

SMALL BUSINESS SIZE STANDARDS BY NAICS INDUSTRY—Continued

NAICS codes	NAICS U.S. industry title	Size standards in millions of dollars	Size standards in number of employees
488510	Freight Transportation Arrangement ¹⁰	10 14.0	

Footnotes

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10. NAICS codes 488510 (part) 531210, 541810, 561510, 561520, and 561920—As measured by total revenues, but excluding funds received in trust for an unaffiliated third party, such as bookings or sales subject to commissions. The commissions received are included as revenues.

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Dated: December 21, 2011.

Karen G. Mills,
Administrator.

[FR Doc. 2012-4330 Filed 2-23-12; 8:45 am]

BILLING CODE 8025-01-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2006-25738; Directorate Identifier 2006-NE-27-AD; Amendment 39-16961; AD 2012-04-05]

RIN 2120-AA64

Airworthiness Directives; General Electric Company (GE) Turbofan Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are superseding an existing airworthiness directive (AD) for all GE CF6-80C2B series turbofan engines. That AD currently requires installing software version 8.2.Q1 to the engine electronic control unit (ECU), which increases the engine's margin to flameout. This new AD requires the removal of the affected ECUs from service. This AD was prompted by two reports of engine flameout events during flight in inclement weather conditions, eight reports of engine in-flight shutdown (IFSD) events caused by dual-channel central processing unit (CPU) faults in the ECU, and four reports of engine flameout ground events. We are issuing this AD to prevent engine flameout or un-commanded engine IFSD

of one or more engines, leading to an emergency or forced landing of the airplane.

DATES: This AD is effective March 30, 2012.

ADDRESSES:

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Tomasz Rakowski, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; phone: 781-238-7735; fax: 781-238-7199; email: tomasz.rakowski@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to supersede AD 2007-12-07, Amendment 39-15085 (72 FR 31174, June 6, 2007). That AD applies to the specified products. The NPRM was published in the **Federal Register** on November 14, 2011 (76 FR 70382). That NPRM proposed to remove the affected ECUs from service.

Comments

We gave the public the opportunity to participate in developing this AD. The following presents the comments received on the proposal and the FAA's response to each comment.

Request To Change Unsafe Condition

Commenter GE stated that in all of the events of flameout the engines relit and in all dual-channel CPU fault in-flight

shutdowns the engines were capable of restarting. GE stated that these events should not be considered unsafe conditions.

We do not agree. Although a flameout with a consecutive relight or an in-flight shutdown with a consecutive restart during cruise flight is not in itself an unsafe condition, these types of loss of thrust can be unsafe conditions during takeoff or during approach and landing. We did not change the AD.

Request To Clarify Engine Flight Cycle and ECU Cycle Count

Commenter All Nippon Airways (ANA) requested that we clarify the relationship between the engine flight cycles and ECU cycles of operation in the engine, and whether previous ECU history affects the flight cycle count.

We do not agree. The flight cycle intervals in paragraph (g) of the AD refer to the engine start-stop cycles with the affected ECU part numbers (P/Ns) installed, rather than ECU operational cycles. Engine flight cycles accrued before the effective date of the AD are not accounted for in the cycle count. We did not change the AD.

Request To Remove Certain Affected ECU P/Ns From the AD

Commenters Atlas Air, ANA, KLM, and China Airlines requested that we remove from the list of affected ECU P/Ns in Table 2 of the AD, ECUs with software version 8.2.Q1 and 8.2.R, a new front panel assembly (FPA) and an old pressure subsystem (PSS), or an old FPA and a new PSS generation circuit boards.

We do not agree. Dual-channel CPU faults have not been ruled out for the new FPA or the new PSS, therefore any ECU with either a new FPA or a new PSS must be addressed regardless of the version of software installed. We did not change the AD.

Request To Add ECU P/Ns to the AD

Commenter Atlas Air stated that ECUs P/Ns 1471M63P41, 1519M89P31, and 1820M33P14 are not listed in the proposed AD, but should be listed.

We do not agree. Those ECUs have the old generation of FPA and PSS circuit

boards and, therefore, are not susceptible to dual-channel CPU faults. The referenced ECUs also have the latest available version of software installed. We did not change the AD.

Request To Mandate Software Version 8.2.R or Later

Commenter Atlas Air requested to add a requirement to install software version 8.2.R or later in all affected engines at specified times, without regard to FPA and PSS circuit board hardware configuration.

We do not agree. Certain ECU P/Ns that have software version 8.2.R are susceptible to CPU channel faults. We did not change the AD.

Request To Modify ECUs

Commenter Atlas Air requested to modify ECU P/Ns 1471M63P42, 1519M89P32, and 1820M33P15 to ECU P/Ns 1471M63P41, 1519M89P31, and 1820M33P14, respectively.

We do not agree. No approved procedure exists to downgrade the ECUs. Engine owners and operators may propose such a procedure for approval, and request an alternative method of compliance to the AD, as specified in paragraph (i) of the AD. We did not change the AD.

Request To Add ECU Rework Procedures

Commenter ANA requested that we add rework procedures to the AD to modify affected ECUs into serviceable configurations of ECUs.

We do not agree. The AD is written to only remove affected ECU P/Ns from service. Refer to the manufacturer's service information for upgrading affected ECUs. We did not change the AD.

Conclusion

We reviewed the relevant data, considered the comments received, and determined that air safety and the public interest require adopting the AD as proposed.

Costs of Compliance

We estimate that this AD will affect 697 GE CF6-80C2B series turbofan engines installed on airplanes of U.S. registry. We also estimate that it will take about 4 work-hours per engine to perform a removal and replacement of the ECU, and that the average labor rate is \$85 per work-hour. A replacement

ECU costs about \$4,600. Based on these figures, we estimate the total cost of the AD to U.S. operators to be \$3,443,180.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in subtitle VII, part A, subpart III, section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator,

the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by removing airworthiness directive (AD) 2007-12-07, Amendment 39-15085 (72 FR 31174, June 6, 2007), and adding the following new AD:

2012-04-05 General Electric Company (GE): Amendment 39-16961; Docket No. FAA-2006-25738; Directorate Identifier 2006-NE-27-AD.

(a) Effective Date

This airworthiness directive (AD) is effective March 30, 2012.

(b) Affected ADs

This AD supersedes AD 2007-12-07, Amendment 39-15085 (72 FR 31174, June 6, 2007).

(c) Applicability

This AD applies to GE CF6-80C2B1F, CF6-80C2B1F1, CF6-80C2B1F2, CF6-80C2B2F, CF6-80C2B3F, CF6-80C2B4F, CF6-80C2B5F, CF6-80C2B6F, CF6-80C2B6FA, CF6-80C2B7F, and CF6-80C2B8F turbofan engines, including engines marked on the engine data plate as CF6-80C2B7F1.

(d) Unsafe Condition

- This AD results from:
- (1) Two reports of engine flameout events during flight in inclement weather conditions; and
 - (2) Eight reports of engine in-flight shutdown (IFSD) events caused by dual-channel central processing unit (CPU) faults in the electronic control unit (ECU); and
 - (3) Four reports of engine flameout ground events.

(e) We are issuing this AD to prevent engine flameout or un-commanded engine IFSD of one or more engines, leading to an emergency or forced landing of the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) ECU Removal

(1) Remove from service ECUs with part numbers (P/Ns) listed in Table 1 of this AD within 6 months or 450 engine flight cycles after the effective date of this AD, whichever occurs first.

TABLE 1—AFFECTED ECU P/Ns

1471M63P01	1471M63P02	1471M63P03	1471M63P04	1471M63P05
1471M63P06	1471M63P07	1471M63P08	1471M63P09	1471M63P10
1471M63P11	1471M63P12	1471M63P13	1471M63P14	1471M63P15
1471M63P16	1471M63P17	1471M63P18	1471M63P23	1471M63P24

TABLE 1—AFFECTED ECU P/Ns—Continued

1471M63P25	1471M63P26	1471M63P27	1471M63P28	1471M63P29
1471M63P30	1471M63P31	1471M63P32	1471M63P33	1471M63P34
1471M63P35	1471M63P36	1519M89P01	1519M89P02	1519M89P03
1519M89P04	1519M89P05	1519M89P06	1519M89P07	1519M89P08
1519M89P09	1519M89P10	1519M89P13	1519M89P14	1519M89P15
1519M89P16	1519M89P17	1519M89P18	1519M89P19	1519M89P20
1519M89P21	1519M89P22	1519M89P23	1519M89P24	1519M89P25
1519M89P26	1820M33P01	1820M33P02	1820M33P03	1820M33P04
1820M33P05	1820M33P06	1820M33P07	1820M33P08	1820M33P09

(2) Remove from service ECUs with P/Ns 2121M37P01, 2121M37P02, 2121M38P01, 2121M38P02, 2121M41P01 and 2121M41P02 within 14 months or 1,050 engine flight

cycles after the effective date of this AD, whichever occurs first.

(3) Remove from service ECUs with P/Ns listed in Table 2 of this AD within 60 months

or 4,500 engine flight cycles after the effective date of this AD, whichever occurs first.

TABLE 2—AFFECTED ECU P/Ns

1471M63P37	1471M63P38	1471M63P39	1471M63P40	1471M63P42
1519M89P27	1519M89P28	1519M89P29	1519M89P30	1519M89P32
1820M33P10	1820M33P11	1820M33P12	1820M33P13	1820M33P15
2121M25P01	2121M25P02	2121M26P01	2121M26P02	2121M29P01
2121M29P02	2121M37P03	2121M38P03	2121M41P03	

(h) Installation Prohibition

(1) After the effective date of this AD, do not install any ECU P/N listed in Table 1 of this AD onto any airplane.

(2) After the effective date of this AD, do not operate any airplane with more than one ECU P/N 2121M37P02, 2121M38P02, or 2121M41P02 installed.

(i) Alternative Methods of Compliance (AMOCs)

The Manager, Engine Certification Office, FAA, may approve AMOCs for this AD. Use the procedures in 14 CFR 39.19 to make your request.

(j) Related Information

For more information about this AD, contact Tomasz Rakowski, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; phone: 781-238-7735; fax: 781-238-7199; email: tomasz.rakowski@faa.gov.

(k) Material Incorporated by Reference

None.

Issued in Burlington, Massachusetts, on February 17, 2012.

Peter A. White,

Manager, Engine & Propeller Directorate, Aircraft Certification Service.

[FR Doc. 2012-4284 Filed 2-23-12; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2008-1245; Directorate Identifier 2008-NE-27-AD; Amendment 39-15912; AD 2009-11-02]

RIN 2120-AA64

Airworthiness Directives; CFM International S.A. Model CFM56 Turbofan Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule; correction.

SUMMARY: The FAA is correcting an airworthiness directive (AD) that published in the **Federal Register**. That AD applies to CFM International S.A. CFM56-2, CFM56-3, CFM56-5A, CFM56-5B, CFM56-5C, and CFM56-7B series turbofan engines with certain part number (P/N) and serial number (SN) high-pressure compressor (HPC) 4-9 spools installed. In Table 1 of the AD, the HPC 4-9 spool SN GWN05AMO in the 2nd column of the Table is incorrect. This document corrects that error. In all other respects, the original document remains the same.

DATES: This final rule is effective February 24, 2012. The effective date for AD 2009-11-02 (74 FR 23305, May 19, 2009) remains June 23, 2009.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD

docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Martin Adler, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; phone: 781-238-7157; fax: 781-238-7199; email: martin.adler@faa.gov.

SUPPLEMENTARY INFORMATION: Airworthiness Directive 2009-11-02, Amendment 39-15912 (74 FR 23305, May 19, 2009), currently requires removing certain HPC 4-9 spools listed by P/N and SN in the AD.

As published, in Table 1 of the AD, the HPC 4-9 spool SN GWN05AMO in the 2nd column of the Table is incorrect.

No other part of the preamble or regulatory information has been changed; therefore, only the changed portion of the final rule is being published in the **Federal Register**.

The effective date of this AD remains June 23, 2009.

Correction of Regulatory Text

§ 39.13 [Corrected]

■ In the **Federal Register** of May 19, 2009, on page 23306, in the 3rd column, in Table 1, under the HPC 4-9 Spool SN heading, in the twentieth line of AD