

TABLE 1—ALL MATERIAL INCORPORATED BY REFERENCE

Airbus service information	Revision/issue level	Date
Service Bulletin A310–57A2088, excluding Appendix 01	Original	November 6, 2006.
Service Bulletin A310–57–2090	01	December 19, 2007.
Service Bulletin A310–57–2091, excluding Appendix 01	Original	May 22, 2007.
A310 Repair Instruction R572–49121	C	May 2007.

(1) The Director of the Federal Register approved the incorporation by reference of the service information specified in Table 2 of this AD under 5 U.S.C. 552(a) and 1 CFR part 51.

TABLE 2—NEW MATERIAL INCORPORATED BY REFERENCE

Airbus service information	Revision/issue level	Date
Service Bulletin A310–57–2090	01	December 19, 2007.
Service Bulletin A310–57–2091, excluding Appendix 01	Original	May 22, 2007.
A310 Repair Instruction R572–49121	C	May 2007.

(2) The Director of the Federal Register previously approved the incorporation by reference of Airbus Service Bulletin A310–57A2088, excluding Appendix 01, dated November 6, 2006, on February 6, 2007 (72 FR 2612, January 22, 2007).

(3) For service information identified in this AD, contact Airbus, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France.

(4) You may review copies at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741–6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on July 31, 2008.

Ali Bahrami,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. E8–18561 Filed 8–12–08; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2007–0043; Directorate Identifier 2007–NM–058–AD; Amendment 39–15632; AD 2008–16–14]

RIN 2120–AA64

Airworthiness Directives; Boeing Model 747–100, 747–100B, 747–100B SUD, 747–200B, 747–200C, 747–200F, 747–300, 747–400, 747SR, and 747SP Series Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: The FAA is superseding an existing airworthiness directive (AD), which applies to certain Boeing Model 747 series airplanes. That AD currently requires inspecting to detect cracking in certain lower lobe fuselage skin lap joints, doing repetitive inspections for cracking at certain fastener locations having countersunk fasteners, and replacing countersunk fasteners with protruding head fasteners at certain fastener locations. This new AD requires replacing a previous high-frequency eddy current (HFEC) inspection method with a new HFEC inspection method, adding a one-time inspection for cracking of certain airplanes, and terminating the adjustment factor for the inspection compliance times based on cabin differential pressure. This AD also requires inspecting additional lap joints. This AD results from reports of fuselage skin cracks found at certain countersunk fastener locations in the upper row of lap joints near the wing-to-body fairings, and from a report that the presence of Alodine-coated rivets could cause faulty results during the required inspections using the optional sliding probe HFEC inspection method specified in the existing AD. We are issuing this AD to prevent reduced structural integrity of the fuselage.

DATES: This AD becomes effective September 17, 2008.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in the AD as of September 17, 2008.

On August 24, 1994 (59 FR 37659, July 25, 1994), the Director of the Federal Register approved the incorporation by reference of Boeing Service Bulletin 747–53A2312, including the “Addendum,” Revision 2, dated October 8, 1992.

ADDRESSES: For service information identified in this AD, contact Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124–2207.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (telephone 800–647–5527) is the Document Management Facility, U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Ivan Li, Aerospace Engineer, Airframe Branch, ANM–120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98057–3356; telephone (425) 917–6437; fax (425) 917–6590.

SUPPLEMENTARY INFORMATION:

Discussion

The FAA issued a supplemental notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that supersedes AD 94–15–06, amendment 39–8977 (59 FR 37659, July 25, 1994). The existing AD applies to certain Boeing Model 747 series airplanes. That supplemental NPRM was published in the **Federal Register** on May 7, 2008 (73 FR 25601). That supplemental NPRM proposed to continue to require inspecting to detect cracking in certain lower lobe fuselage skin lap joints, doing repetitive

inspections for cracking at certain fastener locations having countersunk fasteners, and replacing countersunk fasteners with protruding head fasteners at certain fastener locations. That supplemental NPRM also proposed to require replacing a previous high-frequency eddy current (HFEC) inspection method with a new HFEC inspection method, adding a one-time inspection for cracking of certain airplanes, and terminating the adjustment factor for the inspection compliance times based on cabin differential pressure. That supplemental

NPRM also proposed to require inspecting additional lap joints.

Comments

We provided the public the opportunity to participate in the development of this AD. We have considered the single comment that has been received on the supplemental NPRM. The commenter, Boeing, supports the NPRM.

Conclusion

We have carefully reviewed the available data, including the comment that has been received, and determined

that air safety and the public interest require adopting the AD as proposed.

Costs of Compliance

There are about 348 airplanes in the worldwide fleet. We estimate that this AD affects 90 airplanes of U.S. registry. The issue associated with Alodine-coated aluminum rivets occurs on 162 airplanes in the worldwide fleet and affects 24 airplanes of U.S. registry. The following table provides the estimated costs for U.S. operators to comply with this AD. The average labor rate is \$80 per work hour.

ESTIMATED COSTS

Action	Work hours	Parts	Number of affected airplanes	Cost per airplane	Fleet cost
Inspections (required by AD 94-15-06 and retained in this AD).	14	\$0	90	\$1,120, per inspection cycle	\$100,800, per inspection cycle.
Inspections (required by AD 94-15-06 and retained in this AD).	82	0	90	\$6,560, per inspection cycle	\$590,400, per inspection cycle.
Modification (required by AD 94-15-06 and retained in this AD).	124	(¹)	90	\$9,920	\$892,800.
One-time inspection (new action).	4	0	24	\$320	\$7,680.

¹ Minimal.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, “General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and

responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a “significant regulatory action” under Executive Order 12866;
- (2) Is not a “significant rule” under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this AD and placed it in the AD docket. See the **ADDRESSES** section for a location to examine the regulatory evaluation.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The Federal Aviation Administration (FAA) amends § 39.13 by removing amendment 39-8977 (59 FR 37659, July 25, 1994) and by adding the following new airworthiness directive (AD):

2008-16-14 Boeing: Amendment 39-15632. Docket No. FAA-2007-0043; Directorate Identifier 2007-NM-058-AD.

Effective Date

(a) This AD becomes effective September 17, 2008.

Affected ADs

(b) This AD supersedes AD 94-15-06.

Applicability

(c) This AD applies to Boeing Model 747-100, 747-100B, 747-100B SUD, 747-200B, 747-200C, 747-200F, 747-300, 747-400, 747SR, and 747SP series airplanes, certificated in any category, as identified in Boeing Alert Service Bulletin 747-53A2312, Revision 3, dated February 8, 2007.

Unsafe Condition

(d) This AD results from reports of fuselage skin cracks found at certain countersunk fastener locations in the upper row of lap joints near the wing-to-body fairings, and from a report that the presence of Alodine-coated rivets could cause faulty results during the required inspections using the optional sliding probe high frequency eddy current (HFEC) inspection method specified in AD 94-15-06. We are issuing this AD to prevent reduced structural integrity of the fuselage.

Compliance

(e) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

Requirements of AD 94-15-06 With Revised Body Station and Stringer Locations

Inspections for Airplanes Having Line Numbers 201 Through 765 Inclusive

(f) For airplanes having line numbers 201 through 765 inclusive: Conduct an HFEC inspection to detect cracking of the lower lobe lap joints in the vicinity of the wing-to-body fairings, in accordance with Boeing Alert Service Bulletin 747-53A2312, dated June 12, 1989; Revision 1, dated March 29, 1990; Revision 2, dated October 8, 1992; or Revision 3, dated February 8, 2007; except as provided by paragraph (u) of this AD; at the time specified in paragraph (f)(1), (f)(2), (f)(3), or (f)(4) of this AD, as applicable. As of the effective date of this AD, only Revision 3 shall be used. Repeat this inspection thereafter at intervals not to exceed 4,000 landings until the inspection required by paragraph (j) of this AD is accomplished.

(1) For airplanes that have accumulated less than 11,200 total landings as of February 5, 1990 (the effective date of AD 90-01-07, amendment 39-6440, which was superseded by AD 94-15-06): Prior to the accumulation of 11,000 total landings, or within the next 1,000 landings after February 5, 1990, whichever occurs later.

(2) For airplanes that have accumulated 11,200 or more total landings but less than 15,201 total landings as of February 5, 1990: Within the next 1,000 landings after February 5, 1990, or prior to the accumulation of 15,500 total landings, whichever occurs earlier.

(3) For airplanes that have accumulated 15,201 or more total landings but less than 18,200 total landings as of February 5, 1990: Within the next 300 landings after February 5, 1990, or prior to the accumulation of 18,250 total landings, whichever occurs earlier.

(4) For airplanes that have accumulated 18,200 or more landings as of February 5, 1990: Within the next 50 landings after February 5, 1990.

Repair and Modification for Airplanes Having Line Numbers 201 Through 765 Inclusive

(g) For airplanes having line numbers 201 through 765 inclusive: Accomplish the requirements of paragraphs (g)(1) and (g)(2) of this AD.

(1) If any cracking is detected during the inspections required by paragraph (f) of this AD, prior to further flight, repair in accordance with Boeing Alert Service Bulletin 747-53A2312, dated June 12, 1989; Revision 1, dated March 29, 1990; Revision 2, dated October 8, 1992; or Revision 3, dated February 8, 2007; except as provided by paragraph (u) of this AD. As of the effective date of this AD, only Revision 3 shall be used.

(2) Prior to the accumulation of 20,000 total landings, or within the next 3,000 landings after February 5, 1990 (the effective date of AD 90-01-07), whichever occurs later, modify the airplane by replacing countersunk fasteners in the upper row of the lower lobe lap joints in the vicinity of the wing-to-body fairings with protruding head fasteners, in accordance with the procedures described in Boeing Alert Service Bulletin 747-53A2312, dated June 12, 1989; Revision 1, dated March 29, 1990; Revision 2, dated October 8, 1992; or Revision 3, dated February 8, 2007; except as provided by paragraph (u) of this AD. As of the effective date of this AD, only Revision 3 shall be used.

Adjustments for Cabin Differential Pressure for Airplanes Having Line Numbers 201 Through 765 Inclusive

(h) For airplanes having line numbers 201 through 765 inclusive: Before the effective date of this AD, for purposes of complying with paragraphs (f) and (g) of this AD, the number of landings may be determined to equal the number of pressurization cycles where the cabin pressure differential was greater than 2.0 psi.

(i) For airplanes having line numbers 201 through 765 inclusive: Before the effective date of this AD, for Model 747SR series airplanes only, based on continued mixed operation of lower cabin differentials, the inspection and modification compliance times specified in paragraphs (f) and (g) of this AD may be multiplied by a 1.2 adjustment factor.

General Visual Inspection for Countersunk Fasteners for All Airplanes

(j) For all airplanes: Prior to the accumulation of 11,000 total landings, or within 1,000 landings after August 24, 1994 (the effective date of AD 94-15-06), whichever occurs later, conduct a general visual inspection, unless previously accomplished within the last 3,000 landings prior to August 24, 1994, to determine if countersunk fasteners have been installed in the lap joints listed in paragraph (j)(1) or (j)(2) of this AD, as applicable, in accordance with the procedures described in Boeing Service Bulletin 747-53A2312, Revision 2, dated October 8, 1992; or Revision 3, dated February 8, 2007; except as provided by paragraph (u) of this AD. As of the effective date of this AD, only Revision 3 shall be used. Accomplishment of this inspection terminates the inspection requirements of paragraph (f) of this AD.

(1) For Model 747-100, -200, -300, -400, and 747SR series airplanes: From body stations (BS) 741 to 1000 at stringers (S)-34L, S-34R, S-39L, S-39R, S-44L, and S-44R,

and from BS 1480 to 1741 at S-34L, S-34R, S-40L, and S-40R.

(2) For Model 747SP series airplanes: From BS 560 to 800 at S-34L, S-34R, S-39L, S-39R, S-44L, and S-44R, and from BS 1640 to 1741 at S-34L, S-34R, S-40L, and S-40R.

Corrective Action for Countersunk Fasteners for All Airplanes

(k) For all airplanes: If no countersunk fastener is found in the upper row of a lap joint during the inspection required by paragraph (j) of this AD, no further action is required by this AD for that lap joint.

(l) For all airplanes: If any countersunk fastener is found in the upper row of a lap joint during the inspection required by paragraph (j) of this AD, prior to further flight, perform an HFEC inspection to detect cracking at all fastener locations in the lap joint where a countersunk fastener was found during the inspection required by paragraph (j) of this AD, in accordance with the procedures described in Boeing Service Bulletin 747-53A2312, Revision 2, dated October 8, 1992; or Revision 3, dated February 8, 2007; except as provided by paragraph (u) of this AD. As of the effective date of this AD, only Revision 3 shall be used.

Repetitive Inspections

(m) If no cracking is detected during any inspection required by paragraphs (l) and (q) of this AD, at any fastener location where a countersunk fastener was found during the inspection required by paragraph (j) or (q)(1) of this AD, repeat the HFEC inspection thereafter at intervals not to exceed 4,000 landings, in accordance with the procedures described in Boeing Service Bulletin 747-53A2312, Revision 2, dated October 8, 1992; or Revision 3, dated February 8, 2007; except as provided by paragraph (u) of this AD. As of the effective date of this AD, only Revision 3 shall be used. As an alternative to the HFEC inspection, operators may perform a detailed inspection to detect cracking at any fastener location where a countersunk fastener was found, in accordance with the procedures described in Boeing Service Alert Bulletin 747-53A2312, Revision 3, dated February 8, 2007; except as provided by paragraph (u) of this AD. Perform the detailed inspection within the next 4,000 landings after the HFEC inspection required by paragraph (l) of this AD, and repeat the inspection thereafter at intervals not to exceed 500 landings. At any of the subsequent inspection cycles, operators may use either inspection method provided that the corresponding inspection interval is used to determine the compliance time of the next inspection.

(n) If cracking is detected during any inspection required by paragraph (l), (m), (p), or (q) of this AD, at any fastener location where a countersunk fastener was found during the inspection required by paragraph (j) or (q)(1) of this AD, prior to further flight, repair and modify that lap joint in accordance with Boeing Service Bulletin 747-53A2312, Revision 2, dated October 8, 1992; or Revision 3, dated February 8, 2007; except as provided by paragraph (u) of this AD. As of the effective date of this AD, only Revision 3 shall be used. Accomplishment of

this repair and modification terminates the repetitive inspections required by paragraph (m) of this AD for that lap joint.

Modification of Countersunk Fasteners for All Airplanes

(o) For all airplanes: Prior to the accumulation of 20,000 total landings or within 1,000 landings after August 24, 1994, whichever occurs later, modify all fastener locations where a countersunk fastener was found during the inspections required by paragraph (j) of this AD, in accordance with the procedures described in Boeing Service Bulletin 747-53A2312, Revision 2, dated October 8, 1992; or Revision 3, dated February 8, 2007; except as provided by paragraph (u) of this AD. As of the effective date of this AD, only Revision 3 shall be used. For purposes of complying with the requirements of this paragraph, fastener locations that were previously modified in accordance with paragraph (g) or (n) of this AD do not need to be modified again. Accomplishment of this modification terminates the repetitive inspections required by paragraph (m) of this AD for the modified fastener locations.

Post-Modification Inspections for All Airplanes

(p) For all airplanes: Prior to the accumulation of 10,000 total landings following the modification required by paragraph (g), (n), (o), (q) or (s) of this AD, perform an HFEC inspection to detect cracking at all fastener locations where a countersunk fastener was found during the inspection required by paragraph (j) or (q)(1) of this AD, and repeat this inspection thereafter at intervals not to exceed 4,000 landings, in accordance with the procedures described in Boeing Service Bulletin 747-53A2312, Revision 2, dated October 8, 1992; or Revision 3, dated February 8, 2007; except as provided by paragraph (u) of this AD. As of the effective date of this AD, only Revision 3 shall be used.

New Requirements of This AD

General Visual Inspection for Countersunk Fasteners and Modification for Model 747SP Airplanes at Stringers S-34L, S-34R, S-40L, S-40R, and S-46L

(q) For Model 747SP series airplanes having line numbers 201 through 814 inclusive, do the actions in paragraphs (q)(1) and (q)(2) of this AD at the times specified in those paragraphs.

(1) Prior to the accumulation of 11,000 total landings, or within 1,000 landings as of the effective date of this AD, whichever occurs later, unless previously accomplished within the last 3,000 landings prior to the effective date of this AD, conduct a general visual inspection of the lap joint from BS 1640 to 1901 at S-46L, and from BS 1741 to 1901 at S-34L, S-34R, S-40L, and S-40R, to determine if countersunk fasteners have been installed in the specified area, in accordance with the procedures described in Boeing Alert Service Bulletin 747-53A2312, Revision 3, dated February 8, 2007; except as provided by paragraph (u) of this AD.

(i) If no countersunk fastener is found in the upper row of the lap joint during the

inspection, no further action is required by this AD for the lap joint.

(ii) If any countersunk fastener is found in the upper row of the lap joint, prior to further flight, perform an HFEC inspection to detect cracking at all fastener locations where a countersunk fastener was found, in accordance with the procedures described in Boeing Alert Service Bulletin 747-53A2312, Revision 3, dated February 8, 2007; except as provided by paragraph (u) of this AD.

(A) If no cracking is found, repeat the inspection thereafter in accordance with the requirements of paragraph (m) of this AD.

(B) If any cracking is found, prior to further flight, repair and modify the lap joint as required by paragraph (n) of this AD.

(2) Prior to the accumulation of 20,000 total landings, or within 1,000 landings as of the effective date of this AD, whichever occurs later, modify all fastener locations where a countersunk fastener was found, during the inspection required by paragraph (q)(1) of this AD, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 747-53A2312, Revision 3, dated February 8, 2007; except as provided by paragraph (u) of this AD. For purposes of complying with the requirements of this AD, fastener locations that were previously modified in accordance with paragraph (m) of this AD do not need to be modified again. Accomplishment of this modification terminates the repetitive inspections required by paragraph (m) of this AD for the modified fastener locations.

Adjustments to Compliance Time: Cabin Differential Pressure

(r) For the purposes of calculating the compliance threshold and repetitive intervals for actions required by paragraphs (f) and (g) of this AD, as of the effective date of this AD: All flight cycles, including the number of flight cycles in which cabin differential pressure is at 2.0 psi or less, must be counted when determining the number of flight cycles that have occurred on the airplane, and a 1.2 adjustment factor may not be used. However, for airplanes on which the repetitive intervals for the actions required by paragraph (f) of this AD have been calculated in accordance with paragraphs (h) and/or (i) of this AD by excluding the number of flight cycles in which cabin differential pressure is at 2.0 pounds psi or less, and/or by using a 1.2 adjustment factor: Continue to adjust the repetitive intervals in accordance with paragraphs (h) and/or (i) of this AD until the next inspection required by paragraph (f) of this AD is accomplished. Thereafter, no adjustment to compliance times based on paragraphs (h) and/or (i) of this AD is allowed.

Special One-Time Inspection for Cracking of Certain Airplanes

(s) For airplanes with line numbers 630 through 814 inclusive that meet the conditions specified in paragraphs (s)(1) and (s)(2) of this AD: Within 300 flight cycles after the effective date of this AD, or within 500 flight cycles after the most recent sliding probe inspection done in accordance with Boeing Alert Service Bulletin 747-53A2312, Revision 1, dated March 29, 1990; or

Revision 2, dated October 8, 1992; whichever occurs later, do a special one-time HFEC inspection or a special one-time detailed inspection for cracking, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 747-53A2312, Revision 3, dated February 8, 2007. If any cracking is found in a lap joint, before further flight, repair and modify that lap joint in accordance with Boeing Alert Service Bulletin 747-53A2312, Revision 3, dated February 8, 2007; except as provided by paragraph (u) of this AD. Accomplishment of this repair and modification terminates the repetitive inspections required by paragraph (m) of this AD for that lap joint. This special one-time inspection is not required for lap joints that have been modified in accordance with paragraph (g), (n), (o), or (q) of this AD.

(1) Airplanes that have not been modified in accordance with paragraph (g) or (o) of this AD.

(2) Airplanes on which the sliding probe HFEC inspection method specified in Boeing Alert Service Bulletin 747-53A2312, Revision 1, dated March 29, 1990; or Revision 2, dated October 8, 1992; was used during the last skin inspection required by paragraph (f), (l), or (m) of this AD.

Actions After the Special One-time Inspection if No Cracking Is Found

(t) For airplanes specified in paragraph (s) of this AD on which no cracking is found during the special one-time inspection, do the applicable repetitive inspections specified in paragraph (t)(1) or (t)(2) of this AD.

(1) If the special one-time inspection was done using the HFEC inspection method in accordance with paragraph (s) of this AD, perform the next inspection required by paragraph (m) of this AD within the next 4,000 flight cycles after doing the inspection required by paragraph (s) of this AD, and repeat the inspection thereafter in accordance with paragraph (m) of this AD.

(2) If the special one-time inspection was done using the detailed inspection method in accordance with paragraph (s) of this AD, perform the next inspection required by paragraph (m) of this AD within the next 500 flight cycles after doing the inspection required by paragraph (s) of this AD, and repeat the inspection thereafter in accordance with paragraph (m) of this AD.

Contacting the Manufacturer

(u) Where Boeing Alert Service Bulletin 747-53A2312, Revision 3, dated February 8, 2007, specifies to contact Boeing for appropriate action for a repair or inspection, before further flight, do the applicable action in paragraph (u)(1) or (u)(2) of this AD.

(1) Do the repair using a method approved in accordance with the procedures specified in paragraph (v) of this AD.

(2) Do the inspection using a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA. For a repair method to be approved by the Manager, Seattle ACO, as required by this paragraph, the Manager's approval letter must specifically refer to this AD.

Alternative Methods of Compliance (AMOCs)

(v)(1) The Manager, Seattle ACO, FAA, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(3) An AMOC that provides an acceptable level of safety shall be used for any repair required by this AD, if it is approved by an Authorized Representative for the Boeing Commercial Airplanes Delegation Option Authorization Organization who has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(4) AMOCs approved previously in accordance with AD 94-15-06 for airplane line numbers 630 through 814 inclusive are approved as AMOCs for the corresponding provisions of this AD if the AMOC does not involve using the existing sliding probe HFEC skin inspection method specified in Boeing Service Bulletin 747-53A2312, Revision 2, dated October 8, 1992, or an earlier version. In addition, the provisions of paragraph (r) of this AD must be applied to AMOCs approved previously in accordance with AD 94-15-06, where applicable.

(5) AMOCs approved previously in accordance with AD 94-15-06 for airplane line numbers 201 through 629 inclusive are approved as AMOCs for the corresponding provisions of this AD. In addition, the provisions of paragraph (r) of this AD must be applied to AMOCs approved previously in accordance with AD 94-15-06, where applicable.

Material Incorporated by Reference

(w) You must use Boeing Alert Service Bulletin 747-53A2312, dated June 12, 1989; Boeing Service Bulletin 747-53A2312, Revision 1, including "Addendum," dated March 29, 1990; Boeing Service Bulletin 747-53A2312, including the "Addendum," Revision 2, dated October 8, 1992; or Boeing Service Bulletin 747-53A2312, Revision 3, dated February 8, 2007; as applicable; to perform the actions that are required by this AD, unless the AD specifies otherwise. (The document number and date of Boeing Alert Service Bulletin 747-53A2312, dated June 12, 1989, are identified only on the first page of the document; no other page of the document contains this information.)

(1) The Director of the Federal Register approved the incorporation by reference of Boeing Alert Service Bulletin 747-53A2312, dated June 12, 1989; Boeing Service Bulletin 747-53A2312, Revision 1, including "Addendum," dated March 29, 1990; and Boeing Service Bulletin 747-53A2312, Revision 3, dated February 8, 2007; in accordance with 5 U.S.C. 552(a) and 1 CFR part 51.

(2) On August 24, 1994 (59 FR 37659, July 25, 1994), the Director of the Federal Register approved the incorporation by reference of Boeing Service Bulletin 747-53A2312, including the "Addendum," Revision 2, dated October 8, 1992.

(3) Contact Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124-2207, for a copy of this service information. You may review copies at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on July 23, 2008.

Ali Bahrami,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2008-0626 Directorate Identifier 2008-CE-035-AD; Amendment 39-15637; AD 2008-16-19]

RIN 2120-AA64

Airworthiness Directives; Pilatus Aircraft Ltd. Model PC-6 Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) issued by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

This Airworthiness Directive (AD) is prompted due to the discovery of loose self-locking stop nuts Part Number (P/N) 938.07.65.105 in the tail landing gear fastener assemblies of some PC-6 aircraft.

It is believed that this occurrence could also exist in other fastener assemblies using nuts P/N 938.07.65.105 at various identified locations in the aircraft.

If left uncorrected, the identified assemblies may become loose and not function as designed and could lead to hazardous situations.

We are issuing this AD to require actions to correct the unsafe condition on these products.

DATES: This AD becomes effective September 17, 2008.

On September 17, 2008, the Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4059; fax: (816) 329-4090.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM was published in the **Federal Register** on June 9, 2008 (73 FR 32497). That NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

This Airworthiness Directive (AD) is prompted due to the discovery of loose self-locking stop nuts Part Number (P/N) 938.07.65.105 in the tail landing gear fastener assemblies of some PC-6 aircraft.

It is believed that this occurrence could also exist in other fastener assemblies using nuts P/N 938.07.65.105 at various identified locations in the aircraft.

If left uncorrected, the identified assemblies may become loose and not function as designed and could lead to hazardous situations.

In order to prevent those conditions, the present AD requires you to replace self-locking stop nuts P/N 938.07.65.105 from the Tail Landing Gear Assembly, the Parachute Cable Assembly, the Water Tank Assembly, the Cable Tensioner Assembly, the Fuel Filter Assembly, the Hydraulic Pump Assembly and the Engine Mounts Assembly in accordance with Pilatus PC-6 Service Bulletin No. 53-002 Revision 2.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM or on the determination of the cost to the public.

Conclusion

We reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed.