

Rules and Regulations

Federal Register

Vol. 72, No. 111

Monday, June 11, 2007

This section of the FEDERAL REGISTER contains regulatory documents having general applicability and legal effect, most of which are keyed to and codified in the Code of Federal Regulations, which is published under 50 titles pursuant to 44 U.S.C. 1510.

The Code of Federal Regulations is sold by the Superintendent of Documents. Prices of new books are listed in the first FEDERAL REGISTER issue of each week.

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 23

[Docket No. CE254; Special Conditions No. 23-194-SC]

Special Conditions: Aviation Technology Group (ATG), Inc.; Javelin Model 100 Series Airplane; Acrobatic Spins

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final special conditions.

SUMMARY: These special conditions are issued for the Aviation Technology Group (ATG) Javelin Model 100 Series airplane. This airplane will have a novel or unusual design feature(s) associated with acrobatic spin recovery requirements. The applicable airworthiness regulations do not contain adequate or appropriate safety standards for this design feature. These special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to that established by the existing airworthiness standards.

DATES: *Effective Date:* May 29, 2007.

FOR FURTHER INFORMATION CONTACT: Lowell Foster, Federal Aviation Administration, Aircraft Certification Service, Small Airplane Directorate, ACE-111, 901 Locust, Room 301, Kansas City, Missouri, 816-329-4125, fax 816-329-4090.

SUPPLEMENTARY INFORMATION:

Background

On February 15, 2005, Aviation Technology Group (ATG); 8001 South InterPort Boulevard, Suite 310; Englewood, Colorado 80112-5951, applied for a type certificate for their new Model 100 airplane. ATG seeks certification of the Javelin in both utility and acrobatic categories. The

preliminary design includes the following features:

- Two-place, tandem configuration.
- Maximum takeoff weight of approximately 6,900 pounds.
- Design cruise speed of 500 knots calibrated airspeed.
- Two Williams FJ33-4A-18M turbofan engines with dual channel FADEC controls.
- Major airframe components constructed of carbon fiber composite materials.
- Hydraulically boosted flight control system with floor-mounted control sticks.
- Integrated avionics including electronic displays, autopilot, and flight management system.

Title 14 CFR part 23, § 23.221 contains spin requirements for normal, utility, and acrobatic category airplanes. When part 3 of the Civil Air Regulations was recodified in 1965 as 14 CFR part 23, spin requirements for acrobatic category airplanes were presented in § 23.221(c). Since 1965, the spin requirements in § 23.221(c) have been amended three times.

The original version of § 23.221(c) required an acrobatic category airplane to perform spins of at least six turns and recover without exceeding an airspeed limit or positive load factor limit. Spins were required for flaps-up configuration and flaps-down configuration. In addition, the airplane could not enter an uncontrollable spin with any use of the controls.

Amendment 23-7 revised the presentation of the acrobatic category spin requirements and revised the minimum turn requirement to six turns or three seconds, whichever takes longer. Amendment 23-42 revised § 23.221(c)(3) and clarified the term "controls" in the previous version of the rule by identifying flight controls and engine controls. It also clarified that the use of the controls could be at spin entry or during the spin. Neither of these two amendments changed the basic acrobatic category spin requirements.

In July 1994, the FAA proposed changes to the flight airworthiness standards for normal, utility, acrobatic, and commuter category airplanes. The proposals arose from the joint effort of the FAA and the European Joint Aviation Authorities (JAA) to harmonize 14 CFR regulations and the Joint

Aviation Requirements (JAR). The proposed changes were intended to provide nearly uniform flight airworthiness standards for airplanes certificated in the United States under 14 CFR part 23 and in the JAA countries under JAR 23.

Proposed changes to the introductory paragraph of § 23.221(c) required acrobatic category airplanes to meet the one-turn spin requirements of § 23.221(a) as well as the emergency egress requirements of § 23.807, and to meet the spin requirements of §§ 23.221(c)(1) through (4) in each configuration approved for spins. The addition of normal category spin requirements was necessary because acrobatic category airplanes should have sufficient controllability to recover from the developing one-turn spin under the same conditions as normal category airplanes. The configuration requirement was added to recognize the common practice of approving intentional spins only for a specific configuration (e.g. gear and flaps up). The proposed changes were incorporated into the rule by Amendment 23-50.

The FAA did not intend to approve an acrobatic category airplane that met only the normal category spin requirements. The assumption has always been that an inadvertent spin could result during the performance of a variety of acrobatic maneuvers.

Type Certification Basis

Under the provisions of 14 CFR part 21, § 21.17, ATG must show that the Model 100 meets the applicable provisions of part 23, as amended by Amendment 23-1 through 23-55 thereto. If the Administrator finds that the applicable airworthiness regulations (*i.e.*, 14 CFR part 23) do not contain adequate or appropriate safety standards for the ATG Model 100 series because of a novel or unusual design feature, special conditions are prescribed under the provisions of § 21.16.

Special conditions, as appropriate, as defined in § 11.19, are issued in accordance with § 11.38, and become part of the type certification basis in accordance with § 21.17.

Special conditions are initially applicable to the model for which they are issued. Should the type certificate for that model be amended later to include any other model that incorporates the same novel or unusual

design feature, the special conditions would also apply to the other model under the provisions of § 21.101.

Novel or Unusual Design Features

The ATG Model 100 will incorporate the following novel or unusual design features: High thrust-to-weight ratio, military training jet configuration with a higher fuselage mass compared to typical part 23 acrobatic airplanes.

Discussion

Title 14 CFR part 23, § 23.221(c), as amended by Amendment 23–50, presents acrobatic category airplane spin requirements. As the rule is currently written, the acrobatic category airplane must comply with normal category spin requirements, acrobatic category emergency egress requirements in § 23.807, and acrobatic spin requirements for each configuration requested for spin approval.

ATG proposes to prohibit intentional spins and requests that no configuration be approved for spins. This proposal appears to allow an acrobatic category airplane that meets only normal category spin requirements. This proposal is unacceptable since the FAA has always maintained that an acrobatic category airplane must comply with acrobatic category spin requirements.

Discussion of Comments

A notice of proposed special conditions No. 23–06–06–SC for the Aviation Technology Group (ATG), Inc.; Javelin Model 100 series airplanes was published in the **Federal Register** on February 1, 2007 (72 FR 4661). No comments were received, and the special conditions are adopted as proposed.

Applicability

As discussed above, these special conditions are applicable to the Aviation Technology Group (ATG), Inc.; Javelin Model 100 Series airplane. Should Aviation Technology Group apply at a later date for a change to the type certificate to include another model on the same type certificate incorporating the same novel or unusual design feature, the special conditions would apply to that model as well.

Conclusion

This action affects only certain novel or unusual design features on one model series of airplane. It is not a rule of general applicability.

List of Subjects in 14 CFR Part 23

Aircraft, Aviation safety, Signs and symbols.

■ The authority citation for these special conditions is as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701, 44702, 44704; 14 CFR 21.16 and 21.17 and 14 CFR 11.38 and 11.19.

The Special Conditions

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the following special conditions are issued as part of the type certification basis for the ATG Model 100 airplanes.

Title 14 CFR part 23, § 23.221(c) as amended by Amendment 23–50, presents acrobatic category airplane spin requirements. As the rule is currently written, the acrobatic category airplane must comply with normal category spin requirements, acrobatic category emergency egress requirements in § 23.807, and acrobatic spin requirements for each configuration requested for spin approval.

ATG proposes to prohibit intentional spins and requests that no configuration be approved for spins. This proposal leads to an acrobatic category airplane that meets only normal category spin requirements. This proposal is unacceptable since the FAA has always maintained that an acrobatic category airplane must comply with acrobatic category spin requirements despite the wording in the current rule. The rule's history coupled with preamble information for Amendment 23–50 reveals that the rule was changed to add the normal category spin requirements and to accommodate an applicant's desire to comply with the acrobatic spin requirements for at least one configuration, but not necessarily all configurations.

Since the wording of the current rule combined with ATG's proposal does not provide the level of safety envisioned for an acrobatic category airplane, the FAA adopts the following special condition under the authority of 14 CFR part 21, § 21.16 to replace § 23.221(c) in its entirety:

SC 23.221 Spinning.

(c) *Acrobatic category airplanes.* An acrobatic category airplane must meet the spin requirements of paragraph (a) of this section and § 23.807(b)(6). In addition, the following requirements must be met in an applicant-designated acrobatic configuration, and in each other configuration for which approval for spinning is requested:

(1) The airplane must recover from any point in a spin up to and including six turns, or any greater number of turns for which certification is requested, in not more than one and one-half additional turns after initiation of the first control action for recovery.

However, beyond three turns, the spin may be discontinued if spiral characteristics appear.

(2) The applicable airspeed limits and limit maneuvering load factors must not be exceeded. For flaps extended configurations for which approval is requested, the flaps must not be retracted during the recovery.

(3) It must be impossible to obtain unrecoverable spins with any use of the flight or engine power controls either at the entry into or during the spin.

(4) There must be no characteristics during the spin (such as excessive rates of rotation or extreme oscillatory motion) that might prevent a successful recovery due to disorientation or incapacitation of the pilot.

(5) If the applicant demonstrates that it is impossible for the airplane in the applicant-designated acrobatic configuration, and in each other configuration for which approval for spinning is requested, to enter a spin with any use of the flight or engine power controls, either at or after entry into the stall maneuver, the airplane is considered to meet the requirements of paragraph (c)(1) of this SC. The demonstration must be conducted in accordance with the following—

(i) Reduce the airplane speed using pitch control at a rate of approximately 1 knot per second until the pitch control reaches the stop; then, with the pitch control pulled back and held against the stop, apply full rudder control in a manner to promote spin entry for a period of 7 seconds or through a 360 degree heading change, whichever occurs first. If the 360 degree heading change is reached first, it must have taken no fewer than 4 seconds. This maneuver must be performed first with the ailerons in the neutral position, and then with the ailerons deflected opposite the direction of turn in the most adverse manner.

(ii) Power must be set in accordance with § 23.201(e)(4) without change during the maneuver. At the end of 7 seconds or a 360 degree heading change, the airplane must respond immediately and normally to primary flight controls applied to regain coordinated, unstalled flight without reversal of control effect and without exceeding the temporary control forces specified by § 23.143(c).

We believe that the above special condition, which replaces § 23.221(c) in its entirety, provides the level of safety established for a part 23 airplane certificated in the acrobatic category.

Issued in Kansas City, Missouri on May 29, 2007.

David R. Showers,

Acting Manager, Small Airplane Directorate,
Aircraft Certification Service.

[FR Doc. E7-11152 Filed 6-8-07; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2007-27193; Directorate Identifier 2007-CE-009-AD; Amendment 39-15091; AD 2007-12-13]

RIN 2120-AA64

Airworthiness Directives; Viking Air Limited (Type Certificate No. A-806 Previously Held by deHavilland Inc.) Models DHC-2 Mk. I, DHC-2 Mk. II, and DHC-2 Mk. III Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are superseding an existing airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) issued by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

A report has been received of stress corrosion cracking occurring in the wing lift strut lower clevis fitting, part number C2W-1097A.

We are issuing this AD to require actions to correct the unsafe condition on these products.

DATES: This AD becomes effective July 16, 2007.

On July 16, 2007 the Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD.

ADDRESSES: You may examine the AD docket on the Internet at <http://dms.dot.gov> or in person at the Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: George J. Duckett, Aerospace Engineer, FAA, New York Aircraft Certification Office, 10 Fifth Street, Valley Stream, New York 11581; telephone: (516) 228-7325; fax: (516) 794-5531.

Streamlined Issuance of AD

The FAA is implementing a new process for streamlining the issuance of ADs related to MCAI. The streamlined process will allow us to adopt MCAI safety requirements in a more efficient manner and will reduce safety risks to the public. This process continues to follow all FAA AD issuance processes to meet legal, economic, Administrative Procedure Act, and **Federal Register** requirements. We also continue to meet our technical decision-making responsibilities to identify and correct unsafe conditions on U.S.-certificated products.

This AD references the MCAI and related service information that we considered in forming the engineering basis to correct the unsafe condition. The AD contains text copied from the MCAI and for this reason might not follow our plain language principles.

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM was published in the **Federal Register** on March 22, 2007 (72 FR 13448) and proposed to supersede AD 88-08-02, Amendment 39-5889. That NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

A report has been received of stress corrosion cracking occurring in the wing lift strut lower clevis fitting, part number C2W-1097A.

This AD revision is being issued to allow operators the option of continuing with the existing inspection intervals in accordance with CF-85-08R3 (Part A) or incorporating the improved alternate inspection method in accordance with Part B, to permit an increase in inspection intervals.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM or on the determination of the cost to the public.

Conclusion

We reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed.

Differences Between This AD and the MCAI or Service Information

We have reviewed the MCAI and related service information and, in general, agree with their substance. But we might have found it necessary to use different words from those in the MCAI to ensure the AD is clear for U.S.

operators and is enforceable. In making these changes, we do not intend to differ substantively from the information provided in the MCAI and related service information.

We might also have required different actions in this AD from those in the MCAI in order to follow FAA policies. Any such differences are highlighted in a **Note** within the AD.

Costs of Compliance

We estimate that this AD will affect 392 products of U.S. registry. We also estimate that it will take about 7 work-hours per product to comply with basic requirements of this AD. The average labor rate is \$80 per work-hour. Based on these figures, we estimate the cost of this AD to the U.S. operators to be \$219,520, or \$560 per product.

In addition, we estimate that any necessary follow-on actions will take about 7 work-hours and require parts costing \$6,227 for each wing strut assembly, for a cost of \$6,787 per wing strut assembly. We have no way of determining the number of products that may need these actions.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this AD:

(1) Is not a "significant regulatory action" under Executive Order 12866;