

(2) Have SOLO engine 2 625 02 equipped with optional slip-clutch sets, SOLO P/N 29 00 202, installed; and

(3) Are certificated in any category.

Subject

(d) Air Transport Association of America (ATA) Code 61: Propellers.

Reason

(e) The mandatory continuing airworthiness information (MCAI) states:

Instead of the hub normally used which carries the starter ring gear and the hub for the tooth belt a slip-clutch can be mounted. The unit contains the hub for the tooth belt and the starter ring gear. Occurrences during service have shown that under bad conditions excessive wear on several parts of the clutch can occur. In order to avoid further damages the affected parts of the slip clutch on the engines SOLO 2 625 01 and SOLO 2 652 02 have to be inspected and replaced, if necessary. In order to check the condition of the clutch on all engines SOLO 2 625 in future, additional procedures are installed and additional inspection terms are introduced. The cover plate in front of the slip clutch has to be replaced by a stronger plate with the No. 2042888.

The original Emergency AD has now been revised to indicate that the initial inspection of the installed slip-clutch is required when 12.5 hours in operation have been accumulated.

Actions and Compliance

(f) Unless already done, do the following actions.

(1) Before further flight after September 10, 2007 (the effective date of this AD):

(i) Remove the cover plate of the slip-clutch;

(ii) Inspect the friction pads for wear. Dimension in new condition is .335 inches (8.5 mm), the wear limit is .256 inches (6.5 mm);

(iii) Inspect the slip-clutch shoes on the contact surface to the hub for wear. The wear limit is .039 inches (1 mm);

(iv) Replace any parts found to be outside the wear limit; and

(v) Assemble the slip-clutch with a new cover plate, P/N 2042888.

(2) Every 12.5 hours time-in-service (TIS) after doing the actions required by paragraph (f)(1) of this AD, repetitively inspect the slip-clutch and replace any parts found to be outside the wear limit before further flight after the inspection in which the part(s) exceeds the specified limit.

(3) Every 25 hours TIS after September 10, 2007 (the effective date of this AD), inspect the tiltplay of the clutch drum on the hub for excessive play.

(i) With the tooth belt released, measure the play in the axial direction on the starter gear. The play limit is .024 inches (0.6 mm); and

(ii) Before further flight after any inspection in which excessive play is found, replace with an FAA-approved part that is new or overhauled by the manufacturer.

(4) Every 50 hours TIS after September 10, 2007 (the effective date of this AD), replace the slip-clutch with an FAA-approved part

that is new or overhauled by the manufacturer.

(5) As of September 10, 2007 (the effective date of this AD), only install slip-clutch cover plate P/N 2042888.

(6) Each time before the slip-clutch is mounted, degrease the taper of the crankshaft and the hub of the clutch with thinner following the instructions in the service bulletin specified in paragraph (f)(7) of this AD. The pound inches equivalent to 120 Nm is 1062.1.

(7) Do all actions required by this AD following SOLO Kleinmotoren GmbH Service Bulletin Nr. 4600-2-2, dated December 27, 2006.

FAA AD Differences

Note: This AD differs from the MCAI and/or service information as follows: No differences.

Other FAA AD Provisions

(g) The following provisions also apply to this AD:

(1) Alternative Methods of Compliance (AMOCs): The Manager, Standards Staff, FAA, ATTN: Gregory Davison, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4130; fax: (816) 329-4090, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(2) Airworthy Product: For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(3) Reporting Requirements: For any reporting requirement in this AD, under the provisions of the Paperwork Reduction Act (44 U.S.C. 3501 *et seq.*), the Office of Management and Budget (OMB) has approved the information collection requirements and has assigned OMB Control Number 2120-0056.

Related Information

(h) Refer to MCAI European Aviation Safety Agency (EASA) Emergency AD No. 2007-0001R1-E, dated January 10, 2007, and SOLO Kleinmotoren GmbH Service Bulletin Nr. 4600-2-2, dated December 27, 2006, for related information.

Material Incorporated by Reference

(i) You must use SOLO Kleinmotoren GmbH Service Bulletin Nr. 4600-2-2, dated December 27, 2006, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact SOLO Kleinmotoren GmbH,

Customer Support, 71050 Sindelfingen, Germany; telephone: +49-(0) 7031-301-210; fax: +49-(0) 7031-301-136; e-mail: wolfgang.emmerich@solo-germany.com.

(3) You may review copies at the FAA, Central Region, Office of the Regional Counsel, 901 Locust St., Room 506, Kansas City, Missouri 64016; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Kansas City, Missouri on August 14, 2007.

Terry L. Chasteen,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 07-4090 Filed 8-20-07; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2007-28971; Directorate Identifier 2007-SW-32-AD; Amendment 39-15163; AD 2007-17-05]

RIN 2120-AA64

Airworthiness Directives; Sikorsky Aircraft Corporation Model S92-A Helicopters

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments.

SUMMARY: This amendment adopts a new airworthiness directive (AD) for Sikorsky Aircraft Corporation (Sikorsky) Model S92-A helicopters. This action requires, within a specified time, borescope inspecting a certain part-numbered tail rotor pitch change shaft and bearing assembly (shaft and bearing assembly) and also inspecting after any installation. This amendment is prompted by an incident involving failure of a shaft and bearing assembly and servo clevis shaft resulting in loss of tail rotor control. The actions specified in this AD are intended to prevent failure of a shaft and bearing assembly, loss of tail rotor pitch and yaw control, and subsequent loss of control of a helicopter.

DATES: Effective August 21, 2007.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of August 21, 2007.

Comments for inclusion in the Rules Docket must be received on or before October 22, 2007.

ADDRESSES: Use one of the following addresses to submit comments on this AD:

- *DOT Docket Web site:* Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically;

- *Government-wide rulemaking Web site:* Go to <http://www.regulations.gov> and follow the instructions for sending your comments electronically;

- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590;

- *Hand Delivery:* Deliver to the "Mail" address between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays; or

- *Fax:* 202-493-2251.

You may get the service information identified in this AD from Sikorsky Aircraft Corporation, Attn: Manager, Commercial Technical Support, mailstop s581a, 6900 Main Street, Stratford, Connecticut, phone (203) 383-4866, e-mail address tsslibrary@sikorsky.com.

Examining the Docket: You may examine the docket that contains the AD, any comments, and other information on the Internet at <http://dms.dot.gov>, or in person at the Docket Operations office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Operations office (telephone (800) 647-5527) is located in Room W12-140 on the ground floor of the West Building at the street address stated in the **ADDRESSES** section. Comments will be available in the AD docket shortly after the DMS receives them.

FOR FURTHER INFORMATION CONTACT:

Wayne Gaulzetti, Aviation Safety Engineer, Boston Aircraft Certification Office, 12 New England Executive Park, Burlington, MA 01803, telephone (781) 238-7156, fax (781) 238-7170.

SUPPLEMENTARY INFORMATION: This amendment adopts a new AD for Sikorsky Model S92-A helicopters. This action requires, within a specified time, borescope inspecting a certain part-numbered shaft and bearing assembly and also inspecting after any installation. This amendment is prompted by an incident involving failure of a shaft and bearing assembly and servo clevis shaft resulting in loss of tail rotor control. This condition, if not detected, could result in loss of tail rotor pitch and yaw control and subsequent loss of control of a helicopter.

We have reviewed Sikorsky Alert Service Bulletin No. 92-64-002, dated

August 3, 2007 (ASB), which describes procedures for doing a one-time borescope inspection of the shaft and bearing assembly.

The ASB requires inspecting the shaft and bearing assembly within 50 hours time-in-service (TIS). This AD requires the inspection within 20 hours TIS based on the ease of the inspection, the availability of borescopes, the flight hours per day for the high time helicopters (about 8 hours), and the potential for a helicopter to ditch while servicing the oil rig industry. Also, this AD requires this inspection between 10 and 15 hours TIS following any installation of a shaft and bearing assembly.

The inspections required by this AD are interim actions; the manufacturer continues to investigate failure of the shaft and bearing assembly and we may either develop follow-on actions or a terminating action for the requirements of this AD.

This unsafe condition is likely to exist or develop on other helicopters of the same type design. Therefore, this AD is being issued to prevent failure of a shaft and bearing assembly, loss of tail rotor pitch and yaw control, and subsequent loss of control of a helicopter. This AD requires, within 20 hours TIS, inspecting each affected shaft and bearing assembly at the tail rotor side and on the servo side through the oil filler cap. This AD also requires borescope inspecting each shaft and bearing assembly that is installed as a replacement. This inspection must be done between 10 and 15 hours TIS after installation. Replacing any unairworthy shaft and bearing assembly is required before further flight.

The short compliance time involved is required because the previously described critical unsafe condition can adversely affect the controllability or structural integrity of the helicopter. Therefore, borescope inspecting the affected shaft and bearing assembly within 20 hours TIS and before further flight following any installation of an affected shaft and bearing assembly are required, and this AD must be issued immediately.

Since a situation exists that requires the immediate adoption of this regulation, it is found that notice and opportunity for prior public comment hereon are impracticable, and that good cause exists for making this amendment effective in less than 30 days.

We estimate that this AD will affect 34 helicopters, and the borescope inspection of the shaft and bearing assembly will take about 2 work hours to do at an average labor rate of \$80 per work hour. Required parts will cost

about \$30,864 per helicopter. Based on these figures, we estimate the total cost impact of the AD on U.S. operators to be \$1,054,816.

Comments Invited

This AD is a final rule that involves requirements that affect flight safety and was not preceded by notice and an opportunity for public comment; however, we invite you to submit any written data, views, or arguments regarding this AD. Send your comments to an address listed under **ADDRESSES**. Include "Docket No. FAA-2007-28971; Directorate Identifier 2007-SW-32-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of the AD. We will consider all comments received by the closing date and may amend the AD in light of those comments.

We will post all comments we receive, without change, to <http://dms.dot.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact with FAA personnel concerning this AD. Using the search function of our docket web site, you can find and read the comments to any of our dockets, including the name of the individual who sent the comment. You may review the DOT's complete Privacy Act Statement in the **Federal Register** published on April 11, 2000 (65 FR 19477-78), or you may visit <http://dms.dot.gov>.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared an economic evaluation of the estimated costs to comply with this AD. See the DMS to examine the economic evaluation.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. Section 39.13 is amended by adding a new airworthiness directive to read as follows:

2007-17-05 Sikorsky Aircraft Corporation: Amendment 39-15163. Docket No. FAA-2007-28971; Directorate Identifier 2007-SW-32-AD.

Applicability

Model S-92A helicopter, with a tail rotor pitch change shaft and bearing assembly (shaft and bearing assembly) part number 92358-06303-041, installed, certificated in any category.

Compliance

Required as indicated, unless accomplished previously.

To prevent failure of a shaft and bearing assembly, loss of tail rotor pitch and yaw control, and subsequent loss of control of a helicopter, do the following:

(a) Within 20 hours time-in-service (TIS), borescope inspect as follows:

(1) Inspect each affected shaft and bearing assembly at tail rotor side by following the

Accomplishment Instructions, paragraphs 3.A.(1) through (7) and Figure 4 of Sikorsky Aircraft Corporation Alert Service Bulletin No. 92-64-002, dated August 3, 2007 (ASB). If the shaft bearing fails the inspection, replace the shaft and bearing assembly before further flight.

(2) Inspect each shaft and bearing assembly on the servo side through the oil filler cap by following the Accomplishment Instructions, paragraphs B.(1) through (9) and Figures 2 and 3, of the ASB. If the shaft bearing fails the inspection, replace the shaft and bearing assembly before further flight.

Note: Maintenance Manual SA S92A-ANM-000 pertains to the subject of this AD.

(b) Between 10 and 15 hours TIS after installing a shaft and bearing assembly, borescope inspect it by following paragraph (a) of this AD.

(c) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Contact the Manager, Boston Aircraft Certification Office, FAA, ATTN: Wayne Gaulzetti, Aviation Safety Engineer, 12 New England Executive Park, Burlington, MA 01803, telephone (781) 238-7156, fax (781) 238-7170, for information about previously approved alternative methods of compliance.

(d) The inspections of the shaft and bearing assembly shall be done by following Sikorsky Alert Service Bulletin No. 92-64-002, dated August 3, 2007. The Director of the Federal Register approved this incorporation by reference in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Sikorsky Aircraft Corporation, Attn: Manager, Commercial Technical Support, mailstop s581a, 6900 Main Street, Stratford, Connecticut, phone (203) 383-4866, e-mail address tsslibrary@sikorsky.com. Copies may be inspected at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

(e) This amendment becomes effective on August 21, 2007.

Issued in Fort Worth, Texas, on August 9, 2007.

Mark R. Schilling,

Manager, Rotorcraft Directorate, Aircraft Certification Service.

[FR Doc. E7-15980 Filed 8-20-07; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2007-29014; Directorate Identifier 2007-NM-179-AD; Amendment 39-15165; AD 2007-17-07]

RIN 2120-AA64

Airworthiness Directives; Bombardier Model CL-600-2B19 (Regional Jet Series 100 & 440) Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule; request for comments.

SUMMARY: We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

The Bombardier CL-600-2B19 airplanes have had a history of flap failures at various positions for several years. Flap failure may result in a significant increase in required landing distances and higher fuel consumption than planned during a diversion.

This AD requires actions that are intended to address the unsafe condition described in the MCAI.

DATES: This AD becomes effective September 5, 2007.

The Director of the Federal Register approved the incorporation by reference of certain publications, listed in the AD as of September 5, 2007.

We must receive comments on this AD by September 20, 2007.

ADDRESSES: You may send comments by any of the following methods:

- **DOT Docket Web Site:** Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically.

- **Fax:** (202) 493-2251.

- **Mail:** U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

- **Hand Delivery:** Room W12-140 on the ground floor of the West Building, 1200 New Jersey Avenue, SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

- **Federal eRulemaking Portal:** <http://www.regulations.gov>. Follow the instructions for submitting comments.