

TABLE 1 TO PARAGRAPH (i) OF THIS AD—REPLACEMENT COMPLIANCE TIMES—Continued

Year of manufacture	Compliance time
2009	Before exceeding 10 years since date of manufacture of the passenger chemical oxygen generator.

(j) Definition of Serviceable

For the purpose of this AD, a serviceable unit is a passenger chemical oxygen generator having P/N 117042-XX with a manufacturing date not older than 10 years, or any other approved part number, provided that the generator has not exceeded the life limit established for that generator by the manufacturer.

(k) Reporting

At the applicable time specified in paragraph (k)(1) or (k)(2) of this AD, submit a report of the findings (both positive and negative) of the inspection required by paragraph (g) of this AD, in accordance with paragraph 7., "Reporting," of Airbus AOT A35N006-14, dated December 10, 2014, including Appendix 01. The report must include the information specified in Appendix 1 of Airbus AOT A35N006-14, dated December 10, 2014.

(1) If the inspection was done on or after the effective date of this AD: Submit the report within 30 days after the inspection.

(2) If the inspection was done before the effective date of this AD: Submit the report within 30 days after the effective date of this AD.

(l) Parts Installation Limitation

As of the effective date of this AD, no person may install a passenger chemical oxygen generator, unless it is determined, prior to installation, that the oxygen generator is a serviceable unit as specified in paragraph (j) of this AD.

(m) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs)*: The Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Sanjay Ralhan, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone: 425-227-1405; fax: 425-227-1149. Information may be emailed to: 9-ANM-116-AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of

the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) *Contacting the Manufacturer*: For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA); or Airbus's EASA Design Organization Approval (DOA). If approved by the DOA, the approval must include the DOA-authorized signature.

(3) *Reporting Requirements*: A federal agency may not conduct or sponsor, and a person is not required to respond to, nor shall a person be subject to a penalty for failure to comply with, a collection of information subject to the requirements of the Paperwork Reduction Act unless that collection of information displays a current valid OMB Control Number. The OMB Control Number for this information collection is 2120-0056. Public reporting for this collection of information is estimated to be approximately 5 minutes per response, including the time for reviewing instructions, completing and reviewing the collection of information. All responses to this collection of information are mandatory. Comments concerning the accuracy of this burden and suggestions for reducing the burden should be directed to the FAA at: 800 Independence Ave. SW., Washington, DC 20591, Attn: Information Collection Clearance Officer, AES-200.

(n) Related Information

(1) Refer to Mandatory Continuing Airworthiness Information (MCAI) EASA Airworthiness Directive EASA AD 2015-0117, dated June 24, 2015; corrected August 7, 2015, for related information. This MCAI may be found in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-3989.

(2) For Airbus service information identified in this proposed AD, contact Airbus, Airworthiness Office—EIAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email account.airworth-eas@airbus.com; Internet <http://www.airbus.com>. For BE service identified in this proposed AD, contact B/E Aerospace Inc., 10800 Pflumm Road, Lenexa, KS 66215; telephone: 913-338-9800; fax: 913-469-8419; Internet <http://beaerospace.com/home/globalsupport>. You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

Issued in Renton, Washington, on October 6, 2015.

Jeffrey E. Duven,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2015-26220 Filed 10-16-15; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2015-3988; Directorate Identifier 2015-NM-005-AD]

RIN 2120-AA64

Airworthiness Directives; Bombardier, Inc. Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to supersede Airworthiness Directive (AD) 2014-17-51, for certain Bombardier, Inc. Model CL-600-2B16 airplanes. AD 2014-17-51 currently requires inspecting the inboard flap fasteners of the hinge-box forward fitting at Wing Station (WS) 76.50 and WS 127.25 to determine the orientation and condition of the fasteners, as applicable, and replacement or repetitive inspections of the fasteners if necessary. AD 2014-17-51 also provides for optional terminating action for the requirements of that AD. Since we issued AD 2014-17-51, we have determined that additional action is necessary. This proposed AD would also require accomplishment of the previously optional terminating action. We are proposing this AD to detect and correct incorrectly oriented or fractured fasteners, which could result in premature failure of the fasteners attaching the inboard flap hinge-box forward fitting; failure of the fasteners could lead to the detachment of the flap hinge box and the flap surface, and consequent loss of control of the airplane.

DATES: We must receive comments on this proposed AD by December 3, 2015.

ADDRESSES: You may send comments by any of the following methods:

- *Federal eRulemaking Portal*: Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- *Fax*: 202-493-2251.

- *Mail*: U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

- *Hand Delivery*: U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Bombardier, Inc., 400 Côte-Vertu Road West, Dorval, Québec H4S 1Y9, Canada; telephone 514-855-5000; fax 514-855-7401; email thd.crj@aero.bombardier.com; Internet <http://www.bombardier.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-3988; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone 800-647-5527) is in the ADDRESSES section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Aziz Ahmed, Aerospace Engineer, Propulsion and Services Branch, ANE-173, FAA, New York Aircraft Certification Office (ACO), 1600 Stewart Avenue, Suite 410, Westbury, NY 11590; telephone 516-228-7329; fax 516-794-5531.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the ADDRESSES section. Include "Docket No. FAA-2015-3988; Directorate Identifier 2015-NM-005-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD based on those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

On October 13, 2014, we issued AD 2014-17-51, Amendment 39-17999 (79 FR 64088, October 28, 2014). AD 2014-17-51 requires actions intended to address an unsafe condition on certain

Bombardier, Inc. Model CL-600-2B16 airplanes.

The preamble to AD 2014-17-51, Amendment 39-17999 (79 FR 64088, October 28, 2014), explains that we consider the requirements "interim action" and were considering further rulemaking. We now have determined that further rulemaking is indeed necessary, and this proposed AD follows from that determination.

Transport Canada Civil Aviation (TCCA), which is the aviation authority for Canada, has issued Canadian Airworthiness Directive CF-2014-27R1, dated August 29, 2014 (referred to after this as the Mandatory Continuing Airworthiness Information, or "the MCAI"), to correct an unsafe condition for certain Bombardier, Inc. Model CL-600-2B16 airplanes. The MCAI states:

There have been three in-service reports on 604 Variant aeroplanes of a fractured fastener head on the inboard flap hinge-box forward fitting at Wing Station (WS) 76.50, found during a routine maintenance inspection. Investigation revealed that the installation of these fasteners on the inboard flap hinge-box forward fittings at WS 76.50 and WS 127.25, on both wings, does not conform to the engineering drawings. Incorrect installation may result in premature failure of the fasteners attaching the inboard flap hinge-box forward fitting. Failure of the fasteners could lead to the detachment of the flap hinge box and consequently the detachment of the flap surface. The loss of a flap surface could adversely affect the continued safe operation of the aeroplane.

The original issue of [Canadian] AD CF-2013-39 [<http://www.regulations.gov/#/documentDetail;D=FAA-2014-0054-0002>] [which corresponds to FAA AD 2014-03-17, Amendment 39-17754 (79 FR 9389, February 19, 2014)] mandated a detailed visual inspection (DVI) of each inboard flap hinge-box forward fitting, on both wings, and rectification as required. Incorrectly oriented fasteners require repetitive inspections until the terminating action is accomplished.

After the issuance of [Canadian] AD CF-2013-39, there has been one reported incident on a 604 Variant aeroplane where four fasteners were found fractured on the same flap hinge-box forward fitting. The investigation determined that the fasteners were incorrectly installed.

The original issue of this [Canadian] AD was issued to reduce the initial and repetitive inspection intervals previously mandated in [Canadian] AD CF-2013-39, and to impose replacement of the incorrectly oriented fasteners within 24 months. The CL-600-1A11, -2A12 and -2B16 (601-3A/-3R Variant) aeroplanes are addressed through [Canadian] AD CF-2013-39R1.

Revision 1 of this [Canadian] AD is issued to clarify the requirements for the initial and repetitive inspections.

You may examine the MCAI in the AD docket on the Internet at <http://www.regulations.gov> by searching for

and locating Docket No. FAA-2015-3988.

Related Service Information Under 1 CFR Part 51

Bombardier has issued Alert Service Bulletins:

- A604-57-006, Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013,
- A605-57-004, Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013, and

The service information describes detailed visual inspection of each inboard flap fastener of the hinge-box forward fitting at WS 76.50 and WS 127.25, on both wings, and, if necessary, replacement of the fasteners. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the ADDRESSES section of this NPRM.

FAA's Determination and Requirements of This Proposed AD

This product has been approved by the aviation authority of another country, and is approved for operation in the United States. Pursuant to our bilateral agreement with the State of Design Authority, we have been notified of the unsafe condition described in the MCAI and service information referenced above. We are proposing this AD because we evaluated all pertinent information and determined an unsafe condition exists and is likely to exist or develop on other products of the same type design.

Costs of Compliance

We estimate that this proposed AD affects 285 airplanes of U.S. registry.

The actions required by AD 2014-17-51, Amendment 39-17999 (79 FR 64088, October 28, 2014), and retained in this proposed AD take about 1 work-hour per product, at an average labor rate of \$85 per work-hour. Required parts cost about \$0 per product. Based on these figures, the estimated cost of the actions that are required by AD 2014-17-51 is \$85 per product.

We also estimate that it would take about 58 work-hours per product to comply with the basic requirements of this proposed AD. The average labor rate is \$85 per work-hour. Required parts would cost about \$753 per product. Based on these figures, we estimate the cost of this proposed AD on U.S. operators to be \$1,619,655, or \$5,683 per product.

In addition, we estimate that any necessary follow-on actions would take about 58 work-hours and require parts costing \$753, for a cost of \$5,683 per product. We have no way of determining the number of aircraft that might need this action.

According to the manufacturer, some of the costs of this proposed AD may be covered under warranty, thereby reducing the cost impact on affected individuals. We do not control warranty coverage for affected individuals. As a result, we have included all costs in our cost estimate.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by removing Airworthiness Directive (AD) 2014–17–51, Amendment 39–17999 (79 FR 64088, October 28, 2014), and adding the following new AD:

Bombardier, Inc.: Docket No. FAA–2015–3988; Directorate Identifier 2015–NM–005–AD.

(a) Comments Due Date

We must receive comments by December 3, 2015.

(b) Affected ADs

(1) This AD replaces AD 2014–17–51, Amendment 39–17999 (79 FR 64088, October 28, 2014).

(2) This AD affects AD 2014–03–17, Amendment 39–17754 (79 FR 9389, February 19, 2014), only for the airplanes identified in paragraph (c) of this AD.

(c) Applicability

This AD applies to Bombardier, Inc. Model CL–600–2B16 airplanes, certificated in any category, serial numbers 5301 through 5665 inclusive, and 5701 through 5920 inclusive.

(d) Subject

Air Transport Association (ATA) of America Code 57, Wings.

(e) Reason

This AD was prompted by reports of fractured fastener heads on the inboard flap hinge-box forward fitting at Wing Station (WS) 76.50 due to incorrect installation. We are issuing this AD to detect and correct incorrectly oriented or fractured fasteners, which could result in premature failure of the fasteners attaching the inboard flap hinge-box forward fitting; failure of the fasteners could lead to the detachment of the flap hinge box and the flap surface, and consequent loss of control of the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Retained Inspection, With New Service Information: Airplanes Not Previously Inspected

This paragraph restates the requirements of paragraph (g) of AD 2014–17–51, Amendment 39–17999 (79 FR 64088, October 28, 2014), with new service information. For airplanes on which the actions required by AD 2014–03–17, Amendment 39–17754 (79 FR 9389, February 19, 2014), have not been done as of November 12, 2014 (the effective date of AD 2014–17–51): Within 10 flight cycles after November 12, 2014, or within 100 flight cycles after March 6, 2014 (the effective date of AD 2014–03–17, Amendment 39–17754 (79 FR 9389, February 19, 2014)), whichever occurs first, do a detailed visual inspection of each inboard flap fastener of the hinge-box forward fitting at WS 76.50 and WS 127.25, on both wings, to determine if the fasteners are correctly oriented and intact (non-fractured, with intact fastener head). Do the inspection in accordance with the Accomplishment Instructions of Bombardier Alert Service Bulletin A604–57–006, Revision 01, dated September 26, 2013, including Appendices 1 and 2, dated September 26, 2013, Revision 02, dated January 22, 2014, including Appendices 1 and 2, dated September 26, 2013, or Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013 (for serial numbers 5301 through 5665 inclusive); or Bombardier Alert Service Bulletin A605–57–004, Revision 01, dated September 26, 2013, including Appendices 1 and 2, dated September 26, 2013, Revision 02, dated January 22, 2014, including Appendices 1 and 2, dated September 26, 2013, or Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013 (for serial numbers 5701 through 5920 inclusive). As of the effective date of this AD, only use Bombardier Alert Service Bulletin A604–57–006, Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013; or Bombardier Alert Service Bulletin A605–57–004, Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013; as applicable; for the actions required by this paragraph.

(1) If all fasteners are found intact and correctly oriented, no further action is required by this AD.

(2) If any fastener is found fractured: Before further flight, remove and replace all forward and aft fasteners at WS 76.50 and WS 127.25, regardless of condition or orientation, on both wings, in accordance with the Accomplishment Instructions of the applicable service information identified in paragraph (g) of this AD. As of the effective date of this AD, only use Bombardier Alert Service Bulletin A604–57–006, Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013; or Bombardier Alert Service Bulletin A605–57–004, Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013; as applicable; for the actions required by this paragraph. After replacement of all fasteners as required by this paragraph of this AD, no further action is required by this AD.

(3) If any incorrectly oriented but intact fastener is found, and no fractured fastener is found, repeat the inspection required by paragraph (g) of this AD thereafter at intervals not to exceed 10 flight cycles, until the requirements of paragraph (i)(1) or (k) of this AD have been done.

(h) Retained Actions, With New Service Information: Airplanes Previously Inspected, Having Incorrectly Oriented Fastener(s)

This paragraph restates the requirements of paragraph (h) of AD 2014-17-51, Amendment 39-17999 (79 FR 64088, October 28, 2014), with new service information. For airplanes on which an inspection required by paragraph (g) or (j) of AD 2014-03-17, Amendment 39-17754 (79 FR 9389, February 19, 2014), has been done as of November 12, 2014 (the effective date of AD 2014-17-51), and on which any incorrectly oriented fastener, but no fractured fastener, was found: Except as provided by paragraph (i)(3) of this AD, do a detailed visual inspection of all inboard flap fasteners of the hinge-box forward fitting at WS 76.50 and WS 127.25, on both wings, to determine if the fasteners are intact (non-fractured, with intact fastener head). Inspect within 10 flight cycles after November 12, 2014, or within 100 flight cycles after the most recent inspection done as required by AD 2014-03-17, whichever occurs first. Inspect in accordance with the Accomplishment Instructions of the applicable service information identified in paragraph (g) of this AD. As of the effective date of this AD, only use Bombardier Alert Service Bulletin A604-57-006, Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013; or Bombardier Alert Service Bulletin A605-57-004, Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013; as applicable; for the actions required by this paragraph.

(1) If all fasteners are found intact, repeat the inspection thereafter at intervals not to exceed 10 flight cycles, until the requirements of paragraph (i)(1) or (k) of this AD have been done.

(2) If any fastener is found fractured: Before further flight, remove and replace all forward and aft fasteners at WS 76.50 and WS 127.25, regardless of condition or orientation, on both wings, in accordance with the Accomplishment Instructions of the applicable service information identified in paragraph (g) of this AD. As of the effective date of this AD, only use Bombardier Alert Service Bulletin A604-57-006, Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013; or Bombardier Alert Service Bulletin A605-57-004, Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013; as applicable; for the actions required by this paragraph. After replacement of all fasteners as required by this paragraph, no further action is required by this AD.

(i) Retained Terminating Action, With New Service Information

This paragraph restates the terminating action specified in paragraph (i) of AD 2014-

17-51, Amendment 39-17999 (79 FR 64088, October 28, 2014), with new service information.

(1) Replacement of all forward and aft fasteners at WS 76.50 and WS 127.25, on both wings, in accordance with the Accomplishment Instructions of the applicable service information identified in paragraph (g) of this AD, terminates the requirements of this AD. As of the effective date of this AD, only use Bombardier Alert Service Bulletin A604-57-006, Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013; or Bombardier Alert Service Bulletin A605-57-004, Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013; as applicable; for the actions specified in this paragraph.

(2) Accomplishment of the applicable requirements of this AD constitutes terminating action for the requirements of AD 2014-03-17, Amendment 39-17754 (79 FR 9389, February 19, 2014), for that airplane only.

(3) Replacement of all fractured and incorrectly oriented fasteners before November 12, 2014 (the effective date of AD 2014-17-51, Amendment 39-17999 (79 FR 64088, October 28, 2014)), as provided by paragraph (i) or (k) of AD 2014-03-17, Amendment 39-17754 (79 FR 9389, February 19, 2014), is acceptable for compliance with the requirements of this AD.

(j) Retained Special Flight Permit Prohibition

This paragraph restates the requirements of paragraph (j) of AD 2014-17-51, Amendment 39-17999 (79 FR 64088, October 28, 2014). Special flight permits to operate the airplane to a location where the airplane can be repaired in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) are not allowed.

(k) New Requirement of This AD: Post-Inspection Fastener Replacement

For airplanes on which incorrectly oriented fasteners were found during any inspection required by paragraph (g), (g)(3), (h), or (h)(1) of this AD, but none were found to be fractured: Within 24 months after the effective date of this AD, remove and replace all forward and aft fasteners at WS 76.50 and WS 127.25, regardless of condition or orientation, on both wings, in accordance with the Accomplishment Instructions of Bombardier Alert Service Bulletin A604-57-006, Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013 (for serial numbers 5301 through 5665 inclusive); or Bombardier Alert Service Bulletin A605-57-004, Revision 04, dated November 12, 2014, including Appendices 1 and 2, dated September 26, 2013 (for serial numbers 5701 through 5920 inclusive). Accomplishing the requirements of this paragraph terminates the requirements of this AD.

(l) Credit for Previous Actions

(1) This paragraph provides credit for actions required by paragraphs (g), (h), and (i)(1) of this AD, if those actions were performed before the effective date of this AD

using the applicable service information identified in paragraphs (l)(1)(i) and (l)(1)(ii) of this AD, which are not incorporated by reference in this AD.

(i) Bombardier Alert Service Bulletin A604-57-006, Revision 03, dated August 19, 2014, including Appendices 1 and 2, dated September 26, 2013.

(ii) Bombardier Alert Service Bulletin A605-57-004, Revision 03, dated August 19, 2014, including Appendices 1 and 2, dated September 26, 2013.

(2) This paragraph provides credit for actions required by paragraph (k) of this AD, if those actions were done before the effective date of this AD using the applicable service information identified in paragraphs (l)(2)(i) through (l)(2)(iv) of this AD.

(i) Bombardier Alert Service Bulletin A604-57-006, Revision 01, dated September 26, 2013, including Appendices 1 and 2, dated September 26, 2013, which is incorporated by reference in AD 2014-03-17, Amendment 39-17754 (79 FR 9389, February 19, 2014).

(ii) Bombardier Alert Service Bulletin A604-57-006, Revision 02, dated January 22, 2014, including Appendices 1 and 2, dated September 26, 2013, which is incorporated by reference in AD 2014-17-51, Amendment 39-17999 (79 FR 64088, October 28, 2014).

(iii) Bombardier Alert Service Bulletin A605-57-004, Revision 01, dated September 26, 2013, including Appendices 1 and 2, dated September 26, 2013, which is incorporated by reference in AD 2014-03-17, Amendment 39-17754 (79 FR 9389, February 19, 2014).

(iv) Bombardier Alert Service Bulletin A604-57-004, Revision 02, dated January 22, 2014, including Appendices 1 and 2, dated September 26, 2013, which is incorporated by reference in AD 2014-17-51, Amendment 39-17999 (79 FR 64088, October 28, 2014).

(m) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs)*: The Manager, New York Aircraft Certification Office (ACO), ANE-170, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the ACO, send it to ATTN: Program Manager, Continuing Operational Safety, FAA, New York ACO, 1600 Stewart Avenue, Suite 410, Westbury, NY 11590; telephone 516-228-7300; fax 516-794-5531.

(i) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(ii) AMOCs previously approved for AD 2014-17-51, Amendment 39-17999 (79 FR 64088, October 28, 2014), are acceptable for the corresponding requirements of this AD.

(2) *Contacting the Manufacturer*: As of the effective date of this AD, for any requirement in this AD to obtain corrective actions from

a manufacturer, the action must be accomplished using a method approved by the Manager, New York ACO, ANE-170, FAA; or Transport Canada Civil Aviation (TCCA); or Bombardier, Inc.'s TCCA Design Approval Organization (DAO). If approved by the DAO, the approval must include the DAO-authorized signature.

(n) Related Information

(1) Refer to Mandatory Continuing Airworthiness Information (MCAI) Canadian Emergency Airworthiness Directive CF-2014-27R1, dated August 29, 2014, for related information. This MCAI may be found in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-3988.

(2) For service information identified in this AD, contact Bombardier, Inc., 400 Côte Vertu Road West, Dorval, Québec H4S 1Y9, Canada; telephone 514-855-5000; fax 514-855-7401; email thd.crj@aero.bombardier.com; Internet <http://www.bombardier.com>. You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

Issued in Renton, Washington, on October 6, 2015.

Jeffrey E. Duven,

Manager, Transport Airplane Directorate,
Aircraft Certification Service.

[FR Doc. 2015-26222 Filed 10-16-15; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2015-4112; Directorate Identifier 2014-SW-043-AD]

RIN 2120-AA64

Airworthiness Directives; Airbus Helicopters (previously Eurocopter France)

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to supersede airworthiness directive (AD) 2010-23-02 for Eurocopter France (now Airbus Helicopters) Model SA-365N, SA-365N1, AS-365N2, and AS 365 N3 helicopters. AD 2010-23-02 currently requires amending the Limitations section of the Rotorcraft Flight Manual (RFM) to limit the never-exceed velocity (VNE) to 150 Knots Indicated Air Speed (KIAS) and to add a 1,500 ft/minute rate of descent (R/D) limitation beyond 140 KIAS. Since we issued AD 2010-23-02, a design change designated as

modification (MOD) 0755B28 improved the dynamic behavior of the horizontal stabilizer such that AD actions are not required. This proposed AD would retain the requirements of AD 2010-23-01 and revise the applicability to exclude helicopters with MOD 0755B28. These proposed actions are intended to exclude certain helicopters from the applicability and restrict the VNE on other helicopters to prevent failure of the horizontal stabilizer and subsequent loss of control of the helicopter.

DATES: We must receive comments on this proposed AD by December 18, 2015.

ADDRESSES: You may send comments by any of the following methods:

- **Federal eRulemaking Docket:** Go to <http://www.regulations.gov>. Follow the online instructions for sending your comments electronically.

- **Fax:** 202-493-2251.

- **Mail:** Send comments to the U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590-0001.

- **Hand Delivery:** Deliver to the "Mail" address between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> in Docket No. FAA-2015-4112; or in person at the Docket Operations Office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the European Aviation Safety Agency (EASA) AD, the economic evaluation, any comments received and other information. The street address for the Docket Operations Office (telephone 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

For service information identified in this proposed AD, contact Airbus Helicopters, Inc., 2701 N. Forum Drive, Grand Prairie, TX 75052; telephone (972) 641-0000 or (800) 232-0323; fax (972) 641-3775; or at <http://www.airbushelicopters.com/techpub>. You may review service information at the FAA, Office of the Regional Counsel, Southwest Region, 10101 Hillwood Pkwy., Room 6N-321, Fort Worth, Texas 76177.

FOR FURTHER INFORMATION CONTACT: Robert Grant, Aviation Safety Engineer, Safety Management Group, FAA, 10101 Hillwood Pkwy., Fort Worth, Texas

76177; telephone (817) 222-5110; email robert.grant@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to participate in this rulemaking by submitting written comments, data, or views. We also invite comments relating to the economic, environmental, energy, or federalism impacts that might result from adopting the proposals in this document. The most helpful comments reference a specific portion of the proposal, explain the reason for any recommended change, and include supporting data. To ensure the docket does not contain duplicate comments, commenters should send only one copy of written comments, or if comments are filed electronically, commenters should submit only one time.

We will file in the docket all comments that we receive, as well as a report summarizing each substantive public contact with FAA personnel concerning this proposed rulemaking. Before acting on this proposal, we will consider all comments we receive on or before the closing date for comments. We will consider comments filed after the comment period has closed if it is possible to do so without incurring expense or delay. We may change this proposal in light of the comments we receive.

Discussion

On October 15, 2010, we issued AD 2010-23-02, Amendment 39-16491 (75 FR 68169, November 5, 2010) for Eurocopter France (now Airbus Helicopters) Model SA-365N, SA-365N1, AS-365N2, and AS 365 N3 helicopters. AD 2010-23-02 requires amending the Limitations section of the RFM to limit the VNE to 150 KIAS and to add a 1,500 ft/minute R/D limitation beyond 140 KIAS and installing one or more placards on the cockpit instrument panel in full view of the pilot and copilot. AD 2010-23-01 was prompted by failures of the horizontal stabilizers on then-recently delivered Model AS 365 N3 helicopters due to a vibration phenomenon that may arise during the descent flight phases at high speed regardless of the stabilizer installed. Those actions were intended to prevent failure of the horizontal stabilizer and subsequent loss of control of the helicopter.

Actions Since AD 2010-23-02 Was Issued

Since we issued AD 2010-23-01 (75 FR 68169, November 5, 2010), Eurocopter France changed its name to Airbus Helicopters. EASA, which is the