

publish a regulatory flexibility analysis if the agency certifies that the final rule will not have a significant economic impact on a substantial number of small entities. 5 U.S.C. 605(b).

Pursuant to section 605(b), the Board certifies that this interim final rule will not have a significant economic impact on a substantial number of small entities. The rule increases the interest paid on certain balances held by eligible institutions at the Federal Reserve Banks and will benefit all institutions, small and large, that receive such interest. There are no new reporting, record-keeping, or other compliance requirements associated with this rule.

Paperwork Reduction Act

In accordance with the Paperwork Reduction Act (44 U.S.C. 3506; 5 CFR 1320 Appendix A.1), the Board has reviewed the interim final rule under authority delegated to the Board by the Office of Management and Budget. The rule contains no collections of information pursuant to the Paperwork Reduction Act.

List of Subjects in 12 CFR Part 204

Banks, Banking, Reporting and recordkeeping requirements.

Authority and Issuance

■ For the reasons set forth in the preamble, the Board is amending 12 CFR part 204 as follows:

PART 204—RESERVE REQUIREMENTS OF DEPOSITORY INSTITUTIONS (REGULATION D)

■ 1. The authority citation for part 204 continues to read as follows:

Authority: 12 U.S.C. 248(a), 248(c), 371a, 461, 601, 611, and 3105.

■ 2. In § 204.10, paragraphs (b)(1) and (b)(2) are revised to read as follows:

§ 204.10 Payment of interest on balances.

* * * * *

(b) * * *

(1) For required reserve balances, at the average targeted federal funds rate over the reserve maintenance period; and

(2) For excess balances, at the lowest targeted federal funds rate during the reserve maintenance period.

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By order of the Board of Governors of the Federal Reserve System, November 5, 2008.

Robert deV. Frierson,

Deputy Secretary of the Board.

[FR Doc. E8-26727 Filed 11-14-08; 8:45 am]

BILLING CODE 6210-01-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2008-0492; Directorate Identifier 2008-CE-023-AD; Amendment 39-15734; AD 2008-23-13]

RIN 2120-AA64

Airworthiness Directives; Hawker Beechcraft Corporation Model 390 Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for certain Hawker Beechcraft Corporation (HBC) Model 390 airplanes. This AD requires you to remove the current preformed packing, elbow fitting, and jam nut from the left and right hydraulic pump pressure output port and replace with new parts. This AD also requires you to install a hydraulic pump case drain check valve. This AD results from nine occurrences of hydraulic fluid leaking from the engine hydraulic pump output fitting as a result of an improperly installed elbow connecting the output port to the pulse dampener hose. We are issuing this AD to prevent hydraulic fluid leaks from the left and right hydraulic fluid pump and to prevent the flow of hydraulic fluid into the engine compartment. The loss of hydraulic fluid can result in loss of airplane hydraulic system pressure and the consequent loss of hydraulic system functions including gear extension/retraction, spoiler functions, and anti-skid braking system actuation. The inability of the hydraulic installation to isolate flow of hydraulic fluid could result in a hazardous amount of flammable fluid in the corresponding engine compartment. These conditions, if not corrected, could result in loss of system functions and/or fire in the engine compartment.

DATES: This AD becomes effective on December 22, 2008.

On December 22, 2008, the Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD.

ADDRESSES: For service information identified in this AD, contact Hawker Beechcraft Corporation, 9709 East Central, Wichita, Kansas 67201; telephone: (316) 676-5034; fax: (316)

676-6614; Internet: https://www.hawkerbeechcraft.com/service_support/pubs/.

To view the AD docket, go to U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590, or on the Internet at <http://www.regulations.gov>. The docket number is FAA-2008-0492; Directorate Identifier 2008-CE-023-AD.

FOR FURTHER INFORMATION CONTACT:

Aaron Waters, Aerospace Engineer, 1801 Airport Road, Room 100, Wichita, Kansas 67209; telephone: (316) 946-4174; fax: (316) 946-4107.

SUPPLEMENTARY INFORMATION:

Discussion

On April 24, 2008, we issued a proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an AD that would apply to certain Model 390 airplanes. This proposal was published in the **Federal Register** as a notice of proposed rulemaking (NPRM) on May 1, 2008 (73 FR 23988). The NPRM proposed to require you to remove the current preformed packing, elbow fitting, and jam nut from the left and right hydraulic pump pressure output port and replace with new parts. The NPRM also proposed to require you to install a hydraulic pump case drain check valve.

Comments

We provided the public the opportunity to participate in developing this AD. We received no comments on the proposal or on the determination of the cost to the public.

Conclusion

We have carefully reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed except for minor editorial corrections. We have determined that these minor corrections:

- Are consistent with the intent that was proposed in the NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

Costs of Compliance

We estimate that this AD affects 182 airplanes in the U.S. registry.

We estimate the following costs to do the modifications:

Labor cost	Parts cost	Total cost per airplane	Total cost on U.S. operators
Hydraulic pressure output fitting assembly: 8 work-hours × \$80 per hour = \$640	\$100	\$740	\$134,680
Hydraulic pump case drain check valve: 16 work-hours × \$80 per hour = \$1,280	4,353	5,633	1,025,206

HBC will provide warranty credit as specified in Hawker Beechcraft Mandatory Service Bulletin SB 29–3869, dated January 2008; and Hawker Beechcraft Mandatory Service Bulletin SB 29–3851, dated January 2008.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, Section 106 describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the agency’s authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, “General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this AD.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- 1. Is not a “significant regulatory action” under Executive Order 12866;
- 2. Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
- 3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a summary of the costs to comply with this AD (and other information as included in the Regulatory Evaluation) and placed it in the AD Docket. You may get a copy of this summary by sending a request to us at the address listed under **ADDRESSES**. Include “Docket No. FAA–2008–0492; Directorate Identifier 2008–CE–023–AD” in your request.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, under the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. FAA amends § 39.13 by adding the following new AD:

2008–23–13 Hawker Beechcraft

Corporation: Amendment 39–15734; Docket No. FAA–2008–0492; Directorate Identifier 2008–CE–023–AD.

Effective Date

(a) This AD becomes effective on December 22, 2008.

Affected ADs

(b) None.

Applicability

(c) This AD applies to Model 390 airplanes, serial numbers RB–4 through RB–224, that are certificated in any category.

Unsafe Condition

(d) This AD results from nine occurrences of hydraulic fluid leaking from the engine hydraulic pump output fitting as a result of an improperly installed elbow connecting the output port to the pulse dampener hose. We are issuing this AD to prevent hydraulic fluid leaking from the left and right hydraulic fluid pump and to prevent the flow of hydraulic fluid into the engine compartment. The loss of hydraulic fluid can result in loss of airplane hydraulic system pressure and the consequent loss of hydraulic system functions including gear extension/retraction, spoiler functions, and anti-skid braking system actuation. The inability of the hydraulic installation to isolate flow of hydraulic fluid could result in a hazardous amount of flammable fluid in the corresponding engine compartment. These conditions, if not corrected, could result in loss of system functions and/or fire in the engine compartment.

Compliance

(e) To address this problem, you must do the following, unless already done:

Actions	Compliance	Procedures
(1) Remove current preformed packing (part number (P/N) MS28778–6), elbow fitting (P/N MS21908J6), and jam nut (P/N AN924–6J) from the left and right hydraulic pump pressure output port and install new preformed packing (P/N MS28778–6), union (P/N MS21902J6), and swivel fitting (P/N NAS1762J0606) in the left and right hydraulic pressure pump output port.	Within the next 200 hours time-in-service (TIS) after December 22, 2008 (the effective date of this AD) or within the next 6 months after December 22, 2008 (the effective date of this AD), whichever occurs first.	Follow Hawker Beechcraft Mandatory Service Bulletin SB 29–3869, dated January 2008.
(2) Install hydraulic pump case drain check valve Kit No. 390–5803–0001.	Within the next 200 hours TIS after December 22, 2008 (the effective date of this AD) or within the next 6 months after December 22, 2008 (the effective date of this AD), whichever occurs first.	Hawker Beechcraft Mandatory Service Bulletin SB 29–3851, dated January 2008.

Alternative Methods of Compliance (AMOCs)

(f) The Manager, Wichita Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Aaron Waters, Aerospace Engineer, Wichita ACO, 1801 Airport Road, Room 100, Wichita, Kansas 67209; telephone: (316) 946-4174; fax: (316) 946-4107. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

Material Incorporated by Reference

(g) You must use Hawker Beechcraft Mandatory Service Bulletin SB 29-3869, dated January 2008; and Hawker Beechcraft Mandatory Service Bulletin SB 29-3851, dated January 2008, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Hawker Beechcraft Corporation, 9709 East Central, Wichita, Kansas 67201; telephone: (316) 676-5034; fax: (316) 676-6614; Internet: https://www.hawkerbeechcraft.com/service_support/pubs/.

(3) You may review copies at the FAA, Central Region, Office of the Regional Counsel, 901 Locust, Kansas City, Missouri 64106; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Issued in Kansas City, Missouri, on November 5, 2008.

Patrick R. Mullen,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. E8-26879 Filed 11-14-08; 8:45 am]

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DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2008-0115; Directorate Identifier 2007-NM-240-AD; Amendment 39-15723; AD 2008-23-02]

RIN 2120-AA64

Airworthiness Directives; Saab Model SAAB 2000 Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

One Part Number (P/N) LM-219-92 Centre Bracket from a P/N LM-219-SA28 Aft Engine Mounting assembly was found to be cracked while installed on the aircraft.

This reduces the effectiveness of the mounting assembly and could eventually cause it to fail.

* * * * *

A failed mounting assembly, if not corrected, could result in loss of the engine. We are issuing this AD to require actions to correct the unsafe condition on these products.

DATES: This AD becomes effective December 22, 2008.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of December 22, 2008.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC.

FOR FURTHER INFORMATION CONTACT: Shahram Daneshmandi, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 227-1112; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION:**Discussion**

We issued a supplemental notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That supplemental NPRM was published in the *Federal Register* on August 29, 2008 (73 FR 50903). That supplemental NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

One Part Number (P/N) LM-219-92 Centre Bracket from a P/N LM-219-SA28 Aft Engine Mounting assembly was found to be cracked while installed on the aircraft.

This reduces the effectiveness of the mounting assembly and could eventually cause it to fail.

EASA Airworthiness Directive (AD) was issued to require inspection and rework in order to make the centre bracket less sensitive to external damage that may result in a crack.

This AD, superseding AD 2007-0204, has been issued to introduce an alternative repeatable inspection procedure.

A failed mounting assembly, if not corrected, could result in loss of the engine. The corrective actions include an inspection to determine if there are any sharp edges on the aft engine mounting assembly; repetitive visual inspections, or a combination of visual and fluorescent penetrant inspection, for cracking of the center bracket of the aft engine mounting assembly for both engines; rework of sharp edges; replacement of the aft engine mounting assemblies; and re-identification of engine mounting assemblies and reworked center bracket. You may obtain further information by examining the MCAI in the AD docket.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the supplemental NPRM or on the determination of the cost to the public.

Conclusion

We reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed in the supplemental NPRM.

Differences Between This AD and the MCAI or Service Information

We have reviewed the MCAI and related service information and, in general, agree with their substance. But we might have found it necessary to use different words from those in the MCAI to ensure the AD is clear for U.S. operators and is enforceable. In making these changes, we do not intend to differ substantively from the information provided in the MCAI and related service information.

We might also have required different actions in this AD from those in the MCAI in order to follow our FAA policies. Any such differences are highlighted in a NOTE within the AD.

Costs of Compliance

We estimate that this AD will affect about 6 products of U.S. registry. We also estimate that it will take about 8 work-hours per product to comply with the basic requirements of this AD. The average labor rate is \$80 per work-hour. Based on these figures, we estimate the cost of this AD to the U.S. operators to be \$3,840, or \$640 per product.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of