

Parts Installation

(j) As of the effective date of this AD, no person may install on any airplane a horizontal stabilizer trim actuator unless it is new or has been overhauled as specified in Boeing Service Bulletins 767-27A0194 and 767-27A0195, both Revision 2, both dated July 13, 2006; or has been inspected, lubricated, and measured in accordance with paragraph (f) of this AD.

Alternative Methods of Compliance (AMOCs)

(k)(1) The Manager, Seattle Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

Issued in Renton, Washington, on July 31, 2007.

Ali Bahrami,

Manager, Transport Airplane Directorate,
Aircraft Certification Service.

[FR Doc. E7-16424 Filed 8-20-07; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2007-28844; Directorate Identifier 2007-CE-066-AD]

RIN 2120-AA64

Airworthiness Directives; Aeromot-Industria Mecanico Metalurgica Ltda. Model AMT-100/200/200S/300 Gliders

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to adopt a new airworthiness directive (AD) for the products listed above. This proposed AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

It has been found the occurrence of incorrect use of the self-locking nuts in bolts subject to rotational loads in bolted fittings of some assemblies of metallic components. Such event may result in disconnection of those fittings, which jeopardizes the

structural integrity of the aircraft or its flight controls.

The proposed AD would require actions that are intended to address the unsafe condition described in the MCAI.

DATES: We must receive comments on this proposed AD by September 20, 2007.

ADDRESSES: You may send comments by any of the following methods:

- *DOT Docket Web Site:* Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically.
- *Fax:* (202) 493-2251.
- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.
- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.
- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://dms.dot.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone (800) 647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Greg Davison, Glider Program Manager, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4130; fax: (816) 329-4090.

SUPPLEMENTARY INFORMATION:**Comments Invited**

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2007-28844; Directorate Identifier 2007-CE-066-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this

proposed AD because of those comments.

We will post all comments we receive, without change, to <http://dms.dot.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

The Departamento de Aviacao Civil (DAC), which is the aviation authority for Brazil, has issued AD No. 2005-12-01, dated January 17, 2006 (referred to after this as "the MCAI"), to correct an unsafe condition for the specified products. The MCAI states:

It has been found the occurrence of incorrect use of the self-locking nuts in bolts subject to rotational loads in bolted fittings of some assemblies of metallic components. Such event may result in disconnection of those fittings, which jeopardizes the structural integrity of the aircraft or its flight controls.

Since this condition may occur in other airplanes of the same type and affects flight safety, a corrective action is required. Thus, sufficient reason exists to request compliance with this AD in the indicated time limit.

You may obtain further information by examining the MCAI in the AD docket.

Relevant Service Information

Aeromot has issued Service Bulletin (SB) No. 200-20-102, revision B, dated January 23, 2006. The actions described in this service information are intended to correct the unsafe condition identified in the MCAI.

FAA's Determination and Requirements of the Proposed AD

This product has been approved by the aviation authority of another country, and is approved for operation in the United States. Pursuant to our bilateral agreement with this State of Design Authority, they have notified us of the unsafe condition described in the MCAI and service information referenced above. We are proposing this AD because we evaluated all information and determined the unsafe condition exists and is likely to exist or develop on other products of the same type design.

Differences Between This Proposed AD and the MCAI or Service Information

We have reviewed the MCAI and related service information and, in general, agree with their substance. But we might have found it necessary to use different words from those in the MCAI to ensure the AD is clear for U.S. operators and is enforceable. In making these changes, we do not intend to differ

substantively from the information provided in the MCAI and related service information.

We might also have proposed different actions in this AD from those in the MCAI in order to follow FAA policies. Any such differences are highlighted in a Note within the proposed AD.

Costs of Compliance

Based on the service information, we estimate that this proposed AD would affect about 56 products of U.S. registry. We also estimate that it would take about 8 work-hours per product to comply with the basic requirements of this proposed AD. The average labor rate is \$80 per work-hour. Required parts would cost about \$430 per product.

Based on these figures, we estimate the cost of the proposed AD on U.S. operators to be \$59,920 or \$1,070 per product.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that

section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this proposed AD and placed it in the AD docket.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The FAA amends § 39.13 by adding the following new AD:

Aeromot-Industria Mecanico Metalurgica

Itda.: Docket No. FAA-2007-28844; Directorate Identifier 2007-CE-066-AD.

Comments Due Date

(a) We must receive comments by September 20, 2007.

Affected ADs

(b) None.

Applicability

(c) This AD applies to the following gliders in the table below that:

- (1) are certificated in any category and
- (2) have not incorporated the actions in their entirety of Aeromot SB No. 200-20-102, revision A, dated April 19, 2005.

AIRPLANE APPLICABILITY

| Model | Serial Nos. |
|-------------------------------------|---|
| AMT-100 | 100.001 through 100.003, 100.005 through 100.015, 100.017, 100.019, 100.022 through 100.039, and 100.041 through 100.044. |
| AMT-100 (modified to AMT-200) | 100.004, 100.016, 100.018, 100.020, and 100.021. |
| AMT-200 | 200.040, 200.045 through 200.105, 200.108 through 200.111, 200.113 through 200.118, and 200.121. |
| AMT-200S | 200.119, 200.122 through 200.124, and 200.126 through 200.161. |
| AMT-300 | 300.106, 300.107, 300.115, and 300.125. |

Subject

(d) Air Transport Association of America (ATA) Code 51: Structures.

Reason

(e) The mandatory continuing airworthiness information (MCAI) states:

It has been found the occurrence of incorrect use of the self-locking nuts in bolts subject to rotational loads in bolted fittings of some assemblies of metallic components. Such even may result in disconnection of those fittings, which jeopardizes the structural integrity of the aircraft or its flight controls.

Since this condition may occur in other airplanes of the same type and affects flight safety, a corrective action is required. Thus,

sufficient reason exists to request compliance with this AD in the indicated time limit.

Actions and Compliance

(f) Unless already done, within the next 50 hours time-in-service (TIS) after the effective date of this AD, following Aeromot Service Bulletin No. 200-20-102 Rev. B, dated January 23, 2006, install new bolts, washers, and castellated nuts with cotter pins in the following areas:

- (1) both main landing gear legs,
- (2) swivel tail wheel,
- (3) eye-bolt fittings located at firewall inside cabin,
- (4) left and right rudder pedal assembly,
- (5) bellcranks of the rudder cables assembly,

- (6) bellcranks of the propeller pitch control assembly, and
- (7) left and right wing hinge point.

FAA AD Differences

No differences.

Other FAA AD Provisions

(g) The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, Standards Staff, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Greg Davison, Glider Program Manager, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4130; fax: (816)

329–4090. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(2) *Airworthy Product*: For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(3) *Reporting Requirements*: For any reporting requirement in this AD, under the provisions of the Paperwork Reduction Act (44 U.S.C. 3501 *et seq.*), the Office of Management and Budget (OMB) has approved the information collection requirements and has assigned OMB Control Number 2120–0056.

Related Information

(h) Refer to MCAI Departamento de Aviação Civil (DAC), which is the aviation authority for Brazil, AD No. 2005–12–01; and Aeromot SB No. 200–20–102, revision B, dated January 23, 2006, for related information.

Issued in Kansas City, Missouri, on August 14, 2007.

Terry L. Chasteen,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. E7–16421 Filed 8–20–07; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2007–28730; Directorate Identifier 2007–CE–063–AD]

RIN 2120–AA64

Airworthiness Directives; GARMIN International GSM 85 Servo Gearbox Units

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to adopt a new airworthiness directive (AD) for certain GARMIN International (GARMIN) GSM 85 servo gearbox units that are installed on airplanes. This proposed AD would require you to inspect the GSM 85 servo gearbox for foreign object debris and return the unit to the manufacturer for replacement if you find debris. This proposed AD results from reports of certain GARMIN GSM 85 servo gearbox

units that have foreign object debris inside the assembly. We are proposing this AD to detect and correct defective GARMIN GSM 85 servo gearbox units, which could result in jamming of the gearbox. Jamming of the gearbox could lead to the pilot having to apply excessive manual force to control the airplane.

DATES: We must receive comments on this proposed AD by October 22, 2007.

ADDRESSES: Use one of the following addresses to comment on this proposed AD:

- *DOT Docket Web site:* Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically.

- *Mail:* U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

- *Fax:* (202) 493–2251.

- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue, SE., Washington, DC 20590, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

For service information identified in this proposed AD, contact GARMIN International Inc., 1200 East 151st Street, Olathe, KS 66062; telephone: 913–397–8200; fax: 913–397–8282.

FOR FURTHER INFORMATION CONTACT: Roger A. Souter, Aerospace Engineer, FAA, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Wichita, Kansas 67209; telephone: 316–946–4134; fax: 316–946–4107; e-mail address: roger.souter@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments regarding this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include the docket number, “FAA–2007–28730; Directorate Identifier 2007–CE–063–AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of the proposed AD. We will consider all comments received by the closing date and may amend the proposed AD in light of those comments.

We will post all comments we receive, without change, to <http://dms.dot.gov>, including any personal

information you provide. We will also post a report summarizing each substantive verbal contact we receive concerning this proposed AD.

Discussion

We have received reports of certain GARMIN GSM 85 servo gearbox units having foreign object debris inside the assembly. The debris was found during installation and removal.

The GSM 85 servo gearbox extrusion (housing) is composed primarily of aluminum. The manufacturer selectively uses a tumbling process to deburr the housing, which resulted in foreign object debris collecting in the housing cavities.

We have determined that foreign object debris inside the gear-assembly housing may come loose causing the GSM 85 servo gearbox to jam.

This condition, if not corrected, could result in the GSM 85 servo gearbox unit becoming jammed. Jamming of the servo gearbox could lead to the pilot having to apply excessive manual force to control the airplane.

Relevant Service Information

We have reviewed GARMIN International, Inc. Service Bulletin No. 0713, Revision A, dated May 7, 2007; GARMIN International, Inc. Service Bulletin No. 0713, Revision B, dated May 18, 2007; GARMIN International, Inc. Service Bulletin No. 0713, Revision C, dated May 29, 2007; and GARMIN International, Inc. Service Bulletin No. 0713, Revision D, dated June 13, 2007. These service bulletins describe procedures for inspecting the GSM 85 servo gearbox for foreign object debris and returning the unit to the manufacturer for replacement if debris is found.

FAA’s Determination and Requirements of the Proposed AD

We are proposing this AD because we evaluated all information and determined the unsafe condition described previously is likely to exist or develop on other products of the same type design. This proposed AD would require you to inspect the GSM 85 servo gearbox for foreign object debris and return the unit to the manufacturer for replacement if you find debris.

Costs of Compliance

We estimate that this proposed AD would affect 900 airplanes in the U.S. registry.

We estimate the following costs to do the proposed inspection: