

Effective Date

(a) This airworthiness directive (AD) becomes effective May 7, 2010.

Affected ADs

(b) None.

Applicability

(c) This AD applies to Bombardier, Inc. Model BD-100-1A10 (Challenger 300) airplanes, certificated in any category, serial numbers 20002 through 20153 inclusive.

Subject

(d) Air Transport Association (ATA) of America Code 32: Landing gear.

Reason

(e) The mandatory continuing airworthiness information (MCAI) states:

There has been an incident during a production flight test where the proximity-sensor electronic unit (PSEU) failed. This resulted in unannounced loss of:

- Wheel brakes below 10 knots;
- Thrust reverser;
- Nose wheel steering; and
- Auto-deployment of the multi-function spoilers.

A similar condition, if not corrected, may result in reduced controllability of the aircraft upon landing and possible overrun of the runway.

The original issue of this directive mandated the introduction of non-normal procedures to the airplane flight manual (AFM) as an interim corrective action to address PSEU failures.

Revision 1 of this directive amends the aircraft applicability and introduces a note providing terminating action, for use at operator discretion, if the aircraft has incorporated a PSEU with software version 12 in accordance with Bombardier Service Bulletin (SB) 100-32-12.

Actions and Compliance

(f) Unless already done, within 14 days after the effective date of this AD: Revise the Limitations Section of the Bombardier Challenger 300 AFM, CSP 100-1, to include the information in Bombardier Temporary Revision TR-39, dated March 2, 2005, as specified in the temporary revision. This temporary revision introduces a procedure for "PROX SYS FAULT (A)" and modifies the "WOW FAIL (C)" and "GEAR SYS FAIL (C)" procedures.

Note 1: This may be done by inserting a copy of Bombardier Temporary Revision TR-39, dated March 2, 2005, in the AFM. When this temporary revision has been included in general revisions of the AFM, the general revisions may be inserted in the AFM, provided the relevant information in the general revision is identical to that in Bombardier Temporary Revision TR-39, dated March 2, 2005.

Note 2: If the aircraft has incorporated a PSEU, part number (P/N) 30227-0401, 30227-0402, or 30227-0403, with software version 12, installed in accordance with Bombardier Service Bulletin 100-32-12, dated June 4, 2007, it is permissible to follow the revised AFM procedures included in Bombardier Temporary Revision TR-46,

dated March 27, 2008, in lieu of using Bombardier Temporary Revision TR-39, dated March 2, 2005, specified in paragraph (f) of this AD.

FAA AD Differences

Note 3: This AD differs from the MCAI and/or service information as follows: No differences.

Other FAA AD Provisions

(g) The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, New York Aircraft Certification Office (ACO), ANE-170, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Program Manager, Continuing Operational Safety, FAA, New York ACO, 1600 Stewart Avenue, Suite 410, Westbury, New York 11590; telephone 516-228-7300; fax 516-794-5531. Before using any approved AMOC on any airplane to which the AMOC applies, notify your principal maintenance inspector (PMI) or principal avionics inspector (PAI), as appropriate, or lacking a principal inspector, your local Flight Standards District Office. The AMOC approval letter must specifically reference this AD.

(2) *Airworthy Product:* For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(3) *Reporting Requirements:* For any reporting requirement in this AD, under the provisions of the Paperwork Reduction Act (44 U.S.C. 3501 *et seq.*), the Office of Management and Budget (OMB) has approved the information collection requirements and has assigned OMB Control Number 2120-0056.

Related Information

(h) Refer to MCAI Transport Canada Civil Aviation Airworthiness Directive CF-2005-12R1, dated December 23, 2008; and Bombardier Temporary Revision TR-39, dated March 2, 2005; for related information.

Material Incorporated by Reference

(i) You must use Bombardier Temporary Revision TR-39, dated March 2, 2005, to the Bombardier Challenger 300 Airplane Flight Manual, CSP 100-1, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Bombardier, Inc., 400 Côte-Vertu Road West, Dorval, Québec H4S 1Y9, Canada; telephone 514-855-5000; fax 514-855-7401; e-mail thd.crj@aero.bombardier.com; Internet <http://www.bombardier.com>.

(3) You may review copies of the service information at the FAA, Transport Airplane

Directorate, 1601 Lind Avenue, SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221.

(4) You may also review copies of the service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Issued in Renton, Washington, on March 19, 2010.

Ali Bahrami,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2010-6785 Filed 4-1-10; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2010-0230; Directorate Identifier 2010-NM-071-AD; Amendment 39-16250; AD 2010-06-51]

RIN 2120-AA64

Airworthiness Directives; The Boeing Company Model 737-600, -700, -700C, -800, -900, and -900ER Series Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule; request for comments.

SUMMARY: This document publishes in the **Federal Register** an amendment adopting airworthiness directive (AD) 2010-06-51 that was sent previously to all known U.S. owners and operators of The Boeing Company Model 737-600, -700, -700C, -800, -900, and -900ER series airplanes by individual notices. This AD requires doing a detailed inspection of the inboard and outboard aft attach lugs of the left and right elevator control tab mechanisms for gaps between the swage ring and the aft attach lug, and between the spacer and the aft attach lug; trying to move or rotate the spacer using hand pressure; and replacing any discrepant elevator tab control mechanism, including performing the detailed inspection on the replacement part before and after installation. This AD is prompted by a report of failure of the aft attach lugs on the left elevator tab control mechanism, which resulted in severe elevator vibration. We are issuing this AD to detect and correct a loose bearing in the aft lug of the elevator tab control

mechanism, which could result in unwanted elevator and tab vibration. The consequent structural failure of the elevator or horizontal stabilizer could result in loss of aircraft control and structural integrity.

DATES: This AD becomes effective April 7, 2010 to all persons except those persons to whom it was made immediately effective by emergency AD 2010-06-51, issued March 12, 2010, which contained the requirements of this amendment.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in the AD as of April 7, 2010.

We must receive comments on this AD by May 17, 2010.

ADDRESSES: You may send comments by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- *Fax:* 202-493-2251.

- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, Washington 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; e-mail me.boecom@boeing.com; Internet <https://www.myboeingfleet.com>.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Kelly McGuckin, Aerospace Engineer, Systems and Equipment Branch, ANM-130S, FAA, Seattle ACO, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 917-6490; fax (425) 917-6590.

SUPPLEMENTARY INFORMATION: On March 12, 2010, we issued emergency AD 2010-06-51, which applies to all The Boeing Company Model 737-600, -700, -700C, -800, -900, and -900ER series airplanes.

Background

The FAA received a report of failure of the aft attach lugs on the left elevator tab control mechanism, which resulted in severe elevator vibration. The flightcrew diverted from the intended route and made an uneventful landing. Subsequent investigation revealed extensive damage to the elevator tab control system. Severe vibration in this attach point is suspected of allowing rapid wear of the joint, and resulted in failure of the attach lugs. This condition, if not corrected, could result in a loss of aircraft control and structural integrity.

Relevant Service Information

We have reviewed Boeing Alert Service Bulletin 737-27A1296, dated March 12, 2010. The service bulletin describes procedures for a detailed inspection to detect discrepancies of the inboard and outboard aft attach lugs of the elevator tab control mechanism. Discrepancies include movement or rotation of the spacer, and gaps between the swage ring and the aft attach lug or between the spacer and the aft attach lug. The service bulletin describes procedures for replacing any discrepant elevator tab control mechanism, including performing the detailed inspection on the replacement part before and after installation. For certain airplanes, the compliance time for the inspection is 12 or 30 days, depending on airplane line number, total accumulated flight cycles, and approval for operation under extended twin operations (ETOPS).

FAA's Determination and Requirements of This AD

Since the unsafe condition described is likely to exist or develop on other airplanes of the same type design, we issued emergency AD 2010-06-51 to detect and correct a loose bearing in the aft lug of the elevator tab control mechanism, which could result in unwanted elevator and tab vibration. The consequent structural failure of the elevator or horizontal stabilizer could result in loss of aircraft control and structural integrity. The AD requires accomplishing the actions specified in the service information previously described, except as described in "Differences Between this AD and the Service Bulletin." This AD also requires

reporting the inspection results to Boeing.

We found that immediate corrective action was required; therefore, notice and opportunity for prior public comment thereon were impracticable and contrary to the public interest, and good cause existed to make the AD effective immediately by individual notices issued on March 12, 2010, to all known U.S. owners and operators of The Boeing Company Model 737-600, -700, -700C, -800, -900, and -900ER series airplanes. These conditions still exist, and the AD is hereby published in the **Federal Register** as an amendment to section 39.13 of the Federal Aviation Regulations (14 CFR 39.13) to make it effective to all persons.

Differences Between This AD and the Service Bulletin

The effectivity of Boeing Alert Service Bulletin 737-27A1296, dated March 12, 2010, includes all Model 737-600, -700, -700C, -800, -900, and -900ER series airplanes. The inspection requirements of this AD, however, affect only those airplanes subject to a short compliance time (within 12 or 30 days). Because the suspect components may be installed as replacements on all airplanes subject to this AD, paragraph (l) of this AD requires that the part be inspected before and after installation. We may consider superseding this AD to apply the inspection requirements to the remaining airplanes, which would be subject to a longer compliance time that would allow enough time to provide notice and opportunity for prior public comment on the merits of the inspection for these airplanes.

Interim Action

This AD is considered to be interim action. The inspection reports that are required by this AD will enable the manufacturer to obtain better insight into the nature, cause, and extent of the issue, and eventually to develop final action to address the unsafe condition. Once final action has been identified, we might consider further rulemaking.

Comments Invited

This AD is a final rule that involves requirements affecting flight safety, and we did not provide you with notice and an opportunity to provide your comments before it becomes effective. However, we invite you to send any written data, views, or arguments about this AD. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2010-0230; Directorate Identifier 2010-NM-071-AD" at the beginning of your comments. We specifically invite

comments on the overall regulatory, economic, environmental, and energy aspects of this AD. We will consider all comments received by the closing date and may amend this AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this AD.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

The FAA has determined that this regulation is an emergency regulation that must be issued immediately to correct an unsafe condition in aircraft, and that it is not a "significant regulatory action" under Executive Order 12866. It has been determined further that this action involves an emergency regulation under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979). If this emergency regulation is later deemed significant under DOT Regulatory Policies and Procedures, we will prepare a final regulatory evaluation and place it in the AD Docket. See the **ADDRESSES** section for a location to examine the regulatory evaluation, if filed.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2010-06-51 The Boeing Company:

Amendment 39-16250. Docket No. FAA-2010-0230; Directorate Identifier 2010-NM-071-AD.

Effective Date

(a) This AD becomes effective April 7, 2010, to all persons except those persons to whom it was made immediately effective by emergency AD 2010-06-51, issued on March 12, 2010, which contained the requirements of this amendment.

Affected ADs

(b) None.

Applicability

(c) This AD applies to all The Boeing Company Model 737-600, -700, -700C, -800, -900, and -900ER series airplanes; certificated in any category.

Subject

(d) Air Transport Association (ATA) of America Code 27: Flight controls.

Unsafe Condition

(e) This AD results from a report of failure of the aft attach lugs on the left elevator tab control mechanism, which resulted in severe elevator vibration. The Federal Aviation Administration is issuing this AD to detect and correct a loose bearing in the aft lug of the elevator tab control mechanism, which could result in unwanted elevator and tab vibration. The consequent structural failure of the elevator or horizontal stabilizer could result in loss of aircraft control and structural integrity.

Compliance

(f) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

Inspection and Corrective Action

(g) For Groups 1, 2, and 3; and Group 4, Configuration 2; as identified in Boeing Alert Service Bulletin 737-27A1296, dated March 12, 2010: At the applicable time specified in paragraph 1.E. Compliance of Boeing Alert

Service Bulletin 737-27A1296, dated March 12, 2010, except as required by paragraph (i) of this AD, do a detailed inspection of the inboard and outboard aft attach lugs of the left and right elevator control tab mechanisms for gaps between the swage ring and the aft attach lug, and between the spacer and the aft attach lug; and try to move or rotate the spacer using hand pressure, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 737-27A1296, dated March 12, 2010.

(h) If, during accomplishment of the actions required by paragraph (g) of this AD, any gap is found between the swage ring and the aft attach lug, or between the spacer and the aft attach lug; or if the spacer moves or rotates: Before further flight, do the actions required by paragraphs (h)(1) and (h)(2) of this AD, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 737-27A1296, dated March 12, 2010.

(1) Inspect the replacement elevator tab control mechanism for discrepancies, as specified in paragraph (g) of this AD; and, if no discrepancy is found, install the replacement elevator tab control mechanism.

(2) Re-inspect the installed elevator tab control mechanism, as required by paragraph (g) of this AD.

Exception to Service Bulletin Specifications

(i) Where Boeing Alert Service Bulletin 737-27A1296, dated March 12, 2010, specifies a compliance time after the date of the original issue of the service bulletin, this AD requires compliance within the specified compliance time after the effective date of this AD.

Inspection Done According to Multi Operator Message (MOM)

(j) An inspection done before the effective date of this AD according to Boeing Multi Operator Message Number MOM-MOM-10-0159-01B, dated March 10, 2010, is considered acceptable for compliance with the corresponding inspection specified in paragraph (g) of this AD.

Reporting

(k) At the applicable time specified in paragraph (k)(1) or (k)(2) of this AD: Submit a report of the findings (both positive and negative) of the inspections required by paragraph (g) of this AD to Boeing Commercial Airplanes Group, Attention: Manager, Airline Support, e-mail: rse.boecom@boeing.com. The report must include the inspection results including a description of any discrepancies found, the airplane line number, and the number of flight cycles and flight hours accumulated on the airplane. Under the provisions of the Paperwork Reduction Act (44 U.S.C. 3501 *et seq.*), the Office of Management and Budget (OMB) has approved the information collection requirements contained in this AD and has assigned OMB Control Number 2120-0056.

(1) If the inspection was done on or after the effective date of this AD: Submit the report within 10 days after the inspection.

(2) If the inspection was done before the effective date of this AD: Submit the report

within 10 days after the effective date of this AD.

Parts Installation

(l) For all airplanes: As of the effective date of this AD, no person may install an elevator tab control mechanism, part number 251A2430-(), on any airplane, unless the mechanism has been inspected before and after installation, in accordance with the requirements of paragraph (g) of this AD, and no discrepancies have been found.

Special Flight Permit

(m) Special flight permits, as described in Section 21.197 and Section 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199), are not allowed.

Alternative Methods of Compliance (AMOCs)

(n)(1) The Manager, Seattle Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Kelly McGuckin, Aerospace Engineer, Systems and Equipment Branch, ANM-130S, FAA, Seattle ACO, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone 425-917-6490; fax 425-917-6590. Information may be e-mailed to 9-ANM-Seattle-ACO-AMOC-Requests@faa.gov.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your principal maintenance inspector (PMI) or principal avionics inspector (PAI), as appropriate, or lacking a principal inspector, your local Flight Standards District Office. The AMOC approval letter must specifically refer to this AD.

(3) An AMOC that provides an acceptable level of safety may be used for any repair required by this AD if it is approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

Material Incorporated by Reference

(o) You must use Boeing Alert Service Bulletin 737-27A1296, dated March 12, 2010, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, Washington 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; e-mail me.boecom@boeing.com; Internet <https://www.myboeingfleet.com>.

(3) You may review copies of the service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington. For information on the

availability of this material at the FAA, call 425-227-1221.

(4) You may also review copies of the service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Issued in Renton, Washington, on March 18, 2010.

Ali Bahrami,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2010-6786 Filed 4-1-10; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2009-0684; Directorate Identifier 2008-NM-149-AD; Amendment 39-16247; AD 2010-07-03]

RIN 2120-AA64

Airworthiness Directives; The Boeing Company Model 747-200C and -200F Series Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: The FAA is superseding an existing airworthiness directive (AD), which applies to certain Model 747-200C and -200F series airplanes. That AD currently requires repetitive inspections to find fatigue cracking in the floor panel attachment fastener holes of the upper chord of certain upper deck floor beams in Section 41 (*i.e.*, body station 520 and forward), and repair if necessary. The existing AD also provides optional modifications, which extend the threshold for initiating certain repetitive inspections. This new AD requires additional repetitive inspections to find fatigue cracking in the floor panel attachment fastener holes of the upper chord of certain other upper deck floor beams in Section 41 and Section 42 (*i.e.*, aft of body station 520); repetitive inspections to find fatigue cracking in the permanent fastener holes of the upper chord of certain upper deck floor beams in Section 41; and related investigative and corrective actions. This new AD also provides a new optional modification, which terminates certain repetitive inspections. This AD results from new reports of cracking in the upper chord

of the upper deck floor beams in Sections 41 and 42, and new analysis that shows the permanent fastener holes of the upper chord of certain upper deck floor beams in Section 41 are also susceptible to fatigue cracking. We are issuing this AD to detect and correct cracking in the upper chord of the upper deck floor beams. Such cracking could extend and sever the floor beams, which could result in rapid decompression and loss of controllability of the airplane.

DATES: This AD becomes effective May 7, 2010.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in the AD as of May 7, 2010.

ADDRESSES: For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, Washington 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; e-mail me.boecom@boeing.com; Internet <https://www.myboeingfleet.com>.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (telephone 800-647-5527) is the Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Ivan Li, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 917-6437; fax (425) 917-6590.

SUPPLEMENTARY INFORMATION:

Discussion

The FAA issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that supersedes AD 2006-08-02, amendment 39-14556 (70 FR 18618, April 12, 2006). The existing AD applies to certain Model 747-200C and -200F series airplanes. That NPRM was published in the **Federal Register** on August 12, 2009 (74 FR 40529). That NPRM proposed to continue to require repetitive inspections to find fatigue cracking in the floor panel attachment fastener