

TABLE 1.—SBS ALLOWING PREVIOUS CREDIT—Continued

(5) Engine—Accessory Gearbox Starter Pad Drain—Install the Open Starter Drain Adapter (23077526 or 23083403); RRC SB No. AE 3007C-72-223, dated January 19, 2006.

Alternative Methods of Compliance (AMOC)

(k) The Manager, Chicago Aircraft Certification Office, has the authority to approve AMOCs for this AD if requested using the procedures found in 14 CFR 39.19.

(l) AMOCs currently approved for AD 99-02-51 will remain in effect until the terminating action date for this AD, September 30, 2009. After that date, these AMOCs will expire and will not be approved as AMOCs for this AD.

Related Information

(m) Contact Kyri Zaroyiannis, Aerospace Engineer, Chicago Aircraft Certification

Office, Small Airplane Directorate, FAA, 2300 E. Devon Ave., Des Plaines, IL 60018; e-mail: kyri.zaroyiannis@faa.gov; telephone (847) 294-7836; fax (847) 294-7834, for more information about this AD.

Material Incorporated by Reference

(n) You must use the service information specified in Table 2 to perform the actions required by this AD. The Director of the Federal Register approved the incorporation by reference of the documents listed in Table 2 of this AD in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Contact Rolls-Royce Corporation, P.O. Box 420,

Indianapolis, IN 46206; telephone (317) 230-3774; fax (317) 230-8084; e-mail: indy.pubs.services@rolls-royce.com, for a copy of this service information. You may review copies at the FAA, New England Region, 12 New England Executive Park, Burlington, MA; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

TABLE 2.—INCORPORATION BY REFERENCE

Service Bulletin No.	Page	Revision	Date
AE 3007A-72-274 Total Pages—6	ALL	1	November 12, 2007.
AE 3007A-72-321, AE 3007C-72-250 Total Pages—13	ALL	2	November 12, 2007.
AE 3007A-72-330 Total Pages—6	ALL	1	November 12, 2007.
AE 3007C-72-223 Total Pages—7	ALL	1	November 12, 2007.

Issued in Burlington, Massachusetts, on November 14, 2007.

Peter A. White,

Acting Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. E7-22810 Filed 11-27-07; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2006-26102; Directorate Identifier 2006-NE-36-AD; Amendment 39-15272; AD 2007-24-06]

RIN 2120-AA64

Airworthiness Directives; Societe de Motorisations Aeronautiques (SMA) SR305-230 and SR305-230-1 Reciprocating Engines

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule; request for comments.

SUMMARY: We are superseding an existing airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing

airworthiness information (MCAI) provided by the European Aviation Safety Agency (EASA) to identify and correct an unsafe condition on SMA SR305-230 and SR305-230-1 reciprocating engines. The MCAI states the following:

Over a period of time, the alteration of one electronic control unit (ECU) electronic component can cause a rapid uncontrolled power increase. Several occurrences have already been reported during engine start or during engine warm-up. This condition, if not corrected, could result in the loss of control of the aircraft if the pilot fails to react appropriately by switching to the mechanical backup mode.

We are issuing this AD to prevent a rapid uncontrolled power increase and possible loss of control of the airplane. **DATES:** This AD becomes effective December 13, 2007. The Director of the Federal Register approved the incorporation by reference of SMA Service Bulletin (SB) No. SB-01-76-005, dated December 15, 2006, as of December 13, 2007.

We must receive comments on this AD by December 28, 2007.

ADDRESSES: You may send comments by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov> and follow the instructions for sending your comments electronically.

- *Mail:* Docket Management Facility, U.S. Department of Transportation, 1200 New Jersey Avenue, SE., West Building Ground Floor, Room W12-140, Washington, DC 20590-0001.

- *Hand Delivery:* Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

- *Fax:* (202) 493-2251.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Operations office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone (800) 647-5527) is the same as the Mail address provided in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Christopher Spinney, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; e-mail: Christopher.spinney@faa.gov; telephone (781) 238-7175; fax (781) 238-7199.

SUPPLEMENTARY INFORMATION:

Discussion

On October 31, 2006, we issued AD 2006-23-08, Amendment 39-14820 (71 FR 65041, November 7, 2006). That AD required actions intended to address an unsafe condition on the products listed above. Since we issued AD 2006-23-08, SMA developed a terminating action for the unsafe condition. EASA, which is the Technical Agent for the Member States of the European Community, issued AD 2007-0033, dated February 13, 2007. That AD supersedes EASA EAD 2006-0312-E, which mandated a temporary corrective action to the rapid uncontrolled power increase. EASA AD 2007-0033 retains the requirements of EASA AD 2006-0312-E, and also requires replacing all affected ECUs with a new part number ECU as terminating action. EASA AD 2007-0033 states:

Over a period of time, the alteration of one electronic control unit (ECU) electronic component can cause a rapid uncontrolled power increase. Several occurrences have already been reported during engine start or during engine warm-up. This condition, if not corrected, could result in the loss of control of the aircraft if the pilot fails to react appropriately by switching to the mechanical backup mode.

You may obtain further information by examining the MCAI in the AD docket.

Relevant Service Information

SMA issued SB No. SB-01-76-005, dated December 15, 2006. The actions described in this service information are intended to correct the unsafe condition identified in EASA AD 2007-0033.

FAA's Determination and Requirements of This AD

This product has been approved by the aviation authority of France, and is approved for operation in the United States. Pursuant to our bilateral agreement with France, they have notified us of the unsafe condition described in the EASA AD and service information referenced above. We are issuing this AD because we evaluated all the information provided by EASA and SMA, and determined the unsafe condition exists and is likely to exist or develop on other products of the same type design. This AD requires replacing all affected ECUs with a new part number ECU, as terminating action.

FAA's Determination of the Effective Date

An unsafe condition exists that requires the immediate adoption of this AD. The FAA has found that the risk to the flying public justifies waiving notice and comment prior to adoption of this

rule because of the short compliance provided to correct the unsafe condition. Therefore, we determined that notice and opportunity for public comment before issuing this AD are impracticable and that good cause exists for making this amendment effective in fewer than 30 days.

Comments Invited

This AD is a final rule that involves requirements affecting flight safety, and we did not precede it by notice and opportunity for public comment. We invite you to send any written relevant data, views, or arguments about this AD. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2006-26102; Directorate Identifier 2006-NE-36-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this AD. We will consider all comments received by the closing date and may amend this AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this AD.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and

responsibilities among the various levels of government.

For the reasons discussed above, I certify this AD:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this AD and placed it in the AD docket.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The FAA amends § 39.13 by removing Amendment 39-14820; (71 FR 65041, November 7, 2006) and adding the following new AD:

2007-24-06 Societe de Motorisations Aeronautiques (SMA): Amendment 39-15272; Docket No. FAA-2006-26102; Directorate Identifier 2006-NE-36-AD.

Effective Date

(a) This airworthiness directive (AD) becomes effective December 13, 2007.

Affected ADs

(b) This AD supersedes AD 2006-23-08, Amendment 39-14820.

Applicability

(c) This AD applies to SMA SR305-230 and SR305-230-1 engines equipped with an electronic control unit (ECU) having one of the following part numbers (P/Ns): SF01160009-0, SF01160011-0, SP01160013, SP01160051-0, SP01160051-1, SP01160051-2, SP01160051-3, SP01160051-4, SP01160051-5, SP01160089-0, SP01160089-1, SP01160089-2. These engines are installed on, but not limited to, Cessna 182 series airplanes with Supplemental Type Certificate SA03302AT applied.

Reason

(d) European Aviation Safety Agency (EASA) AD 2007-0033, dated February 13, 2007, states:

Over a period of time, the alteration of one electronic control unit (ECU) electronic component can cause a rapid uncontrolled power increase. Several occurrences have already been reported during engine start or during engine warm-up. This condition, if not corrected, could result in the loss of control of the aircraft if the pilot fails to react appropriately by switching to the mechanical backup mode.

We are issuing this AD to prevent a rapid uncontrolled power increase and possible loss of control of the airplane.

Actions and Compliance

(e) Unless already done, do the following actions:

(1) Before further flight, check if the ECU has a P/N listed in the Applicability section of this AD and a serial number (SN) of 131 or below, except SNs 70, 71, 83, and 88. If it does, then do not operate the engine.

(2) Remove and replace the ECU with an ECU P/N SP01160089-3, using SMA Service Bulletin (SB) No. SB-01-76-005, dated December 15, 2006.

(3) No later than 30 days after the effective date of this AD, replace all remaining affected P/N ECUs with an ECU P/N SP01160089-3, using SMA SB No. SB-01-76-005, dated December 15, 2006.

(4) After the effective date of this AD, do not install a spare ECU having a P/N listed in the Applicability section of this AD as a replacement part on any SMA SR305-230 or SR305-230-1 engine.

FAA AD Differences

(f) This AD differs from the Mandatory Continuing Airworthiness Information (MCAI) and/or service information as follows:

(1) EASA AD No. 2007-0033 requires compliance with the AD by March 31, 2007.

(2) This AD, written later, requires compliance within 30 days after the effective date of the AD.

Other FAA AD Provisions

(g) Alternative Methods of Compliance: The Manager, Engine Certification Office, FAA, has the authority to approve alternative methods of compliance for this AD, if requested using the procedures found in 14 CFR 39.19.

Related Information

(h) Refer to EASA AD 2007-0033, dated February 13, 2007, for related information.

(i) Contact Christopher Spinney, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; e-mail: Christopher.spinney@faa.gov; telephone (781) 238-7175; fax (781) 238-7199 for more information about this AD.

Material Incorporated by Reference

(j) You must use Societe de Motorisations Aeronautiques Service Bulletin No. SB-01-76-005, dated December 15, 2006, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Societe de Motorisations Aeronautiques, 10-12 Rue Didier Daurat, F-18021 Bourges, France—Telephone +33 (0) 2 4867 5600; Fax: +33 (0) 2 4850 0141; e-mail: customer_services@smasr.com.

(3) You may review service information copies at the FAA, New England Region, 12 New England Executive Park, Burlington, MA 01803; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Burlington, MA, on November 15, 2007.

Peter A. White,

Acting Manager, Engine and Propeller Directorate, Aircraft Certification Service.

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2007-28987; Directorate Identifier 2007-NM-127-AD; Amendment 39-15269; AD 2007-24-03]

RIN 2120-AA64

Airworthiness Directives; Empresa Brasileira de Aeronautica S.A. (EMBRAER) Model EMB-135ER, -135KE, -135KL, and -135LR Airplanes and Model EMB-145, -145ER, -145MR, -145LR, -145XR, -145MP, and -145EP Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

It has been found the development of cracks in the forward fuselage right hand (RH) side skin during full-scale fatigue tests. Those cracks may quickly reach their critical length, reducing the aircraft structural integrity, with possible rapid decompression of the aircraft.

We are issuing this AD to require actions to correct the unsafe condition on these products.

DATES: This AD becomes effective January 2, 2008.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of January 2, 2008.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC.

FOR FURTHER INFORMATION CONTACT: Dan Rodina, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 227-2125; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM was published in the **Federal Register** on August 16, 2007 (72 FR 45963). That NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

It has been found the development of cracks in the forward fuselage right hand (RH) side skin during full-scale fatigue tests. Those cracks may quickly reach their critical length, reducing the aircraft structural integrity, with possible rapid decompression of the aircraft.

The corrective action includes rework of the aircraft structure on the forward fuselage LH (left-hand) and RH sides. You may obtain further information by examining the MCAI in the AD docket.

Relevant Service Information

EMBRAER has issued Service Bulletin 145-53-0067, Revision 02, dated August 28, 2007. We referred to EMBRAER Service Bulletin 145-53-0067, Revision 01, dated February 27, 2007, as the appropriate source of service information for doing the actions specified in the NPRM. The procedures in Revision 02 of the service bulletin are essentially the same as those procedures in Revision 01. Revision 02 revises the illustrations and makes editorial changes. We have revised paragraph (f)(1) and Table 1 of this AD to also refer to Revision 02 of the service bulletin.

Comments

We gave the public the opportunity to participate in developing this AD. We considered the comment received.

Request To Extend Grace Period

American Eagle Airlines requests that we extend the grace period specified in