

(1) If the third fire extinguisher bottle is not installed, no further work is required by this paragraph.

(2) For Model GV airplanes in which the third fire extinguisher bottle is installed as a dedicated APU fire bottle configuration, as defined in Gulfstream V Alert Customer Bulletin 30A, dated December 20, 2010 (as a functioning part of the aircraft fire suppression system): Before further flight, revise the Limitations section of the Gulfstream GV AFM to include the information in Gulfstream GV/GV-SP AFM Supplement CE51 628M001, Revision A, dated December 20, 2010 (which is included in Gulfstream V Alert Customer Bulletin 30A, dated December 20, 2010). This AFM supplement adds restrictions for APU usage. Operate the airplane thereafter according to the limitations in this AFM supplement.

Note 1: This may be done by inserting a copy of Gulfstream GV/GV-SP AFM Supplement CE51 628M001, Revision A, dated December 20, 2010, in the applicable AFM. When information in this AFM supplement has been included in general revisions of the applicable AFM, the general revisions may be inserted in the applicable AFM, provided the relevant information in the general revision is identical to that in Gulfstream GV/GV-SP AFM Supplement CE51 628M001, Revision A, dated December 20, 2010, and that AFM supplement may be removed.

(3) For Model GV and GV-SP airplanes in which the third fire extinguisher bottle is installed as a spare fire bottle configuration (not connected to the airplane's electrical or fire suppression systems), as defined in the applicable Gulfstream alert customer bulletin identified in table 1 of this AD: Do the actions required by paragraph (g)(3)(i) or (g)(3)(ii) of this AD.

(i) Before further flight, remove the bottle, in accordance with the Accomplishment Instructions of the applicable Gulfstream alert customer bulletin identified in table 1 of this AD.

(ii) Before further flight, revise the limitations section of the applicable Gulfstream AFM specified in table 1 of this AD to include the information in Gulfstream GV/GV-SP AFM Supplement CE51 628M001, Revision A, dated December 20, 2010. This AFM supplement adds restrictions for APU usage. Operate the airplane thereafter according to the limitations in that AFM supplement.

Note 2: This may be done by inserting a copy of Gulfstream GV/GV-SP AFM Supplement CE51 628M001, Revision A, dated December 20, 2010, in the applicable AFM. When information in this AFM supplement has been included in general revisions of the applicable AFM, the general revisions may be inserted in the applicable AFM, provided the relevant information in the general revision is identical to that in Gulfstream GV/GV-SP AFM Supplement CE51 628M001, Revision A, dated December 20, 2010, and that AFM supplement may be removed.

Credit for Actions Accomplished in Accordance With Previous Service Information

(h) Actions accomplished before the effective date of this AD in accordance with Gulfstream V Alert Customer Bulletin 30 (for Model GV airplanes), dated December 6, 2010, including Gulfstream GV AFM Supplement CE51 628M001, dated November 18, 2010, to the Gulfstream GV AFM; or Gulfstream G550 (for Model GV-SP airplanes) or G500 (for Model GV-SP airplanes) Alert Customer Bulletin 10, both dated December 6, 2010; are acceptable for compliance with the corresponding actions required by paragraph (g) of this AD.

Parts Installation

(i) As of the effective date of this AD, no person may install a third fire extinguisher bottle in the APU fragment impact zone (rotor fragment impact zone) of any airplane.

No Reporting

(j) Although Gulfstream V Alert Customer Bulletin 30A (for Model GV airplanes), Gulfstream G500 Alert Customer Bulletin 10A (for Model GV-SP airplanes), and Gulfstream G550 Alert Customer Bulletin 10A (for Model GV-SP airplanes); all dated December 20, 2010, all including Gulfstream GV/GV-SP AFM Supplement CE51 628M001, Revision A, dated December 20, 2010, to the Gulfstream GV, and GV-SP AFMs; specify to submit certain information to the manufacturer, this AD does not include that requirement.

Special Flight Permit

(k) Special flight permits, as described in Section 21.197 and Section 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199), may be issued to operate the airplane to a location where the requirements of this AD can be accomplished, provided the following conditions are met:

(1) If an airplane is grounded due to a single generator failure, the APU may be operated during a ferry flight, provided no passengers are carried.

(2) Only the minimum required flight crew is allowed on any ferry flight.

Alternative Methods of Compliance (AMOCs)

(l)(1) The Manager, Atlanta Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in the Related Information section of this AD.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

Related Information

(m) For more information about this AD, contact Sanford Proveaux, Aerospace Engineer, Continued Operational Safety and

Certificate Management Branch, ACE-102A, FAA, Atlanta Aircraft Certification Office (ACO) 1701 Columbia Avenue, College Park, Georgia 30337; telephone 404-474-5566; fax 404-474-5606; sanford.proveaux@faa.gov.

(n) For service information identified in this AD, contact Gulfstream Aerospace Corporation, Technical Publications Dept., P.O. Box 2206, Savannah, Georgia 31402-2206; telephone 800-810-4853; fax 912-965-3520; e-mail pubs@gulfstream.com; Internet http://www.gulfstream.com/product_support/technical_pubs/pubs/index.htm. You may review copies of the referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221.

Issued in Renton, Washington on June 10, 2011.

Ali Bahrami,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2011-15537 Filed 6-21-11; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2011-0639; Directorate Identifier 2011-CE-016-AD]

RIN 2120-AA64

Airworthiness Directives; Piper Aircraft, Inc. Models PA-24, PA-24-250, and PA-24-260 Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to adopt a new airworthiness directive (AD) for the products listed above. This proposed AD would require either replacement of the stabilator horn assembly or repetitive inspection of the stabilator horn assembly for corrosion or cracks with replacement of the stabilator horn assembly if any corrosion or cracks are found. This proposed AD was prompted by reports of cracks developing in the stabilator horn assembly. We are proposing this AD to detect and correct corrosion or cracks in the stabilator horn assembly. Corrosion or cracks could lead to failure of the stabilator horn. Consequently, failure of the stabilator horn could lead to a loss of pitch control in flight.

DATES: We must receive comments on this proposed AD by August 8, 2011.

ADDRESSES: You may send comments by any of the following methods:

• *Federal eRulemaking Portal*: Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

• *Fax*: 202-493-2251.

• *Mail*: U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

• *Hand Delivery*: Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Piper Aircraft, Inc., 2926 Piper Drive, Vero Beach, Florida 32960; telephone: (772) 567-4361; fax: (772) 978-6573; Internet: <http://www.newpiper.com/company/publications.asp>. You may review copies of the referenced service information at the FAA, Small Airplane Directorate, 901 Locust St., Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329-4148.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (phone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT:

Gregory K. Noles, Aerospace Engineer, FAA, Atlanta Aircraft Certification Office, 1701 Columbia Avenue, College Park, Georgia 30337; phone: (404) 474-5551; fax: (404) 474-5606; e-mail: gregory.noles@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposal. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2011-0639; Directorate Identifier 2011-CE-016-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://>

www.regulations.gov, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

We issued Special Airworthiness Information Bulletin (SAIB) CE-04-88, dated September 15, 2004. This SAIB alerted owners and operators of Piper Aircraft, Inc. (Piper) Models PA-23, PA-24, PA-30, and PA-39 airplanes of potential corrosion of the stabilator torque tube, attach fittings, and attaching fasteners and recommended inspections of these parts. Based on the information available at issuance of this SAIB, the FAA had determined that an unsafe condition did not exist under 14 CFR part 39.

After reviewing service data for corrosion on the stabilator torque tubes, Piper issued Piper Service Bulletin No. 1160, dated December 26, 2005. This service information is for stabilator torque tube assembly inspection. We then received reports of cracks found in the stabilator horn, part number (P/N) 20397-00, during maintenance inspections per SAIB CE-04-88 or Service Bulletin 1160.

With FAA assistance, the National Institute for Aviation Research (NIAR) investigated and concluded the root cause of the stabilator horn cracking was stress corrosion.

We found two service difficulty reports for this safety issue. In parallel, the International Comanche Society (ICS) surveyed operators and provided additional service data. The ICS survey included approximately 80 targeted inspections and found 18 incidences of stabilator horn cracking, with all incidences occurring on Models PA-24 and PA-24-250 airplanes. The same configuration of horn and torque assembly exists on Model PA-24-260 airplanes.

This condition, if not corrected, could result in failure of the stabilator horn. Consequently, failure of the stabilator horn could lead to a loss of pitch control in flight.

Relevant Service Information

We reviewed Piper Aircraft, Inc. Service Bulletin No. 1189, dated April 29, 2010. The service information describes procedures for stabilator horn assembly inspection.

FAA's Determination

We are proposing this AD because we evaluated all the relevant information and determined the unsafe condition described previously is likely to exist or

develop in other products of the same type design.

Proposed AD Requirements

This proposed AD would require either replacement of the stabilator horn assembly or repetitive inspection of the stabilator horn assembly for corrosion or cracks with replacement of the stabilator horn assembly if any corrosion or cracks are found.

Differences Between the Proposed AD and the Service Information

The service information requires an initial inspection of the stabilator horn assembly upon reaching the initial 1,000 hours time-in-service (TIS), with a repetitive inspection every 100 hours TIS thereafter. After installation of a new stabilator horn assembly, the inspection cycle starts over with an initial inspection at 1,000 hours TIS since the new stabilator horn assembly was installed with the 100-hour TIS repetitive inspections thereafter.

This proposed AD requires either one of the following options: (1) An initial inspection of the stabilator horn assembly upon reaching 1,000 hours TIS or within 100 hours TIS after the effective date of the AD, whichever occurs later, with repetitive inspections every 500 hours TIS or 3 years, whichever occurs first; or (2) replacement of the stabilator horn assembly upon reaching 1,000 hours TIS or within the next 100 hours TIS after the effective date of this AD, whichever occurs later. After replacement of the stabilator horn assembly, within 1,000 hours TIS or 10 years, whichever occurs first, the stabilator horn assembly must be replaced or be initially inspected and start the inspection cycle in option 1.

The service information applies to Piper Models PA-24, PA-24-250, PA-24-260, PA-24-400, PA-30, and PA-39 airplanes. We only have service history on Models PA-24 and PA-24-250 airplanes.

While there is no service history of this unsafe condition on Model PA-24-260 airplanes, we are including it in the AD because it is an identical configuration to Models PA-24 and PA-24-250 airplanes for the horn and torque tube.

There is no service history of this unsafe condition on the Models PA-24-400, PA-30, and PA-39 airplanes, including inspections from the ICS operator survey. Also, these models have a thicker torque tube, which reduces clamp-up forces; clamp-up forces are a key factor of the stress corrosion cracking. Therefore, we are not including the Models PA-24-400,

PA-30, and PA-39 airplanes in the applicability of this AD.

Costs of Compliance

We estimate that this proposed AD affects 3,100 airplanes of U.S registry.

We estimate the following costs to comply with this proposed AD:

ESTIMATED COSTS

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Stabilator horn assembly inspection	12 work-hours × \$85 per hour = \$1,020	Not applicable	\$1,020	\$3,162,000

We estimate the following costs to do any necessary replacements that would

be required based on the results of the proposed inspection. We have no way of

determining the number of airplanes that might need this replacement:

ON-CONDITION COSTS

Action	Labor cost	Parts cost	Cost per product
Stabilator horn assembly replacement	12 work-hours × \$85 per hour = \$1,020		\$572
			\$1,592

According to the manufacturer, some of the costs of this proposed AD may be covered under warranty, thereby reducing the cost impact on affected individuals. We do not control warranty coverage for affected individuals. As a result, we have included all costs in our cost estimate.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: “General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

- (1) Is not a “significant regulatory action” under Executive Order 12866,
- (2) Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

Piper Aircraft, Inc.: Docket No. FAA-2011-0639, Directorate Identifier 2011-CE-016-AD.

Comments Due Date

- (a) We must receive comments by August 8, 2011.

Affected ADs

- (b) None.

Applicability

(c) This AD applies to the following Piper Aircraft, Inc. airplanes, certificated in any category:

- (1) Model PA-24, serial numbers (SNs) 24-1 through 24-3687;
- (2) Model PA-24-250, SNs 24-1, 24-103 through 24-3687; and
- (3) Model PA-24-260, SNs 24-3642 and 24-4000 through 24-5034.

Subject

(d) Joint Aircraft System Component (JASC)/Air Transport Association (ATA) of America Code 27: Flight Controls.

Unsafe Condition

(e) This AD was prompted by reports of cracks developing in the stabilator horn assembly. We are issuing this AD to detect and correct corrosion or cracks in the stabilator horn assembly. Corrosion or cracks could lead to failure of the stabilator horn. Consequently, failure of the stabilator horn could lead to a loss of pitch control in flight.

Compliance

(f) Comply with this AD following Piper Aircraft, Inc. Service Bulletin No. 1189, dated April 29, 2010, within the compliance times specified in this AD, unless already done (does not eliminate the repetitive actions of this AD).

Inspection/Replacement

(g) When the stabilator horn assembly reaches a total of 1,000 hours time-in-service (TIS) or within the next 100 hours TIS after the effective date of this AD, whichever occurs later, do either of the following actions:

- (1) Initially inspect the stabilator horn assembly for corrosion or cracks. Repetitively thereafter inspect at intervals not to exceed 500 hours TIS or 3 years, whichever occurs first; or
- (2) Replace the stabilator horn assembly with a new stabilator horn assembly. Repetitively thereafter replace the stabilator horn assembly with a new stabilator horn assembly within the next 1,000 hours TIS

after the last replacement or within the next 10 years after the last replacement, whichever occurs first.

(h) If any corrosion or cracks are found during any of the inspections required in paragraph (g)(1) of this AD, before further flight, you must replace the stabilator horn assembly with a new stabilator horn assembly. After the new stabilator horn assembly reaches a total of 1,000 hours TIS or within the next 10 years after the last replacement, whichever occurs first, you must do either of the actions required in paragraphs (g)(1) or (g)(2) of this AD.

(i) You may at any time replace the stabilator horn assembly with a new stabilator horn assembly, provided no corrosion or cracks were found during an inspection that would require replacement before further flight. After the new stabilator horn assembly reaches a total of 1,000 hours TIS or within the next 10 years after the last replacement, whichever occurs first, you must do either of the actions required in paragraphs (g)(1) or (g)(2) of this AD.

(j) If you replace the stabilator horn assembly as specified in paragraph (g)(2) of this AD, after the new stabilator horn assembly reaches a total of 1,000 hours TIS or within the next 10 years after the last replacement, whichever occurs first, you may begin the inspection requirements of paragraph (g)(1) instead of the repetitive replacement requirements of paragraph (g)(2).

Note: Piper Aircraft, Inc. Service Bulletin No. 1160, dated December 26, 2005; Special Airworthiness Information Bulletin CE-04-88, dated September 15, 2004; and AD 74-13-03, Amendment 39-2588 (41 FR 17371, April 26, 1976) are related to this AD action. For the attached torque tube, you may consider combining that inspection with the requirements of this AD.

Special Flight Permit

(k) Special flight permits are permitted with the following limitation: flight with known cracks is prohibited.

Alternative Methods of Compliance (AMOCs)

(l)(1) The Manager, Atlanta Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in the Related Information section of this AD.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

Related Information

(m) For more information about this AD, contact Gregory K. Noles, Aerospace Engineer, FAA, Atlanta ACO, 1701 Columbia Avenue, College Park, Georgia 30337; phone: (404) 474-5551; fax: (404) 474-5606; e-mail: gregory.noles@faa.gov.

(n) For service information identified in this AD, contact Piper Aircraft, Inc., 2926

Piper Drive, Vero Beach, Florida 32960; telephone: (772) 567-4361; fax: (772) 978-6573; Internet: <http://www.newpiper.com/company/publications.asp>. You may review copies of the referenced service information at the FAA, Small Airplane Directorate, 901 Locust St., Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329-4148.

Issued in Kansas City, Missouri on June 16, 2011.

John Colomy,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2011-15543 Filed 6-21-11; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2011-0569; Directorate Identifier 2010-NM-240-AD]

RIN 2120-AA64

Airworthiness Directives; BAE SYSTEMS (OPERATIONS) LIMITED Model BAe 146 and Avro 146-RJ Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to adopt a new airworthiness directive (AD) for the products listed above. This proposed AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

BAE Systems have received reports of in-service failure of the Main Landing Gear (MLG) shock absorber lower attachment pin.

This condition, if not detected and corrected, could lead to a MLG collapse on the ground or during landing and consequently damage to the aeroplane or injury to the occupants.

The proposed AD would require actions that are intended to address the unsafe condition described in the MCAI.

DATES: We must receive comments on this proposed AD by August 8, 2011.

ADDRESSES: You may send comments by any of the following methods:

- **Federal eRulemaking Portal:** Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- **Fax:** (202) 493-2251.
- **Mail:** U.S. Department of Transportation, Docket Operations, M-

30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

• **Hand Delivery:** U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-40, 1200 New Jersey Avenue, SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For BAE SYSTEMS (OPERATIONS) LIMITED service information identified in this proposed AD, contact BAE SYSTEMS (OPERATIONS) LIMITED, Customer Information Department, Prestwick International Airport, Ayrshire, KA9 2RW, Scotland, United Kingdom; telephone +44 1292 675207; fax +44 1292 675704; e-mail RApublications@baesystems.com; Internet <http://www.baesystems.com/Businesses/RegionalAircraft/index.htm>.

For Messier-Dowty service information identified in this proposed AD, contact Messier Services Americas, Customer Support Center, 45360 Severn Way, Sterling, Virginia 20166-8910; telephone 703-450-8233; fax 703-404-1621; Internet <https://techpubs.services/messier-dowty.com>.

You may review copies of the referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Operations office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone (800) 647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT:

Todd Thompson, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 227-1175; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No.