

Alternative Methods of Compliance (AMOCs)

(v)(1) The Manager, Seattle ACO, FAA, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(3) An AMOC that provides an acceptable level of safety shall be used for any repair required by this AD, if it is approved by an Authorized Representative for the Boeing Commercial Airplanes Delegation Option Authorization Organization who has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(4) AMOCs approved previously in accordance with AD 94-15-06 for airplane line numbers 630 through 814 inclusive are approved as AMOCs for the corresponding provisions of this AD if the AMOC does not involve using the existing sliding probe HFEC skin inspection method specified in Boeing Service Bulletin 747-53A2312, Revision 2, dated October 8, 1992, or an earlier version. In addition, the provisions of paragraph (r) of this AD must be applied to AMOCs approved previously in accordance with AD 94-15-06, where applicable.

(5) AMOCs approved previously in accordance with AD 94-15-06 for airplane line numbers 201 through 629 inclusive are approved as AMOCs for the corresponding provisions of this AD. In addition, the provisions of paragraph (r) of this AD must be applied to AMOCs approved previously in accordance with AD 94-15-06, where applicable.

Material Incorporated by Reference

(w) You must use Boeing Alert Service Bulletin 747-53A2312, dated June 12, 1989; Boeing Service Bulletin 747-53A2312, Revision 1, including "Addendum," dated March 29, 1990; Boeing Service Bulletin 747-53A2312, including the "Addendum," Revision 2, dated October 8, 1992; or Boeing Service Bulletin 747-53A2312, Revision 3, dated February 8, 2007; as applicable; to perform the actions that are required by this AD, unless the AD specifies otherwise. (The document number and date of Boeing Alert Service Bulletin 747-53A2312, dated June 12, 1989, are identified only on the first page of the document; no other page of the document contains this information.)

(1) The Director of the Federal Register approved the incorporation by reference of Boeing Alert Service Bulletin 747-53A2312, dated June 12, 1989; Boeing Service Bulletin 747-53A2312, Revision 1, including "Addendum," dated March 29, 1990; and Boeing Service Bulletin 747-53A2312, Revision 3, dated February 8, 2007; in accordance with 5 U.S.C. 552(a) and 1 CFR part 51.

(2) On August 24, 1994 (59 FR 37659, July 25, 1994), the Director of the Federal Register approved the incorporation by reference of Boeing Service Bulletin 747-53A2312, including the "Addendum," Revision 2, dated October 8, 1992.

(3) Contact Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124-2207, for a copy of this service information. You may review copies at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on July 23, 2008.

Ali Bahrami,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. E8-17776 Filed 8-12-08; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2008-0626 Directorate Identifier 2008-CE-035-AD; Amendment 39-15637; AD 2008-16-19]

RIN 2120-AA64

Airworthiness Directives; Pilatus Aircraft Ltd. Model PC-6 Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) issued by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

This Airworthiness Directive (AD) is prompted due to the discovery of loose self-locking stop nuts Part Number (P/N) 938.07.65.105 in the tail landing gear fastener assemblies of some PC-6 aircraft.

It is believed that this occurrence could also exist in other fastener assemblies using nuts P/N 938.07.65.105 at various identified locations in the aircraft.

If left uncorrected, the identified assemblies may become loose and not function as designed and could lead to hazardous situations.

We are issuing this AD to require actions to correct the unsafe condition on these products.

DATES: This AD becomes effective September 17, 2008.

On September 17, 2008, the Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4059; fax: (816) 329-4090.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM was published in the **Federal Register** on June 9, 2008 (73 FR 32497). That NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

This Airworthiness Directive (AD) is prompted due to the discovery of loose self-locking stop nuts Part Number (P/N) 938.07.65.105 in the tail landing gear fastener assemblies of some PC-6 aircraft.

It is believed that this occurrence could also exist in other fastener assemblies using nuts P/N 938.07.65.105 at various identified locations in the aircraft.

If left uncorrected, the identified assemblies may become loose and not function as designed and could lead to hazardous situations.

In order to prevent those conditions, the present AD requires you to replace self-locking stop nuts P/N 938.07.65.105 from the Tail Landing Gear Assembly, the Parachute Cable Assembly, the Water Tank Assembly, the Cable Tensioner Assembly, the Fuel Filter Assembly, the Hydraulic Pump Assembly and the Engine Mounts Assembly in accordance with Pilatus PC-6 Service Bulletin No. 53-002 Revision 2.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM or on the determination of the cost to the public.

Conclusion

We reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed.

Differences Between This AD and the MCAI or Service Information

We have reviewed the MCAI and related service information and, in general, agree with their substance. But we might have found it necessary to use different words from those in the MCAI to ensure the AD is clear for U.S. operators and is enforceable. In making these changes, we do not intend to differ substantively from the information provided in the MCAI and related service information.

We might also have required different actions in this AD from those in the MCAI in order to follow FAA policies. Any such differences are highlighted in a NOTE within the AD.

Costs of Compliance

Based on the service information, we estimate that this AD will affect 50 products of U.S. registry. We also estimate that it will take about 7 work-hours per product to comply with basic requirements of this AD. The average labor rate is \$80 per work-hour. Required parts will cost about \$310 per product.

Based on these figures, we estimate the cost of this AD to the U.S. operators to be \$43,500, or \$870 per product.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, part A, subpart III, section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this AD and placed it in the AD Docket.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains the NPRM, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone (800) 647-5527) is in the ADDRESSES section. Comments will be available in the AD docket shortly after receipt.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new AD:

2008-16-19 Pilatus Aircraft Ltd.:

Amendment 39-15637; Docket No. FAA-2008-0626; Directorate Identifier 2008-CE-035-AD.

Effective Date

(a) This airworthiness directive (AD) becomes effective September 17, 2008.

Affected ADs

(b) None.

Applicability

(c) This AD applies to PC-6, PC-6-H1, PC-6-H2, PC-6/350, PC-6/350-H1, PC-6/350-H2, PC-6/A, PC-6/A-H1, PC-6/A-H2, PC-6/B-H2, PC-6/B1-H2, PC-6/B2-H2, PC-6/B2-H4, PC-6/C-H2, and PC-6/C1-H2 airplanes,

manufacturer serial numbers (MSN) MSN 101 through MSN 949 and MSN 2001 through MSN 2092, certificated in any category.

Note 1: These airplanes may also be identified as Fairchild Republic Company PC-6 airplanes, Fairchild Heli Porter PC-6 airplanes, or Fairchild-Hiller Corporation PC-6 airplanes.

Subject

(d) Air Transport Association of America (ATA) Code 53: Fuselage.

Reason

(e) The mandatory continuing airworthiness information (MCAI) states:

This Airworthiness Directive (AD) is prompted due to the discovery of loose self-locking stop nuts Part Number (P/N) 938.07.65.105 in the tail landing gear fastener assemblies of some PC-6 aircraft.

It is believed that this occurrence could also exist in other fastener assemblies using nuts P/N 938.07.65.105 at various identified locations in the aircraft.

If left uncorrected, the identified assemblies may become loose and not function as designed and could lead to hazardous situations.

In order to prevent those conditions, the present AD requires you replace self-locking stop nuts P/N 938.07.65.105 from the Tail Landing Gear Assembly, the Parachute Cable Assembly, the Water Tank Assembly, the Cable Tensioner Assembly, the Fuel Filter Assembly, the Hydraulic Pump Assembly and the Engine Mounts Assembly in accordance with Pilatus PC-6 Service Bulletin No. 53-002 Revision 2.

Actions and Compliance

(f) Unless already done, do the following actions:

(1) Within the next 12 months after September 17, 2008 (the effective date of this AD), inspect and modify the fastener assemblies as instructed in paragraph 3 of Pilatus Aircraft Ltd. Pilatus PC-6 Service Bulletin No. 53-002, Revision No. 2, dated September 24, 2007.

(2) As of September 17, 2008 (the effective date of this AD), no person shall install on any PC-6 series aircraft, water tank assemblies and hydraulic pump assemblies, unless they have been previously modified following paragraph 4 of Pilatus Aircraft Ltd. Pilatus PC-6 Service Bulletin No. 53-002, Revision No. 2, dated September 24, 2007.

FAA AD Differences

Note 2: This AD differs from the MCAI and/or service information as follows: No differences.

Other FAA AD Provisions

(g) The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, Standards Office, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to *ATTN:* Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106;

telephone: (816) 329-4059; fax: (816) 329-4090. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(2) *Airworthy Product*: For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(3) *Reporting Requirements*: For any reporting requirement in this AD, under the provisions of the Paperwork Reduction Act (44 U.S.C. 3501 *et seq.*), the Office of Management and Budget (OMB) has approved the information collection requirements and has assigned OMB Control Number 2120-0056.

Related Information

(h) Refer to MCAI European Aviation Safety Agency EASA AD No.: 2008-0083, dated May 5, 2008; and Pilatus Aircraft Ltd. Pilatus PC-6 Service Bulletin No. 53-002, Revision No. 2, dated September 24, 2007, for related information.

Material Incorporated by Reference

(i) You must use Pilatus Aircraft Ltd. Pilatus PC-6 Service Bulletin No. 53-002, Revision No. 2, dated September 24, 2007, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Pilatus Aircraft Ltd., Customer Liaison Manager, CH-6371 STANS, Switzerland; telephone: +41 41 619 65 80; fax: +41 41 619 65 76; e-mail: fodermatt@pilatus-aircraft.com.

(3) You may review copies at the FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri 64106; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Kansas City, Missouri, on August 1, 2008.

James E. Jackson,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. E8-18236 Filed 8-12-08; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2008-0685 Directorate Identifier 2008-CE-037-AD; Amendment 39-15638; AD 2008-16-20]

RIN 2120-AA64

Airworthiness Directives; Diamond Aircraft Industries GmbH Model DA 42 Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) issued by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

The original designed bellcrank for the aileron control system in the wing needed to be installed with slightly bent rod ends during production of the aircraft to avoid friction and possible chafing. In addition to being a nonpreferable production practice, this creates the risk of replacement parts being installed during subsequent in-service maintenance without being bent or not being bent correctly. This condition, if not detected and corrected, could lead to chafing damage of the aileron control system and consequent loss of control of the aircraft.

We are issuing this AD to require actions to correct the unsafe condition on these products.

DATES: This AD becomes effective September 17, 2008.

On September 17, 2008, the Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Sarjapur Nagarajan, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4145; fax: (816) 329-4090.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM was published in the **Federal Register** on June 23, 2008 (73 FR 35361). That NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

The original designed bellcrank for the aileron control system in the wing needed to be installed with slightly bent rod ends during production of the aircraft to avoid friction and possible chafing. In addition to being a nonpreferable production practice, this creates the risk of replacement parts being installed during subsequent in-service maintenance without being bent or not being bent correctly. This condition, if not detected and corrected, could lead to chafing damage of the aileron control system and consequent loss of control of the aircraft.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM or on the determination of the cost to the public.

Conclusion

We reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed.

Differences Between This AD and the MCAI or Service Information

We have reviewed the MCAI and related service information and, in general, agree with their substance. But we might have found it necessary to use different words from those in the MCAI to ensure the AD is clear for U.S. operators and is enforceable. In making these changes, we do not intend to differ substantively from the information provided in the MCAI and related service information.

We might also have required different actions in this AD from those in the MCAI in order to follow FAA policies. Any such differences are highlighted in a NOTE within the AD.

Costs of Compliance

Based on the service information, we estimate that this AD will affect 156 products of U.S. registry. We also estimate that it will take about 3 work-hours per product to comply with basic requirements of this AD. The average labor rate is \$80 per work-hour. Required parts will cost about \$0 per product (warranty credit given by manufacturer) per product. Where the service information lists required parts costs that are covered under warranty, we have assumed that there will be no