

TABLE 4—DOCUMENTS INCORPORATED BY REFERENCE FOR ACTIONS REQUIRED BY THIS AD

Service bulletin	Date
Fokker Service Bulletin SBF50–24–031	January 29, 2008.
Fokker Service Bulletin SBF100–24–041	January 29, 2008.
Goodrich Power Systems Service Bulletin SG02206–24–01	March 4, 2008.

TABLE 5—DOCUMENTS INCORPORATED BY REFERENCE FOR OPTIONAL ACTIONS SPECIFIED IN THIS AD

Service bulletin	Date
Fokker Service Bulletin SBF50–24–030 including the drawings identified in Table 2 of this AD	November 6, 2003.
Fokker Service Bulletin SBF100–24–037, including Manual Change Notification—Maintenance Documentation MCNM F100–076, dated October 2, 2003, and including the drawings identified in Table 3 of this AD.	October 2, 2003.

Issued in Renton, Washington, on August 17, 2009.

Ali Bahrami,

*Manager, Transport Airplane Directorate,
Aircraft Certification Service.*

[FR Doc. E9–20576 Filed 8–26–09; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2009–0477; Directorate Identifier 2008–NM–191–AD; Amendment 39–16003; AD 2009–18–07]

RIN 2120–AA64

Airworthiness Directives; Boeing Model 747–100, –100B, –100B SUD, –200B, and –300 Series Airplanes; and Model 747SP and 747SR Series Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: The FAA is superseding an existing airworthiness directive (AD), which applies to certain Boeing Model 747 series airplanes. That AD currently requires repetitive inspections to detect cracks in various areas of the fuselage internal structure, and related investigative/corrective actions if necessary. This new AD requires additional repetitive inspections for cracking of certain fuselage structure, and related investigative/corrective actions if necessary. This AD results from fatigue tests and analysis by Boeing that identified areas of the fuselage where fatigue cracks can occur.

We are issuing this AD to prevent the loss of the structural integrity of the fuselage, which could result in rapid depressurization of the airplane.

DATES: This AD becomes effective October 1, 2009.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in the AD as of October 1, 2009.

ADDRESSES: For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H–65, Seattle, Washington 98124–2207; telephone 206–544–5000, extension 1, fax 206–766–5680; e-mail me.boecom@boeing.com; Internet <https://www.myboeingfleet.com>.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (telephone 800–647–5527) is the Document Management Facility, U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Ivan Li, Aerospace Engineer, Airframe Branch, ANM–120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98057–3356; telephone (425) 917–6437; fax (425) 917–6590.

SUPPLEMENTARY INFORMATION:

Discussion

The FAA issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that supersedes AD 2005–20–30, amendment 39–14327 (70 FR 59252, October 12, 2005). The existing AD applies to certain Boeing Model 747 series airplanes. That NPRM was published in the **Federal Register** on May 26, 2009 (74 FR 24712). That NPRM proposed to continue to require repetitive inspections to detect cracks in various areas of the fuselage internal structure, and related investigative/corrective actions if necessary. That NPRM also proposed to require additional repetitive inspections for cracking of certain fuselage structure, and related investigative/corrective actions if necessary.

Comments

We provided the public the opportunity to participate in the development of this AD. We have considered the comment that has been received on the NPRM. Boeing concurs with the NPRM.

Conclusion

We have carefully reviewed the available data, including the comment that has been received, and determined that air safety and the public interest require adopting the AD as proposed.

Costs of Compliance

There are about 209 airplanes of the affected design in the worldwide fleet. The following table provides the estimated costs for U.S. operators to comply with this AD. The average labor rate is \$80 per work hour.

TABLE—ESTIMATED COSTS

Action	Work hours	Cost per air-plane, per in-spection cycle	Number of U.S.-registered airplanes	Fleet cost
Inspections (required by AD 2005–20–30)	130	\$10,400	69	\$717,600
Additional inspections in Area 1 (new action)	6	480	69	33,120
Additional inspections in Area 6 (new action)	1	80	69	5,520

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in subtitle VII, part A, subpart III, Section 44701, “General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a “significant regulatory action” under Executive Order 12866;
- (2) Is not a “significant rule” under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this AD and placed it in the AD docket. See the ADDRESSES section for a location to examine the regulatory evaluation.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The Federal Aviation Administration (FAA) amends § 39.13 by removing amendment 39–14327 (70 FR 59252, October 12, 2005) and by adding the following new airworthiness directive (AD):

2009–18–07 Boeing: Amendment 39–16003. Docket No. FAA–2009–0477; Directorate Identifier 2008–NM–191–AD.

Effective Date

(a) This AD becomes effective October 1, 2009.

Affected ADs

(b) This AD supersedes AD 2005–20–30.

Applicability

(c) This AD applies to Boeing Model 747–100, 747–100B, 747–100B SUD, 747–200B, 747–300, 747SP, and 747SR series airplanes; certificated in any category; identified in Boeing Service Bulletin 747–53A2349, Revision 3, dated October 2, 2008.

Subject

(d) Air Transport Association (ATA) of America Code 53: Fuselage.

Unsafe Condition

(e) This AD results from fatigue tests and analysis by Boeing that identified areas of the fuselage where fatigue cracks can occur. We are issuing this AD to prevent the loss of the structural integrity of the fuselage, which could result in rapid depressurization of the airplane.

Compliance

(f) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

Restatement of Requirements of AD 2002–10–10 With Revised Service Information (Excluding Upper Deck Floor Beams)

Repetitive Inspections

(g) Prior to the accumulation of 22,000 total flight cycles, or within 1,000 flight cycles after June 11, 1993 (the effective date of AD 93–08–12, amendment 39–8559, which was superseded by AD 2002–10–10), whichever occurs later, unless accomplished previously within the last 2,000 flight cycles; and thereafter at intervals not to exceed 3,000 flight cycles: Perform an internal detailed inspection to detect cracks in the areas of the fuselage internal structure specified in paragraphs (g)(1) through (g)(6) of this AD; in accordance with Boeing Service Bulletin 747–53–2349, dated June 27, 1991; Boeing Alert Service Bulletin 747–53A2349, Revision 1, dated October 12, 2000; Boeing Service Bulletin 747–53A2349, Revision 2, dated April 3, 2003; or Boeing Alert Service Bulletin 747–53A2349, Revision 3, dated October 2, 2008. After the effective date of this AD, only Revision 3 of Boeing Alert Service Bulletin 747–53A2349 may be used. Continue doing the inspections until the inspections required by paragraph (j) of this AD are done.

- (1) Section 42 upper lobe frames.
- (2) Section 46 lower lobe frames.
- (3) Section 42 lower lobe frames.
- (4) Main entry door cutouts.
- (5) Section 41 body station 260, 340, and 400 bulkheads.
- (6) Main entry doors.

Note 1: For the purposes of this AD, a detailed inspection is: “An intensive examination of a specific item, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at an intensity deemed appropriate. Inspection aids such as mirror, magnifying lenses, etc., may be necessary. Surface cleaning and elaborate procedures may be required.”

(h) Prior to the accumulation of 25,000 total flight cycles, or within 1,000 flight cycles after June 11, 1993, whichever is later, unless already done within the last 2,000 flight cycles; and thereafter at intervals not to exceed 3,000 flight cycles: Do an internal detailed inspection to detect cracks in the Section 46 upper lobe frames, in accordance with Boeing Service Bulletin 747–53–2349, dated June 27, 1991; Boeing Alert Service Bulletin 747–53A2349, Revision 1, dated October 12, 2000; Boeing Service Bulletin 747–53A2349, Revision 2, dated April 3, 2003; or Boeing Alert Service Bulletin 747–53A2349, Revision 3, dated October 2, 2008. After the effective date of this AD, only

Revision 3 of Boeing Alert Service Bulletin 747-53A2349 may be used.

Repair of Cracks Detected During Paragraph (g) or (h) Inspections

(i) Before further flight, repair any cracks detected during the inspections done per paragraph (g) or (h) of this AD by doing the actions specified in paragraph (i)(1) or (i)(2) of this AD, as applicable.

(1) Repair in accordance with a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA; or using a method approved in accordance with paragraph (p) of this AD.

(2) Repair in accordance with Boeing Service Bulletin 747-53A2349, Revision 2, dated April 3, 2003; or Boeing Alert Service Bulletin 747-53A2349, Revision 3, dated October 2, 2008. After the effective date of this AD, only Revision 3 of Boeing Alert Service Bulletin 747-53A2349 may be used. Where either revision of the service bulletin specifies to contact Boeing for repair instructions, repair in accordance with a method approved by the Manager, Seattle ACO; or use a method approved in accordance with paragraph (p) of this AD.

Restatement of Requirements of AD 2005-20-30 With Revised Service Information

Repetitive Inspections

(j) Do an internal detailed inspection to detect cracking in the areas of the fuselage internal structure specified in paragraphs (j)(1), (j)(2), and (j)(3) of this AD, and internal and external detailed inspections of the areas specified in paragraphs (j)(4), (j)(5), (j)(6), and (j)(7) of this AD. Do the inspections in accordance with Boeing Service Bulletin 747-53A2349, Revision 2, dated April 3, 2003; or Boeing Alert Service Bulletin 747-53A2349, Revision 3, dated October 2, 2008. After the effective date of this AD, only Revision 3 of Boeing Alert Service Bulletin

747-53A2349 may be used. Do the inspections at the applicable time specified in paragraph (k) of this AD. Accomplishment of these inspections terminates the requirements of paragraph (g) of this AD.

(1) Section 42 upper lobe frames.

(2) Section 46 lower lobe frames.

(3) Section 42 lower lobe frames.

(4) Main entry door cutouts.

(5) Nose wheel well bulkheads, sidewall panels, and station (STA) 360 and 380 floor beams. These areas include Section 41 body station 260, 340, and 400 bulkheads.

(6) Main entry doors.

(7) Main electronics bay access door cutout.

(k) Do the inspections required by paragraph (j) of this AD at the applicable time specified in paragraph (k)(1), (k)(2), or (k)(3) of this AD. Repeat the inspections thereafter at intervals not to exceed 3,000 flight cycles.

(1) For airplanes on which the inspections required by paragraphs (g)(1), (g)(2), (g)(3), (g)(4), and (g)(6) of this AD have been done before November 16, 2005 (the effective date of AD 2005-20-30), but the inspections required by paragraphs (j)(5) and (j)(7) of this AD have not been done: Within 3,000 flight cycles since accomplishment of the most recent inspection required by paragraphs (g)(1), (g)(2), (g)(3), (g)(4), and (g)(6) of this AD, except that the inspections specified in paragraphs (j)(5) and (j)(7) of this AD may be done within 3,000 flight cycles since accomplishment of the most recent inspection required by paragraphs (g)(1), (g)(2), (g)(3), (g)(4), and (g)(6) of this AD, or within 1,000 flight cycles after November 16, 2005, whichever is later.

(2) For airplanes on which the inspections required by paragraphs (j)(5) and (j)(7) have been done before November 16, 2005: Within 3,000 flight cycles since accomplishment of the most recent inspection required by paragraphs (j)(5) and (j)(7) of this AD, or

within 1,000 flight cycles after November 16, 2005, whichever is later.

(3) For airplanes on which the inspections required by paragraph (g) of this AD have not been done before November 16, 2005: Prior to the accumulation of 22,000 total flight cycles, or within 1,000 flight cycles after November 16, 2005, whichever is later.

Repair of Cracks Detected During Paragraph (j) Inspection

(l) Before further flight, repair any cracking found during any inspection required by paragraph (j) of this AD in accordance with Boeing Service Bulletin 747-53A2349, Revision 2, dated April 3, 2003; or Boeing Alert Service Bulletin 747-53A2349, Revision 3, dated October 2, 2008. After the effective date of this AD, only Revision 3 of Boeing Alert Service Bulletin 747-53A2349 may be used. Where any revision of the service bulletin specifies to contact Boeing for repair instructions, repair in accordance with a method approved by the Manager, Seattle ACO; or use a method approved in accordance with paragraph (p) of this AD.

New Requirements of This AD

Inspections and Repair

(m) Do initial and repetitive detailed inspections for cracking in the areas specified in Table 1 of this AD using applicable internal and external detailed inspection methods; and repair all cracks, by doing all the applicable actions in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 747-53A2349, Revision 3, dated October 2, 2008, except as required by paragraph (n) of this AD. Do the initial and repetitive inspections at the times specified in paragraph 1.E., "Compliance," of the service bulletin, except as required by paragraph (o) of this AD. Repair all cracks before further flight after detection.

TABLE 1—ADDITIONAL INSPECTIONS

Inspect the addition portion of area 1 and area 6 as specified in Boeing Alert Service Bulletin 747-53A2349, Revision 3, dated October 2, 2008 ("the service bulletin")—	For airplanes identified as these groups in the service bulletin—
<i>In Area 1:</i> Fuselage frames at body stations 260–520 in areas where the upper deck floor beams are attached (Figure 11 of the Accomplishments Instructions of the service bulletin).	1 through 7 inclusive.
<i>In Area 6:</i> Fuselage frames at body stations 400–500 in areas above the Main Entry Door 1 cutouts, from the upper chord of the upper deck floor beams to Stringer 8 (Figure 12 of the Accomplishment Instructions of the service bulletin).	6 and 7.

Exceptions to Certain Procedures

(n) If any crack is found during any inspection required by paragraph (m) of this AD, and Boeing Alert Service Bulletin 747-53A2349, Revision 3, dated October 2, 2008, specifies to contact Boeing for appropriate action: Before further flight, repair the crack using a method approved in accordance with the procedures specified in paragraph (p) of this AD.

(o) Where Boeing Alert Service Bulletin 747-53A2349, Revision 3, dated October 2, 2008, specifies a compliance time after the date on Boeing Alert Service Bulletin 747-53A2349, Revision 3, dated October 2, 2008, this AD requires compliance within the

specified compliance time after the effective date of this AD.

Alternative Methods of Compliance (AMOCs)

(p)(1) The Manager, Seattle ACO, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Ivan Li, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 917-6437; fax (425) 917-6590. Or, e-mail information to 9-ANM-Seattle-ACO-AMOC-Requests@faa.gov.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your principal maintenance inspector (PMI) or principal avionics inspector (PAI), as appropriate, or lacking a principal inspector, your local Flight Standards District Office. The AMOC approval letter must specifically reference this AD.

(3) AMOCs approved previously in accordance with AD 2005-20-30 are approved as AMOCs with the corresponding provisions of this AD.

(4) An AMOC that provides an acceptable level of safety may be used for any repair

required by this AD, if it is approved by an Authorized Representative for the Boeing Commercial Airplanes Delegation Option Authorization Organization who has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

Material Incorporated by Reference

(q) You must use Boeing Alert Service Bulletin 747-53A2349, Revision 3, dated October 2, 2008, as applicable, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, Washington 98124-2207; telephone 206-544-5000, extension 1, fax 206-766-5680; e-mail me.boecom@boeing.com; Internet <https://www.myboeingfleet.com>.

(3) You may review copies of the service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221 or 425-227-1152.

(4) You may also review copies of the service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Issued in Renton, Washington, on August 17, 2009.

Ali Bahrami,

Manager, Transport Airplane Directorate,
Aircraft Certification Service.

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2009-0386; Directorate Identifier 2008-NM-184-AD; Amendment 39-16002; AD 2009-18-06]

RIN 2120-AA64

Airworthiness Directives; Construcciones Aeronauticas, S.A. (CASA), Model CN-235, CN-235-100, CN-235-200, and CN-235-300 Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

During operation in icing conditions, an asymmetric configuration of the de-icing boots was detected, occurring during the inflation and deflation check of the de-icing system. This was found to be due to an unexpected failure mode in the pneumatic and de-icing system's control electronic logic. This condition, if not corrected, could affect the de-icing capabilities of the boots installed on the wing and horizontal stabilizers, potentially leading to loss of control of the aircraft.

* * * * *

We are issuing this AD to require actions to correct the unsafe condition on these products.

DATES: This AD becomes effective October 1, 2009.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of October 1, 2009.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC.

FOR FURTHER INFORMATION CONTACT: Shahram Daneshmandi, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 227-1112; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM was published in the **Federal Register** on April 29, 2009 (74 FR 19460). That NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

During operation in icing conditions, an asymmetric configuration of the de-icing boots was detected, occurring during the inflation and deflation check of the de-icing system. This was found to be due to an unexpected failure mode in the pneumatic and de-icing system's control electronic logic. This condition, if not corrected, could affect the de-icing capabilities of the boots installed on the wing and horizontal

stabilizers, potentially leading to loss of control of the aircraft.

To address and correct this unsafe condition, EADS-CASA developed modification 31558, approved by DGAC-Spain and incorporated into the Type Design Definition through the approval of CN-235-300 version AE02, revision 14 of Spanish Type Certificate DGAC 01/86, dated 22 March 2002, and modification 31607, Minor Change approved by EADS-CASA under their DOA 21J.032 privileges, complementary to modification 31558. The entire modification package consists of an improvement of the de-icing boots electronic control system, making it capable of detecting all possible boot configurations on wings and horizontal stabilizers without affecting pneumatic system functions. The instructions for the in-service accomplishment of this modification have been published as CN-235 Service Bulletin (SB) 235-30-16 dated 21 January 2005.

For the reasons described above, this EASA AD requires the modification of the De-Icing Boots control system in all aircraft that have not yet implemented the modification.

You may obtain further information by examining the MCAI in the AD docket.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM or on the determination of the cost to the public.

Change to Parts Cost

We have revised the parts cost to reflect the price of two kits from the manufacturer. The revised cost is less than the original cost presented in the NPRM.

Conclusion

We reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed.

Differences Between This AD and the MCAI or Service Information

We have reviewed the MCAI and related service information and, in general, agree with their substance. But we might have found it necessary to use different words from those in the MCAI to ensure the AD is clear for U.S. operators and is enforceable. In making these changes, we do not intend to differ substantively from the information provided in the MCAI and related service information.

We might also have required different actions in this AD from those in the MCAI in order to follow our FAA policies. Any such differences are highlighted in a NOTE within the AD.

Costs of Compliance

We estimate that this AD will affect 8 products of U.S. registry. We also