (AA) requested that we clarify that the terminating action can be done instead of the inspections. AA commented to revise the terminating action paragraph to state the following:

If any inboard flap of the main flap having an affected part number and serial number is found during the inspection required by paragraph (g) of this AD: Except as provided by paragraph (l) of this AD, at the applicable time specified in paragraph 1.E., "Compliance," of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, do the terminating action in paragraph (k)(1), (k)(2), or (k)(3) of this AD or do the inspections specified in paragraph (h)(1) or (h)(2) of this AD, in accordance with the Accomplishment Instructions of Boeing Special attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014. Repeat the inspections thereafter at the applicable times specified in paragraph 1.E., "Compliance," of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, until a terminating action in paragraph (k)(1), (k)(2), or (k)(3) of this AD is done.

AA commented that this change would align the proposed rule with paragraph 1.E., "Compliance," table 1 and table 2, of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014. AA also stated that this would clearly indicate that the terminating action would preclude having to accomplish first the initial inspection requirement in paragraph (h) of the proposed AD. Boeing requested that we add a sentence to the end of paragraph (h) to specify that the inspections are not required for any affected inboard flap assembly where one of the terminating actions specified in paragraph (k)(1), (k)(2), or (k)(3) of the proposed AD have been done.

We agree with the commenters' request. We have revised paragraph (h) of this AD to clarify that the terminating actions may preclude the initial inspections because rework or replacement of the parts prevent the unsafe condition.

Request To Clarify the Applicability

Delta Airlines (DAL) requested that we clarify whether paragraph (c) of the proposed AD is intended for all Model 777-200, -200LR, -300, and -300ER series airplanes, or all Model 777-200, -200LR, -300 and -300ER series airplanes, identified as Groups 1, 2, and 4 airplanes in Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014.

DAL stated that paragraph (c) of the proposed AD does not contain Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, to identify the airplane's applicability for the proposed AD. DAL commented that in the proposed rule under "Differences Between This Proposed AD and the Service Information," the airplane's applicability is stated. DAL also stated that the same paragraph states that the proposed AD is applicable only to Groups 1, 2, and 4 airplanes (Model 777-200, -200LR, -300 and -300ER series airplanes) specified in Boeing Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014. DAL commented that a review of the "Compliance" paragraph in the service information indicates that the FAA will possibly release an AD for airplanes in Groups 1, 2, and 4.

We agree with the commenter's request to clarify the applicability of this AD. This AD applies to all Model 777-200, -200LR, -300, and -300ER series airplanes as specified in paragraph (c) of this AD. The description of Groups 1, 2, and 4 airplanes in section 1.A., "Effectivity," of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, specifies Model 777-200, -200LR, -300, and -300ER series airplanes. Additionally, as noted in the service information, the table of variable numbers given in section 1.A., "Effectivity," only reflects up to line number 1322, and affected airplanes after line number 1322 will be added to the profile page of MyBoeingFleet.com. Therefore, this AD applies to all Model 777-200, -200LR, -300, and -300ER series airplanes. We have not changed this AD in this regard.

Conclusion

We reviewed the relevant data, considered the comments received, and determined that air safety and the public interest require adopting this AD with the changes described previously and minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

We also determined that these changes will not increase the economic burden on any operator or increase the scope of this AD.

Related Service Information Under 1 CFR Part 51

We reviewed Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014. The service information describes procedures for an inspection for the affected part number and serial number of the main flap; various additional repetitive inspections of the fitting, if necessary; and replacement of the fitting or nested bushing installation, if necessary, which would terminate the

inspections. The service information also describes an optional terminating action for the repetitive inspections. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the ADDRESSES section.

Costs of Compliance

We estimate that this AD affects 148 airplanes of U.S. registry.

We estimate the following costs to comply with this AD:

ESTIMATED COSTS

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Inspection to determine the part number.	3 work-hours × \$85 per hour = \$255	\$0	\$255	\$37,740.
Additional Inspections	Up to 7 work-hours × \$85 per hour = \$595, per cycle.	0	Up to \$595, per cycle ...	Up to \$88,060, per cycle.

We estimate the following costs to do any necessary replacements that will be required based on the results of the

inspection. The nested bushing installation of the attach fitting and the fitting replacement are also optional

terminating actions. We have no way of determining the number of aircraft on which these actions might be done.

ON-CONDITION COSTS

Action	Labor cost	Parts cost	Cost per product
Nested bushing installation of the attach fitting	40 work-hours × \$85 per hour = \$3,400	\$45	\$3,445.
Fitting replacement	73 work-hours × \$85 per hour = \$6,205	7,400	13,605.

According to the manufacturer, all of the costs of this AD may be covered under warranty, thereby reducing the cost impact on affected individuals. We do not control warranty coverage for affected individuals. As a result, we have included all costs in our cost estimate.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: “General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on

products identified in this rulemaking action.

Regulatory Findings

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a “significant regulatory action” under Executive Order 12866,
- (2) Is not a “significant rule” under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2016-08-03 The Boeing Company:
Amendment 39-18479; Docket No. FAA-2015-3147; Directorate Identifier 2014-NM-094-AD.

(a) Effective Date

This AD is effective May 18, 2016.

(b) Affected ADs

None.

(c) Applicability

This AD applies to all The Boeing Company Model 777-200, -200LR, -300, and -300ER series airplanes, certificated in any category.

(d) Subject

Air Transport Association (ATA) of America Code 57, Wings.

(e) Unsafe Condition

This AD was prompted by reports of fractured forward attach fittings of the inboard flap outboard aft flap track. The fractured fittings were determined to be the result of corrosion pits forming on the inside diameter of the fittings. We are issuing this AD to detect and correct fracture of the fitting, which could result in the loss of the inboard aft flap and could lead to a punctured fuselage, causing injury to the flightcrew and passengers, and damage to the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Inspection To Determine the Part Number

At the applicable time specified in paragraph 1.E., "Compliance," of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, except as provided by paragraph (l) of this AD: Do an inspection of the main flap of the inboard flap assembly for affected part and serial numbers, in accordance with the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014. A review of airplane maintenance records is acceptable in lieu of this inspection if the part number and serial number of the inboard flap can be conclusively determined from that review.

(h) Additional Inspections

If any main flap of the inboard flap assembly having an affected part number and serial number is found during the inspection required by paragraph (g) of this AD: Except as provided by paragraph (l) of this AD, at the applicable time specified in paragraph 1.E., "Compliance," of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, do the terminating action specified in paragraph (k)(1), (k)(2), or (k)(3) of this AD, or do the inspections specified in paragraph (h)(1) or (h)(2) of this AD, in accordance with the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014. Repeat the inspections thereafter at the applicable times specified in paragraph 1.E., "Compliance," of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, until a terminating action in paragraph (k)(1), (k)(2), or (k)(3) of this AD is done. Accomplishing a terminating action specified in paragraph (k)(1), (k)(2), or (k)(3) of this AD terminates the inspections required by this paragraph for that fitting only.

(1) At the forward attach fitting of the aft flap track of the inboard flap: Do a detailed inspection for cracking and bushing migration, and a high frequency eddy current inspection for cracking, in accordance with Part 2 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin

777-57-0094, Revision 1, dated November 5, 2014.

(2) At the forward attach fitting of the aft flap track of the inboard flap: Do a detailed inspection for cracking and bushing migration, and an ultrasound inspection for cracking, in accordance with Part 3 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014.

(i) Corrective Action for Bushing Migration

If any bushing migration but no cracking is found during any inspection required by paragraph (h) of this AD: At the applicable times specified in paragraph 1.E., "Compliance," of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, do the actions specified in paragraphs (i)(1) through (i)(3) of this AD. Accomplishment of a terminating action specified in paragraph (i)(3) or (k) of this AD terminates the actions required by this paragraph for that fitting only.

(1) Apply corrosion inhibiting compound BMS 3-23, Type II, around the bushing flanges on each side of the fitting, in accordance with the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014. Re-apply the corrosion inhibiting compound at the time specified in paragraph 1.E., "Compliance," of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014.

(2) Repeat the inspections specified in paragraph (h)(1) or (h)(2) of this AD, except inspect for cracking only.

(3) Do a terminating action specified in paragraph (i)(3)(i), (i)(3)(ii), or (i)(3)(iii) of this AD.

(i) Install a nested bushing to the forward attach fitting of the aft flap track of the inboard flap, including doing all applicable related investigative actions, in accordance with Part 4 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014.

(ii) Replace the forward attach fitting of the aft flap track of the inboard flap with an aluminum fitting, including doing all applicable related investigative actions, in accordance with Part 5 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014.

(iii) Replace the forward attach fitting of the aft flap track of the inboard flap with a titanium fitting, including doing all applicable related investigative actions, in accordance with Part 6 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014.

(j) Corrective Actions for Cracking

If any cracking is found during any inspection required by paragraph (h) or (i)(3) of this AD: At the applicable time specified in paragraph 1.E., "Compliance," of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, do a terminating action specified in paragraph (j)(1) or (j)(2) of this AD.

Replacement of the forward attach fitting as specified in paragraph (j)(1) or (j)(2) of this AD terminates the actions in this AD for that fitting only.

(1) Replace the forward attach fitting of the aft flap track of the inboard flap with an aluminum fitting, including doing all applicable related investigative and corrective actions, in accordance with Part 5 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014.

(2) Replace the forward attach fitting of the aft flap track of the inboard flap with a titanium fitting, including doing all applicable related investigative and corrective actions, in accordance with Part 6 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014.

(k) Optional Terminating Actions

(1) Installation of the nested bushing to the forward attach fitting of the aft flap track of the inboard flap, in accordance with Part 4 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, terminates the requirements of this AD for that fitting only, provided no cracking is found during any inspection specified in Part 4 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, and all corrosion has been removed.

(2) Replacement of the forward attach fitting of the aft flap track of the inboard flap with an aluminum fitting, in accordance with Part 5 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, terminates the requirements of this AD for that fitting only.

(3) Replacement of the forward attach fitting of the aft flap track of the inboard flap with a titanium fitting, in accordance with Part 6 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, terminates the requirements of this AD for that fitting only.

(l) Exception to the Service Information

Where Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014, specifies a compliance time "after the original issue date of this service bulletin," this AD requires compliance within the specified compliance time after the effective date of this AD.

(m) Credit for Previous Actions

(1) This paragraph provides credit for the actions specified in paragraphs (h)(1) and (h)(2) of this AD, if those actions were performed before the effective date of this AD using Boeing Special Attention Service Bulletin 777-57-0094, dated January 29, 2014, which is not incorporated by reference in this AD.

(2) This paragraph provides credit for the actions specified in paragraph (h)(1) of this AD, if those actions were performed before the effective date of this AD using Boeing Multi Operator Message MOM-MOM-13-

0137-01B, dated February 21, 2013, which is not incorporated by reference in this AD.

(n) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Seattle Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in paragraph (o)(1) of this AD. Information may be emailed to: 9-ANM-Seattle-ACO-AMOC-Requests@faa.gov.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair, modification, or alteration required by this AD if it is approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has been authorized by the Manager, Seattle ACO, to make those findings. To be approved, the repair method, modification deviation, or alteration deviation must meet the certification basis of the airplane and the approval must specifically refer to this AD.

(4) For service information that contains steps that are labeled as Required for Compliance (RC), the provisions of paragraphs (n)(4)(i) and (n)(4)(ii) apply.

(i) The steps labeled as RC, including substeps under an RC step and any figures identified in an RC step, must be done to comply with the AD. An AMOC is required for any deviations to RC steps, including substeps and identified figures.

(ii) Steps not labeled as RC may be deviated from using accepted methods in accordance with the operator's maintenance or inspection program without obtaining approval of an AMOC, provided the RC steps, including substeps and identified figures, can still be done as specified, and the airplane can be put back in an airworthy condition.

(o) Related Information

(1) For more information about this AD, contact Eric Lin, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle ACO, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone: 425-917-6412; fax: 425-917-6590; email: Eric.Lin@faa.gov.

(2) Service information identified in this AD that is not incorporated by reference is available at the addresses specified in paragraphs (p)(3) and (p)(4) of this AD.

(p) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Boeing Special Attention Service Bulletin 777-57-0094, Revision 1, dated November 5, 2014.

(ii) Reserved.

(3) For Boeing service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P. O. Box 3707, MC 2H-65, Seattle, WA 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>.

(4) You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on March 31, 2016.

Victor Wicklund,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2016-08359 Filed 4-12-16; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2015-2959; Directorate Identifier 2015-NM-008-AD; Amendment 39-18470; AD 2016-07-25]

RIN 2120-AA64

Airworthiness Directives; The Boeing Company Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for certain The Boeing Company Model 787-8 airplanes. This AD was prompted by reports indicating that the ram air turbine (RAT) assembly may fail to operate if deployed at low airspeeds. This AD requires replacing either the RAT pump and control module assembly or the entire RAT assembly. We are issuing this AD to prevent failure of the RAT assembly to operate at low air speeds. The volume fuse on the RAT assembly may be activated in-flight before the RAT is deployed. This may lead to improper pump hydraulic pressure offloading when the RAT is needed. Failure of the RAT to operate in an all engine out event would result in loss of control of the airplane.

DATES: This AD is effective May 18, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of May 18, 2016.

ADDRESSES: For service information identified in this final rule, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, WA 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-2959.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-2959; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Sean J. Schauer, Aerospace Engineer, Systems and Equipment Branch, ANM-130S, FAA, Seattle Aircraft Certification Office (ACO), 1601 Lind Avenue SW., Renton, WA 98057-3356; phone: 425-917-6479; fax: 425-917-6590; email: sean.schauer@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to certain The Boeing Company Model 787-8 airplanes. The NPRM published in the **Federal Register** on July 24, 2015 (80 FR 43972) ("the NPRM"). The NPRM was prompted by reports indicating that the RAT assembly may fail to operate if deployed at low airspeeds. The NPRM proposed to require replacing either the RAT pump and control module assembly or