

accordance with the Accomplishment Instructions of Airbus Service Bulletin A320-53-1218, Revision 01, including Appendices 01 and 02, dated June 17, 2010. If any cracking is found, before further flight, repair in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320-53-1218, Revision 01, including Appendices 01 and 02, dated June 17, 2010.

(h) Reporting

Submit a report of the findings of the inspection required by paragraph (g) of this AD to Airbus in accordance with Appendix 01 of Airbus Service Bulletin A320-53-1218, Revision 01, including Appendices 01 and 02, dated June 17, 2010, at the applicable time specified in paragraph (h)(1) or (h)(2) of this AD.

(1) If the inspection was done on or after the effective date of this AD: Submit the report within 90 days after the inspection.

(2) If the inspection was done before the effective date of this AD: Submit the report within 90 days after the effective date of this AD.

(i) Credit for Previous Actions

This paragraph provides credit for inspections and replacements required by paragraph (g) of this AD, if those actions were performed before the effective date of this AD using Airbus Service Bulletin A320-53-1218, including Appendices 01 and 02, dated February 8, 2010.

(j) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs)*: The Manager, International Branch, ANM-116, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Sanjay Ralhan, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, Washington 98057-3356; phone: 425-227-1405; fax: 425-227-1149. Information may be emailed to: 9-ANM-116-AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) *Airworthy Product*: For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(3) *Reporting Requirements*: A federal agency may not conduct or sponsor, and a person is not required to respond to, nor shall a person be subject to a penalty for failure to comply with a collection of

information subject to the requirements of the Paperwork Reduction Act unless that collection of information displays a current valid OMB Control Number. The OMB Control Number for this information collection is 2120-0056. Public reporting for this collection of information is estimated to be approximately 5 minutes per response, including the time for reviewing instructions, completing and reviewing the collection of information. All responses to this collection of information are mandatory. Comments concerning the accuracy of this burden and suggestions for reducing the burden should be directed to the FAA at: 800 Independence Ave. SW., Washington, DC 20591, Attn: Information Collection Clearance Officer, AES-200.

(k) Related Information

Refer to MCAI European Aviation Safety Agency Airworthiness Directive 2011-0120R1, dated July 13, 2011; and Airbus Service Bulletin A320-53-1218, Revision 01, including Appendices 01 and 02, dated June 17, 2010; for related information.

(l) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the following service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use the following service information to do the actions required by this AD, unless the AD specifies otherwise.

(i) Airbus Service Bulletin A320-53-1218, Revision 01, including Appendices 01 and 02, dated June 17, 2010.

(3) For service information identified in this AD, contact Airbus, Airworthiness Office—EAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email account.airworth-eas@airbus.com; Internet <http://www.airbus.com>.

(4) You may review copies of the service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221.

(5) You may also review copies of the service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at an NARA facility, call 202-741-6030, or go to http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Issued in Renton, Washington, on July 5, 2012.

Kalene C. Yanamura,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2012-17389 Filed 7-20-12; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2012-0730; Directorate Identifier 2012-SW-048-AD; Amendment 39-17124; AD 2012-14-10]

RIN 2120-AA64

Airworthiness Directives; Boeing Vertol (Type Certificate Currently Held by Columbia Helicopters, Inc. (CHI)) and Kawasaki Heavy Industries, Limited Helicopters (Kawasaki)

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule; request for comments.

SUMMARY: We are adopting a new airworthiness directive (AD) for CHI Model 107-II and Kawasaki Model KV107-II and KV107-IIA helicopters. This AD requires, before further flight, replacing certain upper collective pitch control yoke bolts. This AD is prompted by three failures of the affected bolts. These actions are intended to prevent failure of an upper collective pitch control yoke bolt (bolt), excessive vibration, migration of the shafts, and subsequent loss of control of the helicopter.

DATES: This AD becomes effective August 7, 2012.

We must receive comments on this AD by September 21, 2012.

ADDRESSES: You may send comments by any of the following methods:

- *Federal eRulemaking Docket:* Go to <http://www.regulations.gov>. Follow the online instructions for sending your comments electronically.

- *Fax:* 202-493-2251.

- *Mail:* Send comments to the U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590-0001.

- *Hand Delivery:* Deliver to the "Mail" address between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the Docket Operations Office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the economic evaluation, any comments received, and other information. The street address for

the Docket Operations Office (telephone 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

For service information identified in this AD, contact Columbia Helicopters, Inc.; 14452 Arndt Road NE., Aurora, OR 97002; telephone (503) 678-1222; email ContactEngineering@colheli.com; or at www.ColHeli.com. You may review the referenced service information at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137.

FOR FURTHER INFORMATION CONTACT:

Kathleen Arrigotti, Aviation Safety Engineer, Seattle Aircraft Certification Office, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, Washington 98057; telephone (425) 917-6426; email kathleen.arrigotti@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

This AD is a final rule that involves requirements affecting flight safety, and we did not provide you with notice and an opportunity to provide your comments prior to it becoming effective. However, we invite you to participate in this rulemaking by submitting written comments, data, or views. We also invite comments relating to the economic, environmental, energy, or federalism impacts that resulted from adopting this AD. The most helpful comments reference a specific portion of the AD, explain the reason for any recommended change, and include supporting data. To ensure the docket does not contain duplicate comments, commenters should send only one copy of written comments, or if comments are filed electronically, commenters should submit them only one time. We will file in the docket all comments that we receive, as well as a report summarizing each substantive public contact with FAA personnel concerning this rulemaking during the comment period. We will consider all the comments we receive and may conduct additional rulemaking based on those comments.

Discussion

We are adopting a new AD for CHI Model 107-II and Kawasaki Model KV107-II and KV107-IIA helicopters. This AD requires replacing certain part-numbered bolts. The bolts are located in the forward and aft rotor upper collective pitch control assemblies. Upon failure, the yoke bolt head shears off, allowing shafts around the bolt to migrate out of place. If the shaft migrates fully out of place, the rotor

pitch cannot be controlled. This AD is prompted by multiple failures of the affected bolts at low flight hours. These actions are intended to prevent failure of a bolt, excessive vibration, migration of the shafts, and subsequent loss of control of the helicopter.

FAA's Determination

We are issuing this AD because we evaluated all the relevant information and determined the unsafe condition described previously is likely to exist or develop in other helicopters of the same type designs.

Related Service Information

CHI issued Service Bulletin No. 107-27-0005 (SB), Revision 0, dated April 26, 2012. The SB specifies replacing all bolts, part number (P/N) 107C2733-1 and P/N 107C2733-2, with bolts, P/N C07C2700-1. The SB also specifies modifying spare assemblies, daily inspections of the yoke bolt retaining nut, and recurring 35-hour inspections of the bolt.

AD Requirements

This AD requires, before further flight, replacing all affected bolts with airworthy bolts, P/N C07C2700-1, and torquing the nut to 450-500 in-lbs. This AD also prohibits installing washer P/N A02C3112-13 with bolt P/N C07C2700-1, and installing bolt P/N 107C2733-1 and P/N 107C2733-2 on any helicopter.

Differences Between This AD and the Service Information

The CHI SB requires a repetitive daily inspection of the yoke bolt retaining nut, and a repetitive 35-hour inspection of the bolts after replacement. This AD does not require those inspections. The SB also specifies modifying spare assemblies. This AD does not address parts that are not installed.

Costs of Compliance

We estimate that this AD will affect 12 helicopters of U.S. Registry. We estimate that operators may incur the following costs in order to comply with this AD: Replacing the bolts will take 4 hours at an average labor rate of \$85 per work hour. Required parts will cost \$2,000. The total cost per helicopter is \$2,340 and the total cost for the entire U.S. fleet is \$28,080.

FAA's Justification and Determination of the Effective Date

Providing an opportunity for public comments before adopting these AD requirements would delay implementing the safety actions needed to correct a previously described known critical unsafe condition, which can

adversely affect the structural integrity and controllability of the helicopter. Therefore, we find that the risk to the flying public justifies waiving notice and comment prior to the adoption of this rule because the required corrective actions must be accomplished before further flight.

Since an unsafe condition exists that requires the immediate adoption of this AD, we determined that notice and opportunity for public comment before issuing this AD are impracticable and that good cause exists for making this amendment effective in less than 30 days.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed, I certify that this AD:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska to the extent that it justifies making a regulatory distinction; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared an economic evaluation of the estimated costs to comply with this AD and placed it in the AD docket.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2012–14–10 Boeing Vertol (Type Certificate Currently Held by Columbia Helicopters, Inc.) and Kawasaki Heavy Industries, Limited Helicopters: Amendment 39–17124; Docket No. FAA–2012–0730; Directorate Identifier 2012–SW–048–AD.

(a) Applicability

This AD applies to Boeing Vertol (type certificate currently held by Columbia Helicopters, Inc. (CHI)) Model 107–II and Kawasaki Heavy Industries, Limited Model KV107–II and KV107–IIA helicopters with an upper collective pitch control assembly, part number (P/N) 107CK003–2 or 107CK002–2, installed, certificated in any category.

(b) Unsafe Condition

This AD defines the unsafe condition as failure of an upper collective pitch control yoke bolt (bolt). This condition could result in excessive vibration, migration of the shafts, and subsequent loss of control of the helicopter.

(c) Effective Date

This AD becomes effective August 7, 2012.

(d) Compliance

You are responsible for performing each action required by this AD within the specified compliance time unless it has already been accomplished prior to that time.

(e) Required Actions

Before further flight, replace bolts, P/N 107C2733–1 and P/N 107C2733–2, with airworthy bolts, P/N C07C2700–1. Torque each nut to 450–500 in-lbs. Do not install a washer, P/N A02C3112–13 with a bolt, P/N C07C2700–1. Do not install bolts, P/N 107C2733–1 and P/N 107C2733–2, on any helicopter.

(f) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Seattle Aircraft Certification Office, FAA, may approve AMOCs for this AD. Send your proposal to: Kathleen Arrigotti, Aviation Safety Engineer, Seattle Aircraft Certification Office, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, Washington 98057; telephone (425) 917–6426; email 9-ANM-Seattle-ACO-AMOC-Requests@faa.gov.

(2) For operations conducted under a 14 CFR part 119 operating certificate or under 14 CFR part 91, subpart K, we suggest that you notify your principal inspector, or lacking a principal inspector, the manager of the local flight standards district office or certificate holding district office, before operating any aircraft complying with this EAD through an AMOC.

(g) Additional Information

CHI Service Bulletin No. 107–27–0005 (SB), Revision 0, dated April 26, 2012, which is not incorporated by reference, contains additional information about the subject of this AD. For service information identified in this AD, contact Columbia Helicopters, Inc.; 14452 Arndt Road NE., Aurora, OR 97002; telephone (503) 678–1222; email ContactEngineering@colheli.com; or at www.ColHeli.com. You may review the referenced service information at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137.

(h) Subject

Joint Aircraft Service Component (JASC) Code: 6230 Main Rotor Mast/Swashplate.

Issued in Fort Worth, Texas, on July 5, 2012.

Kim Smith,

Manager, Rotorcraft Directorate, Aircraft Certification Service.

[FR Doc. 2012–17278 Filed 7–20–12; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2012–0271; Directorate Identifier 2011–NM–196–AD; Amendment 39–17118; AD 2012–14–04]

RIN 2120–AA64

Airworthiness Directives; Bombardier, Inc. Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for certain Bombardier, Inc. Model DHC–8–100, DHC–8–200, and DHC–8–300 series airplanes. This AD was prompted by

reports of hydraulic accumulator screw cap or end cap failure. This AD requires replacing the affected parking brake accumulator. We are issuing this AD to prevent failure of the parking brake accumulator screw caps or end caps, which could result in loss of the number 2 hydraulic system and damage to airplane structures, and could potentially have an adverse effect on the controllability of the airplane.

DATES: This AD becomes effective August 27, 2012.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of August 27, 2012.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE., Washington, DC.

FOR FURTHER INFORMATION CONTACT: Cesar Gomez, Aerospace Engineer, Airframe and Mechanical Systems Branch, ANE–171, FAA, New York Aircraft Certification Office, 1600 Stewart Avenue, Suite 410, Westbury, New York 11590; telephone (516) 228–7318; fax (516) 794–5531.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM was published in the **Federal Register** on March 21, 2012 (77 FR 16488). That NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

Seven cases of on-ground hydraulic accumulator screw cap or end cap failure have been experienced on CL–600–2B19 (CRJ) aeroplanes, resulting in loss of the associated hydraulic system and high-energy impact damage to adjacent systems and structure. To date, the lowest number of flight cycles accumulated at the time of failure has been 6991.

Although there have been no failures to date on any DHC–8 aeroplanes, similar accumulators to those installed on the CL–600–2B19, Part Numbers (P/N) 0860162001 and 0860162002 (Parking Brake Accumulator), are installed on the aeroplanes listed in the Applicability section of this [TCCA] directive.

A detailed analysis of the systems and structure in the potential line of trajectory of a failed screw cap/end cap for the accumulator has been conducted. It has identified that the worst-case scenarios would be the loss of number 2 hydraulic system, and damage to aeroplane structures.