

the compliance times specified, unless the actions have already been done.

#### Service Bulletin References

(f) The term "service bulletin," as used in this AD, means the Accomplishment Instructions of the following service bulletins, as applicable:

(1) For Model A300 airplanes: Airbus Service Bulletin A300-24-0102, including Appendix 01, dated December 15, 2005;

(2) For Model A310 airplanes: Airbus Service Bulletin A310-24-2095, including Appendix 01, dated December 15, 2005; and

(3) For Model A300 B4-601, B4-603, B4-620, B4-622, B4-605R, B4-622R, F4-605R, and F4-622R airplanes, and A300 C4-605R Variant F airplanes: Airbus Service Bulletin A300-24-6092, including Appendix 01, dated December 15, 2005.

#### Inspections and Corrective Actions

(g) Within 44 months after the effective date of this AD, perform detailed inspections for discrepancies of all electrical bundles located in the leading and trailing edges of the wings, and all applicable corrective actions, by doing all of the actions in the service bulletin, except as provided by paragraph (h) of this AD. All corrective actions must be done before further flight.

**Note 1:** For the purposes of this AD, a detailed inspection is: "An intensive examination of a specific item, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at an intensity deemed appropriate. Inspection aids such as mirror, magnifying lenses, etc., may be necessary. Surface cleaning and elaborate procedures may be required."

#### Exception to Corrective Action Instructions

(h) If inadequate clearance is found between any electrical wire harness and adjacent components or structure: Before further flight, correct the inadequate clearance using a method approved by either the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA) (or its delegated agent).

#### Reporting

(i) Within 30 days after doing the inspections required by this AD, or within 30 days after the effective date of the AD, whichever is later: Submit a report of the findings (both positive and negative) of the inspections required by paragraph (g) of this AD to Airbus Engineering, c/o SE-E54, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. The report must include the airplane serial number or registration number, the number of flight cycles and flight hours on the airplane, the date of the inspection, the location of the defect, the conditions found, and the type of repair. Submitting Appendix 01 of the service bulletin to Airbus is acceptable for compliance with this requirement. Under the provisions of the Paperwork Reduction Act (44 U.S.C. 3501 *et seq.*), the Office of Management and Budget (OMB) has approved the information collection

requirements contained in this AD and has assigned OMB Control Number 2120-0056.

#### Alternative Methods of Compliance (AMOCs)

(j)(1) The Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

(2) Before using any AMOC approved in accordance with § 39.19 on any airplane to which the AMOC applies, notify the appropriate principal inspector in the FAA Flight Standards Certificate Holding District Office.

#### Related Information

(k) EASA airworthiness directive 2006-0076, dated April 3, 2006, also addresses the subject of this AD.

#### Material Incorporated by Reference

(l) You must use Airbus Service Bulletin A300-24-0102, including Appendix 01, dated December 15, 2005; Airbus Service Bulletin A310-24-2095, including Appendix 01, dated December 15, 2005; or Airbus Service Bulletin A300-24-6092, including Appendix 01, dated December 15, 2005; as applicable; to perform the actions that are required by this AD, unless the AD specifies otherwise. The Director of the Federal Register approved the incorporation by reference of these documents in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Contact Airbus, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France, for a copy of this service information. You may review copies at the Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street, SW., Room PL-401, Nassif Building, Washington, DC; on the Internet at <http://dms.dot.gov>; or at the National Archives and Records Administration (NARA). For information on the availability of this material at the NARA, call (202) 741-6030, or go to [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

Issued in Renton, Washington, on October 17, 2006.

**Jeffrey E. Duven,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. E6-17747 Filed 10-26-06; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2006-25171; Directorate Identifier 2006-CE-35-AD; Amendment 39-14807; AD 2006-22-10]

RIN 2120-AA64

#### Airworthiness Directives; Schempp-Hirth GmbH & Co. KG Models Mini-Nimbus B and Mini-Nimbus HS-7 Sailplanes

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) issued by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as a failure in the flap actuating circuit. An investigation showed that the lever at the torsional drive in the fuselage failed at the weld. We are issuing this AD to require actions to correct the unsafe condition on these products.

**DATES:** This AD becomes effective December 1, 2006.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of December 1, 2006.

**ADDRESSES:** You may examine the AD docket on the Internet at <http://dms.dot.gov> or in person at the Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street SW., Nassif Building, Room PL-401, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Gregory Davison, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4130; facsimile: (816) 329-4090.

#### SUPPLEMENTARY INFORMATION:

##### Streamlined Issuance of AD

The FAA is implementing a new process for streamlining the issuance of ADs related to MCAI. The streamlined process will allow us to adopt MCAI safety requirements in a more efficient manner and will reduce safety risks to the public. This process continues to follow all FAA AD issuance processes to meet legal, economic, Administrative Procedure Act, and **Federal Register**

requirements. We also continue to meet our technical decision-making responsibilities to identify and correct unsafe conditions on U.S.-certificated products.

This AD references the MCAI and related service information that we considered in forming the engineering basis to correct the unsafe condition. The AD contains text copied from the MCAI and for this reason might not follow our plain language principles.

#### Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM was published in the **Federal Register** on August 10, 2006 (71 FR 45744). That NPRM proposed to require reinforcing the flap drive.

#### Comments

We gave the public the opportunity to participate in developing this AD. We considered the comments received.

Jack Buster with the Modification and Replacement Parts Association (MARPA) provides comments to the MCAI AD process pertaining to how the FAA addresses publishing manufacturer service information as part of a proposed AD action. The commenter states that the rule, as proposed, attempts to require compliance with a public law by reference to a private writing (as referenced in paragraph (e) of the proposed AD). The commenter would like the FAA to incorporate by reference (IBR) the Schempp-Hirth Flugzeugbau GmbH. Technical Note.

We agree with Mr. Buster. However, we do not IBR any document in a proposed AD action, instead we IBR the document in the final rule. Since we are issuing the proposal as a final rule AD action, Schempp-Hirth Flugzeugbau GmbH. Technical Note No. 286-35/No. 328-13, EASA approved on: July 1, 2005, is incorporated by reference.

Mr. Buster requests IBR documents be made available to the public by publication in the **Federal Register** or in the Docket Management System (DMS).

We are currently reviewing issues surrounding the posting of service bulletins in the Department of Transportation's DMS as part of the AD docket. Once we have thoroughly examined all aspects of this issue and have made a final determination, we will consider whether our current practice needs to be revised.

#### Conclusion

We reviewed the available data, including the comments received, and determined that air safety and the

public interest require adopting the AD as proposed.

#### Differences Between this AD and the MCAI or Service Information

We have reviewed the MCAI and related service information and, in general, agree with their substance. But we might have found it necessary to use different words from those in the MCAI to ensure the AD is clear for U.S. operators and is enforceable in a U.S. court of law. In making these changes, we do not intend to differ substantively from the information provided in the MCAI and related service information.

We might also have required different actions in this AD from those in the MCAI in order to follow FAA policies. Any such differences are described in a separate paragraph of the AD. These requirements, if any, take precedence over the actions copied from the MCAI.

#### Costs of Compliance

We estimate that this AD will affect 13 products of U.S. registry. We also estimate that it will take about 6 work-hours per product to comply with this AD. The average labor rate is \$80 per work-hour. Required parts will cost about \$13 per product. Where the service information lists required parts costs that are covered under warranty, we have assumed that there will be no charge for these costs. As we do not control warranty coverage for affected parties, some parties may incur costs higher than estimated here. Based on these figures, we estimate the cost of this AD to the U.S. operators to be \$6,409, or \$493 per product.

#### Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

#### Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this AD and placed it in the AD Docket.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://dms.dot.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains the NPRM, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone (800) 647-5227) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

- Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new AD:

**2006-22-10 Schempp-Hirth Gmbh & Co. KG:** Amendment 39-14807; Docket No. FAA-2006-25171; Directorate Identifier 2006-CE-35-AD.

#### Effective Date

- (a) This airworthiness directive (AD) becomes effective December 1, 2006.

**Affected ADs**

(b) None.

**Applicability**

(c) This AD applies to Models Mini-Nimbus B and Mini-Nimbus HS-7 sailplanes, all serial numbers, that are certificated in any U.S. category.

**Reason**

(d) The mandatory continuing airworthiness information (MCAI) states that the aircraft manufacturer has identified, during the daily check after assembling a Mini Nimbus C, a failure in the flap actuating circuit. An investigation showed that the lever at the torsional drive in the fuselage failed at the weld. If not corrected, this condition could lead to a failure in the flap actuating circuit, which could result in reduced controllability of the sailplane.

**Actions and Compliance**

(e) Unless already done, do the following except as stated in paragraph (f) below.

(1) Within the next 90 days after December 1, 2006 (the effective date of this AD), reinforce the flap drive.

(2) Do the reinforcement following Schempp-Hirth Flugzeugbau GmbH. Technical Note No. 286-35/No. 328-13, EASA approved on: July 1, 2005.

**FAA AD Differences**

(f) None.

**Other FAA AD Provisions**

(g) The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs)*: The Manager, Standards Staff, FAA, Small Airplane Directorate, ATTN: Gregory Davison, Aerospace Engineer, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4130; fax: (816) 329-4090, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19.

(2) *Return to Airworthiness*: When complying with this AD, perform FAA-approved corrective actions before returning the product to an airworthy condition.

(3) *Reporting Requirements*: For any reporting requirement in this AD, under the provisions of the Paperwork Reduction Act, the Office of Management and Budget (OMB) has approved the information collection requirements and has assigned OMB Control Number 2120-0056.

**Related Information**

(h) This AD is related to German AD D-2005-239, Effective Date: July 22, 2005, which references Schempp-Hirth Flugzeugbau GmbH. Technical Note No. 286-35/No. 328-13, EASA approved on: July 1, 2005.

**Material Incorporated by Reference**

(i) You must use Schempp-Hirth Flugzeugbau GmbH. Technical Note No. 286-35/No. 328-13, EASA approved on: July 1, 2005, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of

this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Schempp-Hirth, Flugzeugbau GmbH, Postfach 14 43, D-73222 Kirchheim/Teck, Germany; telephone: ++ 49 7021 7298-0; fax: ++ 49 7021 7298-199; Web site: <http://www.schempp-hirth.com>, e-mail: [info@schempp-hirth.com](mailto:info@schempp-hirth.com).

(3) You may review copies at the FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri 64106; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Kansas City, Missouri, on October 19, 2006.

**James E. Jackson,**

*Acting Manager, Small Airplane Directorate, Aircraft Certification Service.*

[FR Doc. E6-17870 Filed 10-26-06; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

**[Docket No. FAA-2006-25841; Directorate Identifier 86-ANE-7; Amendment 39-14809; AD 2006-22-12]**

**RIN 2120-AA64**

**Airworthiness Directives; Hartzell Propeller Inc. Model HC-B5MP-3( )/M10282A( )+6 and HC-B5MP-3( )/M10876( ) ( ) Five-Bladed Propellers.**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule; request for comments.

**SUMMARY:** The FAA is superseding an existing airworthiness directive (AD) for Hartzell Propeller Inc. model HC-B5MP-3( )/M10282A( )+6 five-bladed propellers. That AD currently requires initial and repetitive torque check inspections on the mounting bolts on certain model Hartzell Propeller Inc. HC-B5MP-3( )/M10282A( )+6 five-bladed propellers, replacement of mounting bolts if necessary, and inspection and resurfacing of the engine and propeller mounting flanges if necessary. This AD requires the same actions but requires more detailed overhaul inspections and maintenance than the previous AD, AD 2004-21-01. This AD also adds Hartzell Propeller Inc. HC-B5MP-3( )/M10876( ) ( ) five-bladed propellers to the applicability. This AD results from

reports of fretting wear still occurring between the engine and propeller mounting flanges. The fretting wear results in loss of mounting bolt preload, causing failure of the mounting bolts. We are issuing this AD to prevent propeller separation from the airplane.

**DATES:** Effective November 13, 2006.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in the regulations as of November 13, 2006.

We must receive any comments on this AD by December 26, 2006.

**ADDRESSES:** Use one of the following addresses to comment on this proposed AD.

- *DOT Docket Web site:* Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically.

- *Government-wide rulemaking Web site:* Go to <http://www.regulations.gov> and follow the instructions for sending your comments electronically.

- *Mail:* Docket Management Facility; U.S. Department of Transportation, 400 Seventh Street, SW., Nassif Building, Room PL-401, Washington, DC 20590-0001.

- *Fax:* (202) 493-2251.

- *Hand Delivery:* Room PL-401 on the plaza level of the Nassif Building, 400 Seventh Street, SW., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

Contact Hartzell Propeller Inc. Technical Publications Department, One Propeller Place, Piqua, OH 45356; telephone (937) 778-4200; fax (937) 778-4391, for the service information identified in this AD.

**FOR FURTHER INFORMATION CONTACT:** Tim Smyth, Aerospace Engineer, Chicago Aircraft Certification Office, FAA, Small Airplane Directorate, 2300 East Devon Avenue, Des Plaines, IL 60018; *telephone:* (847) 294-7132; *fax:* (847) 294-7834.

**SUPPLEMENTARY INFORMATION:** On October 4, 2004, the FAA issued AD 2004-21-01, Amendment 39-13822 (69 FR 62179, October 25, 2004). That AD requires initial and repetitive torque check inspections on the mounting bolts on certain model Hartzell Propeller Inc. model HC-B5MP-3( )/M10282A( )+6 five-bladed propellers, and replacement of mounting bolts if necessary. That AD also reduces compliance time from the previous AD, for the initial inspection on certain Short Brothers Ltd. Model SD3-30 airplanes to before further flight and within 100 hours time-in-service for propellers installed on certain Aerospatiale (Nord) Model 262A airplanes. That AD also requires repetitive torque check inspections of