

this AD, except for the inspections specified in paragraph (r) of this AD.

Post-Modification Inspection and Repair

(r) Within 20,000 flight cycles after the modification required by paragraph (p) or (q) of this AD, as applicable, inspect the BS 2598 bulkhead for cracks, and repair any crack before further flight, in accordance with a method approved by the Manager, Seattle ACO.

Open Hole HFEC Inspection(s) and Terminating Repair

(s) For airplanes on which the terminating modification required by paragraph (l) or (q) of this AD has not been done: Do an initial open hole HFEC inspection to detect cracks in the bulkhead splice fitting, frame support fitting, and forward and aft inner chords on the left and right sides of the BS 2598 bulkhead, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 747-53A2427, Revision 4, dated March 6, 2008. Do the initial inspection at the applicable time specified in Table 1 or 3 of paragraph 1.E., "Compliance," of the service bulletin; except, where the service bulletin specifies a compliance time after the date on the service bulletin, this AD requires compliance within the specified compliance time after the effective date of this AD.

(1) If no crack is detected, repeat the open hole HFEC inspection thereafter at intervals not to exceed 1,500 flight cycles.

(2) If any crack is detected, before further flight, repair it in accordance with the service bulletin; except, where the service bulletin specifies to contact Boeing for appropriate action, before further flight, repair the crack using a method approved in accordance with the procedures specified in paragraph (w) of this AD. Accomplishment of the aft inner chord repair in accordance with the applicable SRM specified in the Accomplishment Instructions of the service bulletin ends the repetitive open hole HFEC inspections required by paragraphs (h) and (s)(1) of this AD for that side of the bulkhead only.

Interim Modification

(t) For Group 1 airplanes, as identified in Boeing Alert Service Bulletin 747-53A2427, Revision 4, dated March 6, 2008, on which the terminating modification required by paragraph (l) or (q) of this AD has not been done: Before the accumulation of 12,000 total flight cycles, or within 1,500 flight cycles after the effective date of this AD, whichever occurs later, install the interim modification for the aft inner chords, in accordance with the Accomplishment Instructions of the service bulletin. Accomplishment of the interim modification ends the repetitive open hole and surface HFEC inspections required by paragraphs (h) and (s)(1) of this AD.

Post-Interim Modification/Repair Repetitive Surface and Open Hole HFEC Inspections

(u) For airplanes on which the interim modification required by paragraph (t) of this AD has been done or the repair of any cracked aft inner chord has been done in accordance with the SRM specified in the Accomplishment Instructions of Boeing Alert

Service Bulletin 747-53A2427, Revision 4, dated March 6, 2008, as required by paragraph (i) or (s)(2) of this AD; and on which the terminating modification required by paragraph (l) or (q) of this AD has not been done: At the applicable times specified in Table 1, 2, or 3 of paragraph 1.E., "Compliance," of the service bulletin, do a surface HFEC inspection to detect cracks on the forward side (unmodified area) of the bulkhead and open hole and surface HFEC inspections to detect cracks in the modified or repaired area, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 747-53A2427, Revision 4, dated March 6, 2008. Repeat the open hole and surface HFEC inspections thereafter at intervals not to exceed 1,500 flight cycles, until the modification required by paragraph (q) of this AD is done, as applicable; except, for airplanes on which the repair of any cracked aft inner chord has been done on only one side of the bulkhead in accordance with the applicable SRM as required by paragraph (i) or (s)(2) of this AD, the repetitive open hole and surface HFEC inspections required by paragraph (h) and (s)(1) of this AD must continue to be done for the other side of the bulkhead.

Repair of Any Cracked Inner Chord, Splice Fitting, or Frame Support Fitting

(v) If any crack is detected during any open hole or surface HFEC inspection required by paragraph (u) of this AD, before further flight, repair any cracked inner chord, splice fitting, or frame support fitting, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 747-53A2427, Revision 4, dated March 6, 2008; except, where the service bulletin specifies to contact Boeing for appropriate action, before further flight, repair the crack using a method approved in accordance with the procedures specified in paragraph (w) of this AD.

Alternative Methods of Compliance (AMOCs)

(w)(1) The Manager, Seattle ACO, FAA, ATTN: Ivan Li, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle ACO, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 917-6437; fax (425) 917-6590; has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(3) An AMOC that provides an acceptable level of safety may be used for any repair required by this AD, if it is approved by an Authorized Representative for the Boeing Commercial Airplanes Delegation Option Authorization Organization who has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(4) AMOCs approved previously in accordance with AD 2006-05-06 are approved as AMOCs for the corresponding provisions of this AD.

Issued in Renton, Washington, on September 11, 2008.

Michael Kaszycki,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. E8-22215 Filed 9-22-08; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2008-1006; Directorate Identifier 2008-NM-110-AD]

RIN 2120-AA64

Airworthiness Directives; Boeing Model 747-100, 747-100B, 747-200B, 747-200C, 747-200F, 747-300, 747SR, and 747SP Series Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: The FAA proposes to supersede an existing airworthiness directive (AD) that applies to certain Boeing Model 747-100, 747-100B, 747-200B, 747-200C, 747-200F, 747-300, 747SR, and 747SP series airplanes. The existing AD currently requires an inspection to determine if acceptable external skin doublers are installed at the stringer 6 (S-6) lap splices, between station (STA) 340 and STA 400. For airplanes without the acceptable external skin doublers, the existing AD requires repetitive related investigative actions and corrective actions if necessary. The existing AD also provides an optional terminating modification for the repetitive related investigative actions. This proposed AD would mandate the optional terminating modification. This proposed AD results from a report of cracked fastener holes at the right S-6 lap splice between STA 340 and STA 380. We are proposing this AD to prevent cracking in the fuselage skin, which could result in rapid decompression and loss of structural integrity of the airplane.

DATES: We must receive comments on this proposed AD by November 7, 2008.

ADDRESSES: You may send comments by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- *Fax:* 202-493-2251.
- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.
- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this AD, contact Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124-2207.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Ivan Li, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 917-6437; fax (425) 917-6590.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2008-1006; Directorate Identifier 2008-NM-110-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

On May 13, 2008, we issued AD 2008-10-15, amendment 39-15522

(73 FR 29042, May 20, 2008), for certain Boeing Model 747-100, 747-100B, 747-200B, 747-200C, 747-200F, 747-300, 747SR, and 747SP series airplanes. That AD requires an inspection to determine if acceptable external skin doublers are installed at the stringer 6 (S-6) lap splices, between station (STA) 340 and STA 400. For airplanes without the acceptable external skin doublers, that AD requires repetitive related investigative actions and corrective actions if necessary. That AD also provides an optional terminating modification for the repetitive related investigative actions. That AD resulted from a report of cracked fastener holes at the right S-6 lap splice between STA 340 and STA 380. We issued that AD to detect and correct cracking in the fuselage skin, which could result in rapid decompression and loss of the airplane.

Actions Since Existing AD Was Issued

The preamble to AD 2008-10-15 specifies that we consider the requirements "interim action" and that we were considering requiring the modification (installation of acceptable external skin doublers), which would terminate the repetitive related investigative actions. We now have determined that further rulemaking is indeed necessary, and this proposed AD follows from that determination.

FAA's Determination and Requirements of the Proposed AD

We have evaluated all pertinent information and identified an unsafe condition that is likely to develop on other airplanes of the same type design. For this reason, we are proposing this AD, which would supersede AD 2008-10-15 and would retain the requirements of the existing AD. This proposed AD would also mandate the terminating action that was optional in AD 2008-10-15.

Costs of Compliance

There are about 501 airplanes of the affected design in the worldwide fleet. This proposed AD would affect about 174 airplanes of U.S. registry.

The inspection for acceptable external skin doublers that is required by AD 2008-10-15 and retained in this proposed AD takes about 2 work hours per airplane, at an average labor rate of \$80 per work hour. Based on these figures, the estimated cost of the inspection for U.S. operators is \$27,840, or \$160 per airplane.

The cost for the proposed terminating action depends upon the results of the inspections. Therefore, we cannot calculate those costs because we do not

know what doubler conditions operators will find.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the proposed regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this proposed AD and placed it in the AD docket. See the **ADDRESSES** section for a location to examine the regulatory evaluation.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The Federal Aviation Administration (FAA) amends § 39.13 by removing amendment 39–15522 (73 FR 29042, May 20, 2008) and adding the following new airworthiness directive (AD):

Boeing: Docket No. FAA–2008–1006; Directorate Identifier 2008–NM–110–AD.

Comments Due Date

(a) The FAA must receive comments on this AD action by November 7, 2008.

Affected ADs

(b) This AD supersedes AD 2008–10–15.

Applicability

(c) This AD applies to Boeing Model 747–100, 747–100B, 747–200B, 747–200C, 747–200F, 747–300, 747SR, and 747SP series airplanes, certificated in any category; as identified in Boeing Alert Service Bulletin 747–53A2748, dated May 9, 2008.

Unsafe Condition

(d) This AD results from a report of cracked fastener holes at the right stringer 6 (S–6) lap splice between station (STA) 340 and STA 380. We are issuing this AD to prevent cracking in the fuselage skin, which could result in rapid decompression and loss of structural integrity of the airplane.

Compliance

(e) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

Requirements of AD 2008–10–15**Service Bulletin Reference Paragraph**

(f) The term “alert service bulletin,” as used in this AD, means the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2748, dated May 9, 2008.

Inspection for Acceptable External Skin Doublers

(g) For airplanes identified as Group 1, Configuration 2, in Boeing Alert Service Bulletin 747–53A2748, dated May 9, 2008: At the latest of the times specified in paragraphs (g)(1), (g)(2), and (g)(3) of this AD, do an external general visual inspection to determine if acceptable external skin doublers are installed at the left- and right-side S–6 lap splices, in accordance with Part 1 of the alert service bulletin.

(1) Prior to the accumulation of 10,000 total flight cycles.

(2) Within 8,000 flight cycles after a modification was done in accordance with Boeing Service Bulletin 747–53–2253.

(3) Within 15 days or 100 flight cycles after May 20, 2008 (the effective date of AD 2008–10–15), whichever occurs first.

Acceptable External Skin Doublers Found at Both Sides

(h) If, during the inspection required by paragraph (g) of this AD, acceptable external skin doublers in accordance with the alert service bulletin are found installed at both the left- and right-side S–6 lap splices, no further work is required by this AD.

Acceptable External Skin Doublers Not Found—Repetitive Related Investigative Actions and Corrective Actions

(i) If, during the inspection required by paragraph (g) of this AD, acceptable external skin doublers in accordance with alert service bulletin are not found installed at either the left- or right-side S–6 lap splice: Before further flight, do all applicable related investigative and corrective actions by doing all actions specified in Part 2 of the alert service bulletin. Repeat the applicable related investigative actions thereafter at intervals not to exceed 300 flight cycles until the modification specified in paragraph (j) of this AD is done.

New Requirement of This AD**Terminating Modification**

(j) If, during the inspection required by paragraph (g) of this AD, acceptable external skin doublers as specified in the alert service bulletin are not found installed at either the left- or right-side S–6 lap splice: Within 3,000 flight cycles after doing the initial related investigative actions in paragraph (i) of this AD, or within 300 flight cycles after the effective date of this AD, whichever occurs later, install acceptable external skin doublers at both the left- and right-side S–6 lap splices, as applicable. The installation of the acceptable skin doublers is required on the side of the airplane that does not have the doublers already. The installation includes doing an open-hole high-frequency eddy current (HFEC) inspection of the skin for cracking, and trimming out cracking as applicable. Do all actions in accordance with the alert service bulletin. Doing this installation terminates the repetitive related investigative actions required by paragraph (i) of this AD.

Note 1: The alert service bulletin refers to Boeing Service Bulletins 747–53–2253, Revision 3, dated March 24, 1994; and 747–53–2272, Revision 18, dated May 16, 2002; as additional sources of service information for accomplishment of the modification (installation of acceptable external skin doublers).

Note 2: AD 90–06–06, amendment 39–6490, requires, among other actions, a modification as specified in Boeing Service Bulletin 747–53–2253, dated December 14, 1984.

Note 3: AD 90–23–14, amendment 39–6801, requires inspections as specified in Boeing Service Bulletin 747–53–2253, Revision 2, dated March 29, 1990.

Alternative Methods of Compliance (AMOCs)

(k)(1) The Manager, Seattle Aircraft Certification Office (ACO), FAA, ATTN: Ivan Li, Aerospace Engineer, Airframe Branch,

ANM–120S, FAA, Seattle ACO, 1601 Lind Avenue, SW., Renton, Washington 98057–3356; telephone (425) 917–6437; fax (425) 917–6590; has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(3) An AMOC that provides an acceptable level of safety may be used for any repair required by this AD, if it is approved by an Authorized Representative for the Boeing Commercial Airplanes Delegation Option Authorization Organization who has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane and the approval must specifically refer to this AD.

(4) AMOCs approved previously in accordance with AD 2008–10–15 are approved as AMOCs for the corresponding provisions of this AD.

Issued in Renton, Washington, on September 12, 2008.

Michael Kaszycki,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. E8–22211 Filed 9–22–08; 8:45 am]

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DEPARTMENT OF HOMELAND SECURITY**Coast Guard****33 CFR Part 165**

[Docket No. USCG–2008–0070]

RIN 1625–AA87

Security Zone; Port of Mayaguez, Puerto Rico

AGENCY: Coast Guard, DHS.

ACTION: Notice of proposed rulemaking.

SUMMARY: The Coast Guard proposes to establish moving and fixed security zones around cruise ships entering, departing, mooring or anchoring at the Port of Mayaguez, Puerto Rico. This proposed regulation is necessary to protect cruise ships operating in this port. All vessels, with the exception of servicing pilot boat and assisting tug boats, would be prohibited from entering the security zones without the express permission of the Captain of the Port San Juan or a designated representative.

DATES: Comments and related material must reach the Coast Guard on or before November 24, 2008.