

or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared an economic evaluation of the estimated costs to comply with this AD. See the DMS to examine the economic evaluation.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in subtitle VII, part A, subpart III, section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. Section 39.13 is amended by adding a new airworthiness directive to read as follows:

2006–26–51 Eurocopter Deutschland

GmbH: Amendment 39–14961. Docket No. FAA–2006–26721; Directorate Identifier 2006–SW–28–AD.

Applicability

Model MBB–BK 117 C–2 helicopters, serial number (S/N) 9075 and higher, and those helicopters from S/N 9004 up to and including 9074, on which Service Bulletin MBB BK117 C–2–67–006 has been accomplished, with a tail rotor control lever B642M1009103, installed, certificated in any category.

Note 1: The referenced Eurocopter Alert Service bulletin MBB BK 117 C–2–67–006, dated July 25, 2006, pertains to the subject of this AD. That ASB states that it offers improvements with regard to a reduction of the operational pedal control forces and involves two modifications. One of the modifications involves installing an optimized tail rotor control lever assembly "consisting of modified weights and modified control lever pre-assys." Installing the optimized tail rotor control lever assembly, in part, involved replacing the 2 control levers, part number (P/N) B642M1009102, with 2 control levers, P/N B642M1009103, and replacing the 4 weights, P/N B642M1011201, with 4 weights, P/N B642M1011202.

Compliance

Required as indicated, unless accomplished previously.

To prevent separation of the tail rotor control lever dynamic weights (weights) in flight, severe vibration, and subsequent loss of control of the helicopter, accomplish the following:

(a) Before further flight, mark the position of the weights, remove the split pins, remove the weights, and visually inspect the tail rotor control lever in the area around the split pin bore for score marks, notching, scratching, or a crack. Conduct the inspection by following the Accomplishment Instructions, paragraph 3A(1) and Figure 1 of Eurocopter Alert Service Bulletin No. MBB BK117 C–2–64A–002, dated December 21, 2006 (ASB).

(1) If you find score marks, notching, or scratches, that exceed the maintenance manual limits, or find a crack, replace the tail rotor control lever with an airworthy tail rotor control lever before further flight.

(2) If you do not find score marks, notching, scratches, or a crack, within 10 hours time-in-service (TIS), and thereafter at intervals not to exceed 25 hours TIS, repeat the visual inspection of the tail rotor control lever as described in paragraph (a) of this AD.

(3) After any repetitive inspection, if you find score marks, notching, or scratches, that exceed the maintenance manual limits or find a crack, replace the tail rotor control lever with an airworthy tail rotor control lever before further flight.

(4) Reassemble the tail rotor control lever by following the appropriate maintenance instructions and the Accomplishment Instructions, paragraph 3A(4) and Figure 1 of the ASB.

(b) To request an alternative method of compliance or a different compliance time

for this AD, follow the procedures in 14 CFR 39.19. Contact the Manager, Safety Management Group, FAA, ATTN: Charles Harrison, Aviation Safety Engineer, FAA, Rotorcraft Directorate, Fort Worth, Texas 76193–0110, telephone (817) 222–5128, fax (817) 222–5961, for information about previously approved alternative methods of compliance.

(c) Special flight permits will not be issued.

(d) The inspections and reassembly shall be done in accordance with the specified portions of Eurocopter Alert Service Bulletin No. MBB BK117 C–2–64A–002, dated December 21, 2006. The Director of the Federal Register approved this incorporation by reference in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from American Eurocopter Corporation, 2701 Forum Drive, Grand Prairie, Texas 75053–4005, telephone (972) 641–3460, fax (972) 641–3527. Copies may be inspected at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202–741–6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

(e) This amendment becomes effective on April 9, 2007, to all persons except those persons to whom it was made immediately effective by Emergency AD 2006–26–51, issued December 22, 2006, which contained the requirements of this amendment.

Note 2: The subject of this AD is addressed in Luftfahrt-Bundesamt (Federal Republic of Germany) AD D–2006–428, dated December 22, 2006.

Issued in Fort Worth, Texas, on March 14, 2007.

David A. Downey,

Manager, Rotorcraft Directorate, Aircraft Certification Service.

[FR Doc. E7–5139 Filed 3–22–07; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2006–26272; Directorate Identifier 2006–NM–153–AD; Amendment 39–14999; AD 2007–06–18]

RIN 2120–AA64

Airworthiness Directives; Airbus Model A318, A319, A320, and A321 Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: The FAA is adopting a new airworthiness directive (AD) for all

Airbus Model A318, A319, A320, and A321 airplanes. This AD requires repetitive inspections of the operation of the main landing gear (MLG) door opening sequence to determine if a defective actuator is installed, and replacing any defective actuator with a new actuator. This AD results from reports of slow operation of the MLG door opening/closing sequence due to a defective actuator. We are issuing this AD to detect and correct defective actuators of the MLG door, which could result in slow operation of the MLG door and consequent non-extension of the MLG during an emergency freefall operation.

DATES: This AD becomes effective April 27, 2007.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in the AD as of April 27, 2007.

ADDRESSES: You may examine the AD docket on the Internet at <http://dms.dot.gov> or in person at the Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street, SW., Nassif Building, Room PL-401, Washington, DC.

Contact Airbus, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France, for service information identified in this AD.

FOR FURTHER INFORMATION CONTACT: Tim Dulin, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 227-2141; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION:

Examining the Docket

You may examine the airworthiness directive (AD) docket on the Internet at <http://dms.dot.gov> or in person at the Docket Management Facility office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Management Facility office (telephone (800) 647-5227) is located on the plaza level of the Nassif Building at the street address stated in the **ADDRESSES** section.

Discussion

The FAA issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to all Airbus Model A318, A319, A320, and A321 airplanes. That NPRM was published in the **Federal Register** on December 8, 2006 (71 FR 71101). That NPRM proposed to require repetitive inspections of the operation of the main landing gear (MLG) door opening sequence to determine if a

defective actuator is installed, and to replace any defective actuator with a new actuator.

Relevant Service Information

Since we issued the NPRM, we have reviewed Airbus Service Bulletin A320-32-1309, Revision 01, dated June 19, 2006. Revision 01 specifies that no additional work is necessary for airplanes modified by any previous issue. We referred to Airbus Service Bulletin A320-32-1309, dated March 7, 2006, in the NPRM for accomplishing the specified actions. We have changed this AD to refer to Revision 01 of the service bulletin for accomplishing the specified actions, and to give credit for accomplishing the specified actions in accordance with the original issue before the effective date of this AD.

Comments

We provided the public the opportunity to participate in the development of this AD. We received no comments on the NPRM or on the determination of the cost to the public.

Conclusion

We have carefully reviewed the available data and determined that air safety and the public interest require adopting the AD with the change described previously. This change will neither increase the economic burden on any operator nor increase the scope of the AD.

Interim Action

We consider this AD interim action. If final action is later identified, we may consider further rulemaking then.

Costs of Compliance

This AD affects about 700 airplanes of U.S. registry. The inspection takes about 1 work hour per airplane, at an average labor rate of \$80 per work hour. Based on these figures, the estimated cost of the AD for U.S. operators is \$56,000, or \$80 per airplane, per inspection cycle.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in subtitle VII, part A, subpart III, section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations

for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this AD and placed it in the AD docket. See the **ADDRESSES** section for a location to examine the regulatory evaluation.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The Federal Aviation Administration (FAA) amends § 39.13 by adding the following new airworthiness directive (AD):

2007-06-18 Airbus: Amendment 39-14999. Docket No. FAA-2006-26272; Directorate Identifier 2006-NM-153-AD.

Effective Date

- (a) This AD becomes effective April 27, 2007.

Affected ADs

- (b) None.

Applicability

(c) This AD applies to all Airbus Model A318, A319, A320, and A321 airplanes, certificated in any category.

Unsafe Condition

(d) This AD results from reports of slow operation of the main landing gear (MLG) door opening/closing sequence due to a defective actuator. We are issuing this AD to detect and correct defective actuators of the MLG door, which could result in slow operation of the MLG door and consequent non-extension of the MLG during an emergency freefall operation.

Compliance

(e) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

Repetitive Inspections/Replacement

(f) At the time specified in paragraph (f)(1) or (f)(2) of this AD, as applicable: Do a general visual inspection of the operation of the MLG door opening sequence to determine if a defective actuator is installed by doing all the applicable actions, including replacing the door actuator, as applicable, specified in the Accomplishment Instructions of Airbus Service Bulletin A320-32-1309, Revision 01, dated June 19, 2006. Do all applicable replacements before further flight. Repeat the inspection thereafter at intervals not to exceed 900 flight cycles. Accomplishing the actions before the effective date of this AD in accordance with Airbus Service Bulletin A320-32-1309, dated March 7, 2006, is acceptable for compliance with the corresponding requirements in this paragraph.

(1) For airplanes on which a record of the total number of flight cycles on the MLG door actuator is available: Before the accumulation of 3,000 total flight cycles on the MLG door actuator, or within 800 flight cycles after the effective date of this AD, whichever is later.

(2) For airplanes on which a record of the total number of flight cycles on the MLG door actuator is not available: Within 800 flight cycles after the effective date of this AD.

Note 1: For the purposes of this AD, a general visual inspection is: "A visual examination of an interior or exterior area, installation, or assembly to detect obvious damage, failure, or irregularity. This level of inspection is made from within touching distance unless otherwise specified. A mirror may be necessary to enhance visual access to all exposed surfaces in the inspection area. This level of inspection is made under normally available lighting conditions such as daylight, hangar lighting, flashlight, or droplight and may require removal or opening of access panels or doors. Stands, ladders, or platforms may be required to gain proximity to the area being checked."

No Reporting/Parts Return Required

(g) Although the Accomplishment Instructions of Airbus Service Bulletin A320-32-1309, Revision 01, dated June 19, 2006, specify submitting certain information to the manufacturer and sending defective actuators back to the component manufacturer for

investigation, this AD does not include those requirements.

Alternative Methods of Compliance (AMOCs)

(h)(1) The Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

(2) Before using any AMOC approved in accordance with § 39.19 on any airplane to which the AMOC applies, notify the appropriate principal inspector in the FAA Flight Standards Certificate Holding District Office.

Related Information

(i) European Aviation Safety Agency airworthiness directive 2006-0112, dated May 15, 2006, also addresses the subject of this AD.

Material Incorporated by Reference

(j) You must use Airbus Service Bulletin A320-32-1309, Revision 01, dated June 19, 2006, to perform the actions that are required by this AD, unless the AD specifies otherwise. The Director of the Federal Register approved the incorporation by reference of this document in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Contact Airbus, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France, for a copy of this service information. You may review copies at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on March 13, 2007.

Ali Bahrami,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. E7-5221 Filed 3-22-07; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2006-26725; Directorate Identifier 2006-NM-161-AD; Amendment 39-15000; AD 2007-06-19]

RIN 2120-AA64

Airworthiness Directives; Bombardier Model DHC-8-102, -103, and -106 Airplanes and Model DHC-8-200 and DHC-8-300 Series Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: The FAA is adopting a new airworthiness directive (AD) for certain Bombardier Model DHC-8-102, -103, and -106 airplanes and Model DHC-8-200 and DHC-8-300 series airplanes. This AD requires modifying the main landing gear (MLG) and nose landing gear (NLG) handle assemblies for alternate release and the MLG retaining plate. This AD also requires doing a related investigative action and corrective action if necessary. This AD results from reports of broken or damaged MLG and NLG alternate release cables caused by rubbing and fraying at the cable-to-handle interface. We are issuing this AD to prevent breakage of the MLG and NLG alternate release cables, which, if the normal gear extension fails, could result in the inability to extend the MLG or NLG and consequent collapse of the landing gear during ground maneuvers or upon landing.

DATES: This AD becomes effective April 27, 2007.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in the AD as of April 27, 2007.

ADDRESSES: You may examine the AD docket on the Internet at <http://dms.dot.gov> or in person at the Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street, SW., Nassif Building, Room PL-401, Washington, DC.

Contact Bombardier, Inc., Bombardier Regional Aircraft Division, 123 Garratt Boulevard, Downsview, Ontario M3K 1Y5, Canada, for service information identified in this AD.

FOR FURTHER INFORMATION CONTACT: Ezra Sasson, Aerospace Engineer, Systems and Flight Test Branch, ANE-172, FAA, New York Aircraft Certification Office, 1600 Stewart Avenue, Suite 410, Westbury, New York 11590; telephone (516) 228-7320; fax (516) 794-5531.

SUPPLEMENTARY INFORMATION:**Examining the Docket**

You may examine the airworthiness directive (AD) docket on the Internet at <http://dms.dot.gov> or in person at the Docket Management Facility office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Management Facility office (telephone (800) 647-5227) is located on the plaza level of the Nassif Building at the street address stated in the **ADDRESSES** section.

Discussion

The FAA issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would